EXTRA 200

PILOT'S OPERATING HANDBOOK
AND
LBA APPROVED AIRPLANE FLIGHT MANUAL

Doc-No. EA - 07701

NATIONALITY AND REGISTRATION MARKS

USA / N149C

DESIGNATION OF AIRCRAFT

EXTRA 300/200

SERIAL NO/YEAR OF MANUFACTURE

08 / 1997

MANUFACTURER

EXTRA FLUGZEUGBAU GmbH
Flugplatz Dinslaken
46569 Hünxe, West Germany

APPROVED IN NORMAL AND ACROBATIC
CATEGORY BASED ON LBA REGULATIONS
AND US FAR 23 AMDT 34.

FAA APPROVED FOR U.S. REGISTERED AIRCRAFT
IN ACCORDANCE WITH FAR 21.29.

LBA-approved:

Date: 12-Aug-96

Signed:

Date: 12.08.96

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THIS HANDBOOK SHALL ALWAYS BE CARRIED ON BOARD DURING FLIGHT.

PILOTS OPERATING HANDBOOK PREPARED BY:

EXTRA Flugzeugproduktions- und Vertriebs- GmbH

THIS MANUAL IS FURNISHED TO THE CIVIL AVIATION AUTHORITIES AS A PART OF THE CERTIFICATION-MATERIAL FOR THIS MODEL.

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PUBLICATION GUIDANCE

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1 INTRODUCTION

This handbook contains 10 sections, and includes the material required to be furnished to the pilot by FAR Part 23 . It also contains supplementary data supplied by EXTRA Flugzeugproduktions- und Vertriebs- GmbH.

2 NOTES

2.1 This Flight Manual applies only to the aircraft whose nationality and registration marks are noted on the title page.

2.2 It is the responsibility of the pilot to be familiar with the contents of this Flight Manual including revisions and any relevant supplements.

2.3 Pages of this Airplane Flight Manual must not be exchanged and no alterations of or additions to the approved contents may be made without the EXTRA Flugzeugproduktions- und Vertriebs- GmbH/LBA approval.
The editor has the copyright of this Flight Manual and is responsible for edition of revisions/ amendments and supplements.

2.4 Amendments, which affect the airworthiness of the aircraft will be announced in the publication Lufttüchtigkeitsanweisung (airworthiness directive) issued by LBA, Luftfahrt-Bundesamt, or by the manufacturer EXTRA Flugzeugproduktions- und Vertriebs- GmbH. The owner is responsible for incorporating prescribed amendments and should make notes about these on the records of amendments.

2.5 Should this Flight Manual get lost, inform EXTRA Flugzeugproduktions- und Vertriebs- GmbH, Dinslaken 46569 Hünxe, Federal Republic of Germany.

2.6 Should this Flight Manual be found, kindly forward it to the civil board of aviation in the country the aircraft is registered.

2.7 This Flight Manual is only valid in connection with the latest, new LBA approved revision.

3 WARNINGS, CAUTIONS AND NOTES

The following definitions apply to Warnings, Cautions, and Notes:

WARNING

=> Operating procedures, techniques, etc which could result in personal injury or loss of life if not carefully followed

CAUTION

=> Operating procedures, techniques, etc , which could result in damage to equipment if not carefully followed.

NOTE

=> An operating procedures, technique, etc which is considered essential to emphasize.

"Shall, "Will", "Should" and "May"

The words "Shall" or, "will" shall be used to express a mandatory requirement. The word "should" shall be used to express nonmandatory provisions. The word "may" shall be used to express permissible.

4 LOG OF EFFECTIVE PAGES

Dates of issue for original and revised pages:	Date and sign of LBA approval:
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SECTION 1

GENERAL

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1.0 DESCRIPTION

This description belongs to aircraft type EXTRA 200 with nationality and registration marks:

Manufacturing

The airframe is built of tig-welded steel-tube construction. Wings, rudder and landing gear are manufactured of composite material.

The aircraft is a two-seater with the rear seat instrumented for pilot in command.

1.1 SPECIFICATION OF CLASS

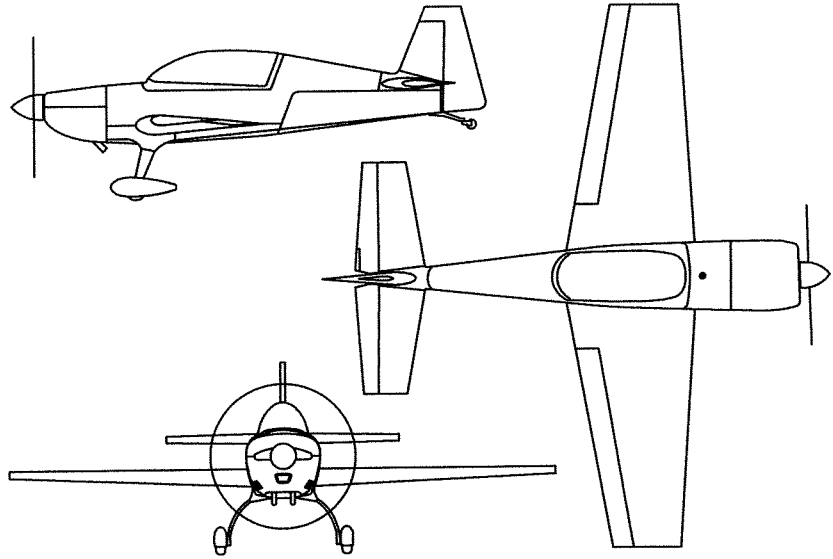
The aircraft is certified in normal and acrobatic category. LBA - Certificate No. 1086.

1.2 MANUFACTURER

Manufacturer EXTRA Flugzeugproduktions- und Vertriebs- GmbH, Flugplatz Dinslaken, 46569 Hünxe, Federal Republic of Germany.

1.3 TECHNICAL DATA

1.3.1 3-VIEW DRAWING



1.3.2 MAIN DATA

- Length	6,51 m (22,25 ft)
- Height	2,67m (8,76 ft)
- Span	7.50 m (24.61 ft)
- Wheelbase	1,80 m (5,91 ft)
- Wheel-track	4.91 m (16.10 ft)

1.3.3 WING

- Wing span	7.5 m (24.61 ft)
- Wing-area	10,44 m² (111.6 ft²)
- Airfoil	Root: MA 15 S. Tip, MA 12 S
- Chord	Root: 1,85 m. Tip, 0,93m
- MAC	1,39 m (4,56 ft)
- Aileron area	2 x 0,855 m² (2 x 9,20 ft²)
- Aileron deflection	up 30°, down 30°, tolerance -2°

1.3.4 HORIZONTAL TAIL

- Span	3,20 m (10,50 ft)
- Area	2,56 m² (27.56 ft²)
- Airfoil	Wortmann FX 71-L-150/30

1.3.5 ELEVATOR

- Area	0,77 m² (8,29 ft²)
- Elevator-deflection	up 22°; down 22°; tolerance -2°
- Trim-tab-deflection	> 15°,

1.3.6 VERTICAL TAIL

- Area	1,39 m² (14,96 ft²)
- Airfoil	Wortmann FX 71-L-150/30

1.3.7 RUDDER

- Area	0,51 m² (5,49 ft²)
- Rudder deflection	left 30°; right 30°; tolerance -2°

1.4 ENGINE

Manufacturer: Textron-Lycoming Williamsport Plant PA 17701 USA.

Type: Textron Lycoming AEIO-360-A1E

Max. Takeoff Power:	200 HP @ 2700 RPM with full throttle in MSL
Max. Continuous Power	185 HP @ 2500 RPM with full throttle in MSL

1.5 PROPELLER

Manufacturer: MT-Propeller Entwicklung GmbH, Federal Republic of Germany.

Type: MTV-12-B-C/C183-17e 3-blade constant speed.

1.5.1 EXHAUST SYSTEMS (OPTIONAL)

Manufacturer: Gomolzig Flugzeug- und Maschinenbau GmbH, Federal Republic of Germany

Exhaust Silencer for Standard System:	PN: EA 200-606000
Complete 4 in 1 System with integrated Silencer:	PN: EA 200-606000/1

1.6 FUEL

Fuel type AVGAS 100/100 LL (for alternate fuel grades see latest issue of Textron Lycoming S.I. No 1070)

Total fuel capacity:	122 liters	(32.1 US Gallon)
- Wingtanks (2 x 43 l)	86 liters	(22.6 US Gallon)
- Acro tank	36 liters	(9.5 US Gallon)
Usable fuel capacity:	117 L	(30.8 US Gallon)
- Wingtanks:	85 L	(22.4 US Gallon)
- Acro tank :	32 L	(8.5 US Gallon)

1.7 OIL

Sump capacity:	8 qts.
Minimum capacity for acro:	6 qts
Normal:	4 qts.

Average ambient air temperature	Mil-L6082 grades	Mil-22851 ashless dispersant grades
All temperatures	----	SAE 15W50 or 20W50
> 27°C (80°F)	SAE 60	SAE 60
> 16°C (60°F)	SAE 50	SAE 40 or 60
- 1°C till 32°C (30°F - 90°F)	SAE 40	

1.7 **Oil (Cont.)**

Average ambient air temperature	Mil-L6082 grades	Mil-22851 ashless dispersant grades
- 18°C till 21°C (0°F - 70°F)	SAE 30	SAE 30, 40 or 20W40
- 18°C till 32°C (0°F - 90°F)	SAE 20W50	SAE 20W50 or 15W50
< -12°C (10°F)	SAE 20	SAE 30 or 20W30

(single or multi - viscosity aviation grade oils see latest issue of Textron Lyc. S.I. No. 1014)

1.8 **LOADING**

Wing loading	80.5 kg/m²	Normal
	67.0 / 76.6 kg/m²	Acrobatic (1 seat / 2 seats)
Power loading	4.2 kg/hp	Normal
	3.5 / 4.0 kg/hp	Acrobatic (1 seat / 2 seats)

1.9 **TERMINOLOGY**

Air Speeds

CAS	Calibrated Air Speed. CAS is the same as TAS (True Air Speed) in standard atmospheric condition at sea level
KCAS	Calibrated speed in knots
GS	Ground speed
IAS	Indicated air speed
KIAS	Indicated speed in knots
TAS	True air speed. It's the same as CAS compensated for altitude, temperature and density
V_A	Maneuvering speed
V_{NE}	Never exceed speed
V_{NO}	Maximum structural cruising speed
V_S	Stalling speed or minimum steady flight speed
V_X	Best angle-of-climb speed

Meteorological terminology

ISA	International standard atmospheric condition
OAT	Outside air temperature

1.10 SECONDARY TERMINOLOGY

FPM	Feet/minute
ft	Feet = 0.3048 m
inch	inch = 2.54 cm
m	Meter
L	Liters
Gal	US gallon = 3.79 liters
Qts	US quart = 0.946 liters
hp	Horse power (english)
h	Hour
kts	Knots (NM/h) = 1.852 kilometer per hour
km/h	kilometer per hour
Lbs	English pound = 0.4536 kg
hPA	hekto Pascal
IN HG	Inches of mercury
MP	Manifold pressure
PA	Pressure altitude (ft)
NM	Nautical miles = 1.852 km
RPM	Revolutions per minute
C.G.	Center of gravity
Arm	Arm is the horizontal distance from reference datum
Moment	is the product of weight of an item multiplied by its arm.

1.11 CONVERSION TABLE

knots <> km/h		km/h <> knots		ft <> m		m <> ft		NM <> km		km <> NM	
60	111	100	54	500	152	250	820	10	19	10	5
65	120	110	59	1000	305	375	1230	20	37	20	11
70	130	120	65	1500	457	500	1640	30	56	30	16
75	139	130	70	2000	610	625	2051	40	74	40	22
80	148	140	76	2500	762	750	2461	50	93	50	27
85	157	150	81	3000	914	875	2871	60	111	60	32
90	167	160	86	3500	1067	1000	3281	70	130	70	38
95	176	170	92	4000	1219	1125	3691	80	148	80	43
100	185	180	97	4500	1372	1250	4101	90	167	90	49
105	194	190	103	5000	1524	1375	4511	100	185	100	54
110	204	200	108	5500	1676	1500	4921	110	204	110	59
115	213	210	113	6000	1829	1625	5331	120	222	120	65
120	222	220	119	6500	1981	1750	5741	130	241	130	70
125	232	230	124	7000	2134	1875	6152	140	259	140	76
130	241	240	130	7500	2286	2000	6562	150	278	150	81
135	250	250	135	8000	2438	2125	6972	160	296	160	86
140	259	260	140	8500	2591	2250	7382	170	315	170	92
145	269	270	146	9000	2743	2375	7792	180	333	180	97
150	278	280	151	9500	2896	2500	8202	190	352	190	103
155	287	290	157	10000	3048	2625	8612	200	370	200	108
160	296	300	162	10500	3200	2750	9022	220	407	250	135
165	306	310	167	11000	3353	2875	9432	240	444	300	162
170	315	320	173	11500	3505	3000	9843	260	482	350	189
175	324	330	178	12000	3658	3125	10253	280	519	400	216
180	333	340	184	12500	3810	3250	10663	300	556	450	243
185	343	350	189	13000	3962	3375	11073	320	593	500	270
190	352	360	194	13500	4115	3500	11483	340	630	550	297
195	361	370	200	14000	4267	3625	11893	360	667	600	324
200	370	380	205	14500	4420	3750	12303	380	704	650	351
205	380	390	211	15000	4572	3875	12713	400	741	700	378
210	389	400	216	15500	4724	4000	13123	420	778	750	405
215	398	410	221	16000	4877	4125	13533	440	815	800	432
220	407	420	227	16500	5029	4250	13944	460	852	850	459
225	417	430	232	17000	5182	4375	14354	480	889	900	486
230	426	440	238	17500	5334	4500	14764	500	926	950	513
235	435	450	243	18000	5486	4625	15174	520	963	1000	540

SECTION 2

LIMITATIONS

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SECTION 2

LIMITATIONS

2.1 GENERAL

This section includes operating limitations, instrument markings, and basic placards necessary for the safe operation of the aircraft, its engine, standard systems, and standard equipment. The limitations included in this section have been approved by the Luftfahrt-Bundesamt (LBA). Observance of these operating limitations is required by national aviation regulations.

NOTE

In case of an aircraft equipped with specific options additional information required for safe operation will be contained in Section 9 "Supplements".

Instrument markings and placards are provided for the acrobatic category only; for normal category refer to corresponding limitations. This aircraft is certified under LBA-Flugzeugkennnblatt No. 1086, Type Certification Data Sheet (T.C.D.S.).

Any exceeding of given limitations have to be reported by the pilot and considered by corresponding maintenance or inspection procedure according to the *SERVICE MANUAL* .

2.2 AIR SPEED (IAS)

Never Exceed Speed	V_{NE}	217 KIAS	(402 km/h)
Max. Structural Cruising Speed	V_{NO}	154 KIAS	(285 km/h)
Maneuver Speed (Normal Cat.)	V_A	138 KIAS	(256 km/h)
Maneuver Speed (Acro I , Acro II)	V_A	154 KIAS	(285 km/h)

2.3 CROSSWIND COMPONENT

Max. demonstrated crosswind component for takeoff and landing 15 knots (27 km/h).

2.4 ENGINE

Engine-type: Textron-Lycoming AEIO-360-A1E with

Max. Takeoff Power:	200 HP @ 2700 RPM with full throttle in MSL
Max. Continuous Power	185 HP @ 2500 RPM with full throttle in MSL

2.4.1 FUEL

Minimum grade aviation gasoline : 100/100LL; for alternate fuelgrades see latest revision of Lyc. S.I. No. 1070
Total fuel capacity: 122 litres (32.1 US Gallon)
Usable fuel capacity: 117 litres (30.8 US Gallon)

For acrobatic flight wing tanks must be empty.
Total fuel capacity for acrobatic 36 litres (9.5 US Gallon) in acro tank.
Usable fuel capacity for acrobatic 32 litres (8.5 US Gallon) in acro tank.

2.4.2 ENGINE LIMITATIONS

a) Power in MSL

-Max. Takeoff Power: 200 HP @ 2700 RPM with full throttle
-Max. Continuous Power: 185 HP @ 2500 RPM with full throttle

b) Oil-temperature gauge

-Max	118°C	245°F

c) Oil capacity

-Maximum sump capacity:		8 qts.
-Minimum sump capacity:	-Acrobatic:	6 qts.
	-Normal:	4 qts.

d) Oil pressure

-Minimum Idling	172 kPa	25 Psi
-Normal	414 - 621 kPa	60 - 90 Psi
-Starting, Warm up, Taxi and Takeoff	379 - 689 kPa	55-100 Psi

CAUTION

It is normal for the oil pressure to "flicker" from 10 to 30 psi (69 to 207 kPa) when going from upright to inverted flight. During knife edge flights and zero-g flights oil pressure may drop and the oil system may not scavenge resulting in engine failure or damage if flight is prolonged. Knife edge and zero-g flight should not exceed 10 seconds.

WARNING

If oil pressure drops to 0 psi the propeller pitch changes automatically to coarse (high) pitch with a corresponding decrease in RPM. Apply positive g to avoid engine stoppage.

e) Fuel pressure

- Max.	83 kPa	12 Psi

f) Cylinder head temperature

- Max	260°C	500°F

2.5 PROPELLER

MT-Propeller Entwicklung GmbH, Federal Republic of Germany

Type: MTV-12-B-C/C183-17e

Max. Takeoff RPM:	2700
Max. Continuous RPM:	2500

2.6 WEIGHT LIMITS

Max. allowed **empty** weight:	
-Normal category	646 kg (1424lbs)
-Acrobatic category (1 seat)	591 kg (1302lbs)
(2 seats)	606 kg (1335lbs)
Max. allowed **T/O** weight:	
-Normal category	840 kg (1858 lbs)
-Acrobatic category (1 seat)	700 kg (1549 lbs)
(2 seats)	800 kg (1770 lbs)
Max. allowed **landing** weight:	840 kg (1858 lbs)

2.7 WEIGHT AND C.G. ENVELOPE

Vertical reference = firewall.
Horizontal reference = upper longerons in cockpit.

2.7.1 NORMAL FLIGHT

Max T/O Weight:	forward C.G.	rear C.G.
840 kg (1858 lbs) (and below)	73,2 cm (28,8")	89.1 cm (35.0")

2.7.2 ACROBATIC FLIGHT (1 SEAT)

Max T/O Weight:	forward C.G.	rear C.G.
700 kg (1549 lbs) (and below)	73,2 cm (28,8")	89.1 cm (35.0")

2.7.3 ACROBATIC FLIGHT (2 SEATS)

Max T/O Weight:	forward C.G.	rear C.G.
800 kg (1770 lbs) (and below)	73,2 cm (28,8")	89.1 cm (35.0")

2.8 ACROBATIC MANEUVERS

2.8.1 NORMAL FLIGHT

All acrobatic maneuvers are prohibited except stall, chandelle, lazy eight and turns up to 60 degrees bank angle.

2.8.2 ACROBATIC FLIGHT

The plane is designed for unlimited acrobatics within the approved operating limitations. Inverted flight manoeuvers are limited to max 4 min. Recommended basic maneuver entry speeds are listed in the following list.

NOTE

If acrobatic maneuvers will be performed with copilot or passenger, the pilot has to check and attend the physiological capability before and during acrobatic maneuvers due to the high possible g-loads.

Check weight and C.G.position !

Maneuvers	Recommended entry speeds IAS min knots (km/h)	max knots (km/h)	Symbol	Remarks
Segment:				
horizontal Line	V_S	V_{NE}		
45°climbing	80 (148)	V_{NE}		
90° up	V_A	V_{NE}		
45° diving	V_S	$< V_{NE}$		reduce throttle
90° diving	V_S	$< V_{NE}$		reduce throttle
1/4 Loop climb.	100 (185)	185 (343)		
Looping	100 (185)	185 (343)		
Stall turn	100 (185)	185 (343)		
Aileron roll	80 (148)	V_A		full deflection
Snap roll	80 (148)	135 (250)		
"tail slide"	100 (185)	185 (343)		
Spin	V_S			
Inverted Spin	V_S			
Knife edge	>150 (278)			< 10 s
Inverted Flight	$>V_S$	185 (343)		< 4 min

CAUTION

Particular caution must be exercised when performing maneuvers at speeds above V_A [154 KIAS (285 km/h)]. Large or abrupt control inputs above this speed may impose unacceptably high loads which exceed the structural capability of the aircraft.

NOTE

For Acrobatic Maneuvers see Section 4. All maneuvers can be performed in upright and inverted flight attitude.

2.9 LOAD FACTOR

2.9.1 NORMAL FLIGHT

+ 6 g / - 3 g for MTOW: 840 kg (1852 lbs)

2.9.2 ACROBATIC FLIGHT

+ 10 g / - 10 g for 1 seat occupied (MTOW 700 kg / 1543 lbs)
+ 8 g / - 8 g for 2 seat occupied (MTOW 800 kg / 1763 lbs)

2.10 FLIGHT CREW LIMITS

Minimum crew is one pilot in the rear seat. 2 persons in both categories (Normal and Acrobatic). Pilot in command seat is the rear seat, Copilot or passenger seat is the front seat. Noise optimized headsets are required.

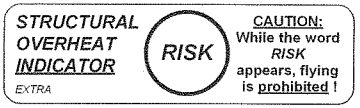
2.11 KINDS OF OPERATIONAL LIMITS

Only **VFR flights at day** are allowed. The A/C engine may be started on ground at OAT from -20°C (-4°F) to +44°C (+111°F). Below temperatures of -10°C (+14°F) the oil vent line must be modified by the low temperature kit (breather line). Flight in known icing-conditions is prohibited. Smoking is prohibited.

2.11.1 STRUCTURAL TEMPERATURE/COLOUR LIMITATION

Structure is qualified up to 72°C (161.6°F). Structure temperatures (composite) above 72°C (161.6°F) are not permitted. Not to exceed this temperature limit, colour specification for composite structure (manufacturer document EA-03205.19) has to be complied with.

To check the temperature inside the cockpit (potential "green house" effect) a reversible temperature indicator (*STRUCTURAL OVERHEAT INDICATOR*) is applied on the root rib of the right wing. After reaching the temperature limit of 72°C (161,6°F) the word "*RISK*" appears and flying is prohibited.



2.12 MAXIMUM OPERATING ALTITUDE

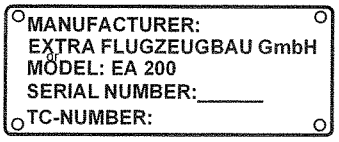
Max. certified operating altitude is 16000 ft MSL (4877 m)

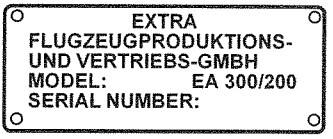
2.13 TIRE PRESSURE

The tire pressure is 2,6 bar (37,7 Psi).

2.14 MARKINGS AND PLACARDS

2.14.1 AIRCRAFT IDENTITY PLACARD







2.14.2 OPERATING PLACARDS

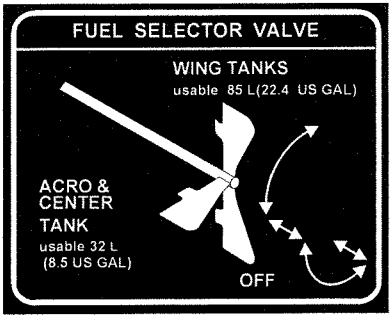
V_A = 154 KIAS (Acro)
V_A = 138 KIAS (Normal)
or
V_A = 285 km/h (Acro)
V_A = 256 km/h (Normal)
— (near airspeed indicator)

THE MARKINGS AND PLACARDS INSTALLED IN THIS AIRPLANE CONTAIN OPERATING LIMITATIONS WHICH MUST BE COMPLIED WITH WHEN OPERATING THIS AIRPLANE IN THE ACROBATIC CATEGORY. OTHER LIMITATIONS WHICH MUST BE COMPLIED WITH WHEN OPERATING THIS AIRPLANE IN THIS CATEGORY OR IN THE NORMAL CATEGORY ARE CONTAINED IN THE AIRPLANE FLIGHT MANUAL. APPLICABLE RPM LIMITATION MUST BE OBSERVED.
— (in the rear cockpit)

THIS AIRPLANE IS CERTIFICATED FOR VFR, DAY OPERATION. OPERATION IN KNOWN ICING CONDITIONS IS PROHIBITED.
— (on the rear instrument panel)

FUEL
AVGAS 100/100 LL
— (near each filler cap)

OIL
— (on the separate hatch / upper cowling)

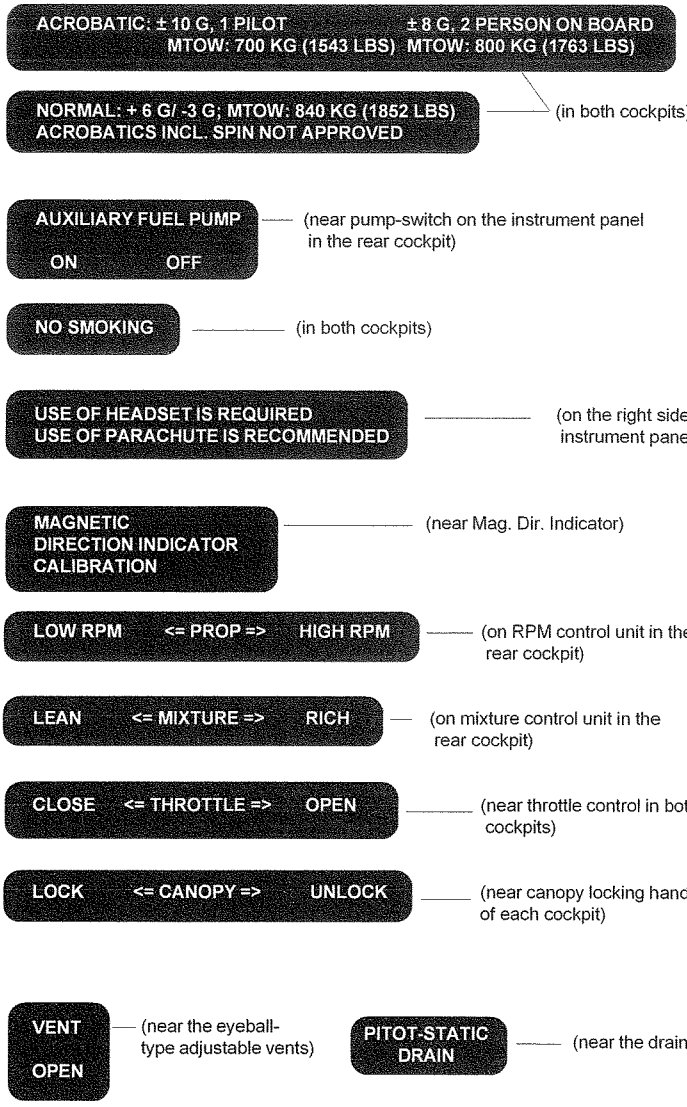


— (in both cockpits near selector valve handle)

NOSE DOWN <= NEUTRAL=> NOSE UP
TRIM
— (near the handle at the right side in the rear cockpit)

WING TANK
MUST BE EMPTY FOR ACROBATICS
(on the rear instrument panel under the fuel capacity indicators)

ACRO & CENTER TANK
WHEN THE INDICATOR READS *"ZERO"* IN LEVEL FLIGHT THE REMAINING USABLE FUEL IS 8 LITERS (2 US GAL.)


ACROBATIC: ± 10 G, 1 PILOT MTOW: 700 KG (1543 LBS)
± 8 G, 2 PERSON ON BOARD MTOW: 800 KG (1763 LBS)
NORMAL: + 6 G/ -3 G; MTOW: 840 KG (1852 LBS)
ACROBATICS INCL. SPIN NOT APPROVED
(in both cockpits)
AUXILIARY FUEL PUMP
ON OFF
(near pump-switch on the instrument panel in the rear cockpit)
NO SMOKING
(in both cockpits)
USE OF HEADSET IS REQUIRED
USE OF PARACHUTE IS RECOMMENDED
(on the right side of both instrument panels)
MAGNETIC DIRECTION INDICATOR CALIBRATION
(near Mag. Dir. Indicator)
LOW RPM <= PROP => HIGH RPM
(on RPM control unit in the rear cockpit)
LEAN <= MIXTURE => RICH
(on mixture control unit in the rear cockpit)
CLOSE <= THROTTLE => OPEN
(near throttle control in both cockpits)
LOCK <= CANOPY => UNLOCK
(near canopy locking handles of each cockpit)
VENT
OPEN
(near the eyeball-type adjustable vents)
PITOT-STATIC DRAIN
(near the drain-point)

APPROVED ACROBATIC MANEUVERS AND RECOMMENDED ENTRY AIRSPEEDS

MANEUVERS	SPEEDS min KIAS	SPEEDS max KIAS
Segment: horizontal Line	V_S	V_{NE}
45°climbing	80	V_{NE}
90° up	V_A	V_{NE}
45° diving	V_S	$<V_{NE}$
90° diving	V_S	$<V_{NE}$
1/4 Loop climb.	100	185
Loop	100	185
Stall turn	100	185
Aileron roll	80	V_A
Snap roll	80	138
"Tail-slide"	100	185
Spin	V_S	
Inverted spin	V_S	
Inverted flight Less than 4 min	$> V_S$	185
Knife edge Less than 10 s	>150	

(in both cockpits)
(alternative SI units next page)

CAUTION
Particular caution must be exercised when performing maneuvers at speeds above V_A (154 KIAS). Large or abrupt control inputs above this speed may impose unacceptably high loads which exceed the structural

WARNING:
SOLO FLYING FROM REAR SEAT ONLY!

(on front instrument panel)

APPROVED ACROBATIC MANEUVERS AND RECOMMENDED ENTRY AIRSPEEDS

MANEUVERS	AIRSPEEDS min km/h	AIRSPEEDS max km/h
Segment: horizontal Line	V_S	V_{NE}
45°climbing	148	V_{NE}
90° up	V_A	V_{NE}
45° diving	V_S	$<V_{NE}$
90° diving	V_S	$<V_{NE}$
1/4 Loop climb.	185	343
Loop	185	343
Stall turn	185	343
Aileron roll	148	V_A
Snap roll	148	256
"Tail-slide"	185	343
Spin	V_S	
Inverted spin	V_S	
Inverted flight Less than 4 min	$> V_S$	343
Knife edge Less than 10 s	>278	

(in both cockpits)
(alternative US units previous page)

CAUTION

Particular caution must be exercised when performing maneuvers at speeds above V_A (285 km/h). Large or abrupt control inputs above this speed may impose unacceptably high loads which exceed the structural

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2.14.3 INSTRUMENT MARKINGS

AIRSPEED INDICATOR

green arc	60 KIAS (111 km/h) - 154 KIAS (285 km/h)
yellow arc	154 KIAS (285 km/h) - 217 KIAS (402 km/h)
red line	217 KIAS (402 km/h)

OIL PRESSURE INDICATOR

red line	25 Psi (172 kPa)
yellow arc	25 Psi (172 kPa) - 55 Psi (379 kPa)
green arc	55 Psi (379 kPa) - 90 Psi (621 kPa)
yellow arc	90 Psi (621 kPa) - 100Psi (689 kPa)
red line	100 Psi (689 kPa)

OIL TEMPERATURE INDICATOR

yellow arc	<140°F
green arc	140°F - 210°F
yellow arc	210°F - 245°F
red line	245°F

CYLINDERHEAD TEMPERATURE INDICATOR

yellow arc	<150°F
green arc	150°F - 400°F
yellow arc	400°F - 500°F
red line	500°F

RPM INDICATOR

green arc	700 - 2500 RPM
yellow arc	2500 - 2700 RPM
red line	2700 RPM

G - METER

green arc	- 5 g - + 8 g
yellow arc	+ 8 g - + 10 g
red line	+ 10 g

FUEL PRESSURE INDICATOR

red line	0 PSI
green arc	0 PSI - 12 PSI
red line	12 PSI

FUEL CAPACITY INDICATOR "WING TANK"

red line	0

FUEL CAPACITY INDICATOR "ACRO & CENTER TANK"

yellow line	0
red line *	0

NOTE

When the Fuel Capacity Indicator "ACRO & CENTER TANK" reads "ZERO" in level flight the remaining usable fuel is 8 Liters (2 US GAL.)

* For the US. registred aircrafts only

2.15 KINDS OF OPERATION EQUIPMENT LIST

The aircraft may be operated in day VFR when the appropriate equipment is installed and operable. Flying under icing conditions is prohibited.

The following equipment list identifies the systems and equipment upon which type certification for each kind of operation was predicated. The following systems and items of equipment must be installed and operable for the particular kind of operation indicated.

	NORMAL	ACROBATIC	
		1 seat	2 seats
COMMUNICATION			
1. Transceiver-VHF	1	1	1
ELECTRICAL POWER			
1. Battery	1	1	1
2. Alternator	1	1	1
3. Ammeter	1	1	1
FLIGHT CONTROL SYSTEM			
1. Elevator-trim control	1	1	1
2. Stall warning	1	1	1
FUEL			
1. Boost pump	1	1	1
2. Fuel quantity indicator	2	2	2
3. Manifold pressure	1	1	1
4. Fuel flow indicator	0	0	0
5. Fuel pressure	1	1	1
LIGHT			
1.single strobe light	1	1	1
2. Wing-tip position / strobe light	0	0	0
NAVIGATION			
1. Altimeter	1	1	1
2. Airspeed indicator	1	1	1
3. Mag. direction indicator	1	1	1
4. OAT indicator	0	0	0
5. Vertical speed indicator	0	0	0
6. Turn and bank indicator	0	0	0
7. Artificial horizon	0	0	0
8. Directional gyro	0	0	0
9. Transponder[1]	1	1	1

[1]) In some airspaces Mode S Elementary Surveillance functionality is required

	NORMAL	ACROBATIC	
		1 seat	2 seats
ENGINE CONTROL			
1. RPM indicator	1	1	1
2. Exhaust gas temperature ind. (EGT)	0	0	0
3. Cylinder head temperature ind.	0	0	0
OIL			
1. Oil temperature indicator	1	1	1
2. Oil pressure indicator	1	1	1
FLIGHT CREW EQUIPMENT			
1. Parachute rear	0	*	*
2. Parachute front	0	0	*
3. Seat belt rear	1	1	1
4. Seat belt front	1	0	1
5. Headset rear	1	1	1
6. Headset front	1	0	1

NOTE

The zeros (**0**) used in the above list mean that the equipment and/or system was not required for type certification for that kind of operation.

The asterisks (*) used in the above list mean that latest national aviation regulations must be observed in determining whether the equipment and/or system are required.
According FAR Part 91 „General Operating and Flight Rules" each occupant of an US registered airplane must wear an approved parachute when performing acrobatic maneuvers.
Extra Flugzeugproduktions- und Vertriebs- GmbH considers acrobatics without wearing an approved parachute to be unsafe.

2.16 NOISE LEVEL

The noise level with standard silencer has been established in accordance

a) with FAR 36 Appendix G, as 68,4 dB(A).

b) with ICAO Annex 16, as 73,1 dB(A).

No determination has been made by the LBA for the FAA that the noise levels of this airplane are or should be acceptable or unacceptable for operation at, into, or out, any airport.

SECTION 3

EMERGENCY PROCEDURES

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SECTION 3

EMERGENCY PROCEDURES

3.0 INTRODUCTION

3.0.1 GENERAL

This section contains the checklist and procedures coping with emergencies that may occur. This checklist must be followed in various emergencies to ensure maximum safety for the crew and/or aircraft.
Thorough knowledge of these procedures will enable the aircrew to better cope with an emergency. The steps should be performed in the listed sequence. However the procedures do not restrict the aircrew from taking any additional action necessary to deal with the emergency. The procedures contain items classified as critical or noncritical. The critical items are actions that shall be performed immediately to avoid aggravating the emergency.

3.0.2 GENERAL BEHAVIOUR IN EMERGENCY SITUATIONS

As soon as one of the crew member becomes aware that an emergency situation exists, he must immediately alert the other crew member of the situation. In any emergency situation, contact should be established with a ground station as soon as possible after completing the initial corrective action. Include position, altitude, heading, speed, nature of the emergency and pilot's intentions in the first transmission. There after the ground station should be kept informed of the progress of the flight and of any changes or developments in the emergency. Three basic rules apply to most emergencies and should be observed by each aircrew member:

1. Maintain aircraft control
2. Analyse the situation and take proper action
3. Land as soon as possible/as soon as practical

The meaning of "*as soon as possible*" and "*as soon as practical*" as used in this section is as follows:

Land AS SOON AS POSSIBLE (ASAP) =	Emergency conditions are urgent and require an immediate landing at the nearest suitable airfield, considering also other factors, such as weather conditions and aircraft mass.
Land AS SOON AS PRACTICAL=	Emergency conditions are less urgent and in the aircrews judgement the flight may be safely continued to an airfield where more adequate facilities are available.

3.1 AIRSPEEDS FOR EMERGENCY OPERATION

If nothing is mentioned a weigh of 840 kg (1852 lbs) is valid.

Stall speed	59 KIAS (109 km/h)
Engine failure after takeoff	80 KIAS (148 km/h)
Best recommended gliding speed (glide angle 1 : 6,2)	
-Normal	90 KIAS (167km/h)
-Acro (800 kg)	80 KIAS (148 km/h)
Precautionary landing with engine power	80 KIAS (148 km/h)
Landing without engine power	80 KIAS (148 km/h)
Maximum demonstrated cross wind component	15 Knots (27 km/h)

3.2 OPERATIONAL CHECKLIST

3.2.1 ENGINE FAILURE DURING TAKEOFF ROLL

1. Throttle	IDLE
2. Brakes	APPLY
3. Mixture	IDLE CUT OFF
4. Ignition switch	OFF
5. Master switch	OFF

3.2.2 ENGINE FAILURE IMMEDIATELY AFTER TAKEOFF

Stall speed: 59 KIAS (109 km/h)

1. Airspeed	80 KIAS (148 km/h)
2. Mixture	IDLE CUT OFF
3. Fuel shutoff valve	OFF (Pull & Turn)
4. Ignition switch	OFF
5. Master switch	OFF

WARNING

The stall warning is deactivated!

6. Forced landing	PERFORM as practical

3.2.3 ENGINE FAILURE DURING FLIGHT (RESTART PROCESS)

1. Airspeed	80 KIAS (148 km/h)
2. Fuel shutoff valve	"ACRO & CENTER" tank
3. Mixture	RICH
4. Boost pump	ON
5. Ignition switch	BOTH (or START if propeller has stopped)

3.2.4 OIL SYSTEM MALFUNCTION

If oil pressure indicates low:	Apply positive "g"
If oil pressure is not regained than:	
1. Airspeed	80 KIAS (148 km/h)
2. Throttle	REDUCE TO IDLE
3. Engine oil temperature	OBSERVE INDICATION
4. Land	ASAP

WARNING

If oil pressure drops to 0 psi the propeller pitch changes automatically to coarse (high) pitch with a corresponding decrease in RPM.

3.2.5 ALTERNATOR FAILURE

An alternator failure is indicated by:
- the red light of the low voltage monitor (if installed) or
- the ampermeter showing negativ current

1. Alternator	SWITCH OFF AND ON
2. Low voltage monitor / ampermeter	CHECK INDICATION
3. Red light off / positive current	CONTINUE FLIGHT
If red light illuminates again / negativ current :	
4. Land	AS SOON AS PRACTICAL

3.3 FORCED LANDINGS

3.3.1 EMERGENCY LANDING WITHOUT ENGINE POWER

1. Seat belts, shoulder harnesses	SECURE
2. Airspeed	80 KIAS (148 km/h)
3. Mixture	IDLE CUT OFF
4. Fuel shutoff valve	OFF (Pull & Turn)
5. Ignition switch	OFF
6. Master switch	OFF

WARNING

The stall warning is deactivated!

7. Touchdown	3-POINT LANDING
8. Brakes	AS REQUIRED

3.3.2 PRECAUTIONARY LANDING WITH ENGINE POWER

1. Seat belts, shoulder harnesses	SECURE
2. Airspeed	80 KIAS (148 km/h)
3. Selected field	FLY OVER, noting terrain and obstructions, then reaching a safe altitude and airspeed

4. Master switch	OFF

WARNING

The stall warning is deactivated!

5. Touchdown	3-POINT LANDING
6. Ignition switch	OFF
7. Mixture	IDLE CUT OFF
8. Fuel shutoff valve	OFF (Pull & Turn)
9. Brakes	AS REQUIRED

3.4 FIRES

3.4.1 DURING START ON GROUND

1. Cranking	CONTINUE to get a start which would suck the flames and accumulated fuel through the air inlet and into the engine.
2. Fuel shutoff valve	OFF (Pull & Turn)
3. Power	1700 RPM for one minute.
4. Engine	SHUT DOWN
5. After engine stop	ABANDON aircraft and inspect for damage
6. Fire	EXTINGUISH using fire extinguisher if available

WARNING

Do not open engine compartment access doors while engine is on fire

3.4.2 IF ENGINE FAILS TO START

1. Cranking	CONTINUE
2. Throttle	FULL OPEN
3. Mixture	IDLE CUT OFF
4. Fuel shutoff valve	OFF (Pull & Turn)

If fire is extinguished

5. Master switch	OFF
6. Ignition switch	OFF
7. Engine compartment	INSPECT

3.4.3 ENGINE FIRE IN FLIGHT

1. Mixture	IDLE CUT OFF
2. Fuel shutoff valve	OFF (Pull & Turn)
3. Master switch	OFF

WARNING

The stall warning is deactivated!

4. Airspeed	100 KIAS, find your airspeed/attitude will keep the fire away from the cockpit

5. Land as soon as possible

3.5 ICING

3.5.1 INADVERTENT ICING ENCOUNTER

1. Turn back or change altitude to obtain an outside temperature that is less conductive to icing.
2. Plan a landing at the nearest airfield. With extremely rapid ice build-up select a suitable "off airport" landing field.

3.6 UNINTENTIONAL SPIN

Refer to section 4 (Normal Procedures) acrobatic maneuvers, spin recovery.

WARNING

The engine may stop, during unintentional spin with fuel shutoff valve in "Wingtank" position. After immediate spin recovery start the engine according section 3.2.3 "ENGINE FAILURE DURING FLIGHT".

3.7 MANUAL BAIL-OUT

When in an emergency situation that requires abandoning the aircraft and while wearing a parachute, which is at least strongly recommended for acrobatics:

- Inform your passenger
- Reduce speed to 100 KIAS if possible
- Pull mixture to lean
- Open canopy (the lowpressure over the canopy in normal flight will flip the canopy full open immediately)
- Take off headset
- Open seat belt
- Leave airplane
- Try to avoid wing and tail
- Open parachute

3.8 EMERGENCY EXIT AFTER TURN OVER

1. Master switch	OFF
2. Fuel shutoff valve	OFF (Pull & Turn)
3. Seat belts	OPEN
4. Parachute harnesses	OPEN
5. Canopy handle	PULL TO OPEN

NOTE

If canopy fails to open break the canopy.

6. Aircraft	EVACUATE ASAP

3.9 ELEVATOR CONTROL FAILURE

In case of elevator control failure the aircraft can be flown with the elevator trim. In this case trim nose up to the desired speed and control horizontal flight or descend with engine power. For landing trim nose up and establish a shallow descend by adjusting throttle. To flair the plane gently increase power to bring the nose up to landing attitude.

SECTION 4

NORMAL PROCEDURES

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SECTION 4

NORMAL PROCEDURE

4.0 GENERAL

4.0.1 AIRSPEEDS FOR NORMAL OPERATION

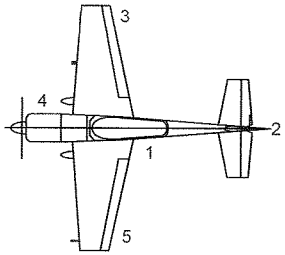
CATEGORY	ACRO		NORMAL
	1 seat KIAS (km/h)	2 seats KIAS (km/h)	KIAS (km/h)
Takeoff:			
-Rotating Speed	70 (130)	72 (133)	73 (135)
Climb:			
-Vx	72 (133)	74 (137)	75 (139)
-Vy	84 (156)	85 (157)	86 (159)
-Recommended Normal Climb Speed	95 (176)	100 (185)	105 (194)
-Max. Cruise	154 (285)	154 (285)	154 (285)
Landing:			
-Approach	80 (148)	85 (157)	90 (167)
-on Final	70 (130)	72 (133)	73 (135)
-Go-Around Speed	90 (167)	95 (176)	100 (185)
Recommended Airspeed For Flight In Rough Air (maximum)	130 (241)	130 (241)	130 (241)
Max. Demonstrated Cross Wind Component	15 kts (27)	15 kts (27)	15 kts (27)

4.0.2 CHECKLIST AND PROCEDURES

This handbook contains the checklist and procedures to operate the aircraft in normal and acrobatic operation. The pilot should be familiar with all procedures contained in this Pilot's Operating Handbook, which should be carried on board. The pilot has to comply with Check-list for daily check and inspections (see Section 8, Handling, Servicing and Maintenance).

4.1 PREFLIGHT INSPECTION

4.1.1 EXTERIOR INSPECTION ILLUSTRATION



4.1.2 GENERAL

Visually check airplane for general condition during walk around inspection. Perform exterior check as outlined in the picture above in counterclockwise direction.

4.2 CHECKLIST PROCEDURES

1) Cockpit

1. Pilot's Operating Handbook	(AVAILABLE)
2. Airplane weight and balance	CHECKED
3. Ignition switch	OFF
4. Master switch	ON
5. Fuel quantity indicators	CHECK
6. Master switch	OFF
7. Fuel selector *	"ACRO & CENTER"

NOTE *

Although safe operation did not require the use of the tanks in a specific sequence, it is recommended to set fuel selector to "ACRO & CENTER" Tank position!

8. Pitot-Static-Drain	CHECK FOR WATER ACCUMULATIONS

NOTE

If water exists, drain by removing the plug. Reinstall plug.

9. "Acro & Center" Tank	CHECK FOR LEAKAGE

NOTE

Pay attention for fuel accumulations between alu-tank and GRP-cover. (This is indicated by a blue color)

2) Empennage

1. All round inspection, canopy, surfaces, stabilizer, elevator, trim rudder and tailwheel	CHECK
2. Horizontal stabilizer attachment bolts	CHECK FOR FREEPLAY BY MOVING THE TIP OF THE HORIZ. STABILIZER UP- AND DOWNWARDS

3) Right Wing

1. Aileron, freedom of movement and security	CHECK
2. Trailing edge	CHECK
3. Fuel tank vent opening (right landing gear)	CHECK
4. Fuel quantity	CHECK
5. Fuel tank filler cap	CHECK
6. Right landing gear, wheel and brake	CHECK
7. Stall warning vane	CHECK

4) Nose

1. Engine oil dipstick	CHECK
2. Propeller and spinner	CHECK
3. Air inlet	CHECK
4. Fuel quantity ("Acro & Center" tank)	CHECK
5. Fuel tank filler cap	CHECK
6. "Acro & Center" tank drain	DRAIN FOR AT LEAST 4 SECONDS TO CLEAR SUMP OF POSSIBLE WATER; CHECK CLOSED
7. "Wing tank" drain	DRAIN FOR AT LEAST 4 SECONDS TO CLEAR SUMP OF POSSIBLE WATER; CHECK CLOSED
8. Fuel filter drain	DRAIN FOR AT LEAST 4 SECONDS TO CLEAR FILTER OF POSSIBLE WATER; CHECK CLOSED

5) Left wing

1. Left landing gear, wheel and brakes	CHECK
2. Fuel quantity	CHECK
3. Fuel tank filler cap	CHECK
4. Pitot cover	REMOVE
5. Trailing edge	CHECK
6. Aileron, freedom of movement and security	CHECK

6) Before starting engine

1. Preflight inspection	COMPLETE
2. Passenger briefing	COMPLETE
3. Parachute handling briefing	COMPLETE
4. Seats, seatbelts, shoulder harnesses	ADJUST AND LOCK
5. Canopy	CLOSE AND LOCK
6. Brake	CHECK
7. Master switch	ON
8. Avionics power switch	OFF
9. Electrical equipment	OFF
10. Alternator	ON
11. Wingtip position / Strobe lights	ON

4.3 STARTING PROCEDURES

4.3.1 COLD ENGINES

The following starting procedures are recommended, however, the starting conditions may necessitate some variation from these procedures.

1. Perform preflight inspection.

2. Set propeller governor control to "High RPM" position.

3. Open throttle approximately 1/4 travel.

4. Turn boost pump "ON".

5. Move mixture control to "FULL RICH" until a slight but steady fuel flow is noted (approximately 3 to 5 seconds) and return mixture control to "IDLE CUT-OFF".
   Turn boost pump "OFF".

6. Engage starter.

7. When engine fires release the ignition switch back to "BOTH".

8. Move mixture control slowly and smoothly to "FULL RICH".

9. Check the oil pressure gauge. If minimum oil pressure is not indicated within 30 seconds, shut off the engine and determine trouble.

4.3.2 HOT ENGINES

Because of the fact that the fuel percolates and the system must be cleared of vapour, it is recommended to use the same procedure as outlined for cold engine start.

4.4 TAXIING THE AIRCRAFT

1. Canopy	CLOSE AND LOCK
2. Brake	CHECK
3. Altimeter	Set on QFE or QNH Scale error max. +60 ft.
4. Avionic master switch	ON
5. Electrical equipment	ON
6. Radio	Set and test
7. Mixture	Leave in "FULL RICH" position

Operate only with the propeller in minimum blade angle (High RPM).
Warm-up at approximately 1000-1200 RPM. The engine is ready for takeoff when the throttle can be opened without the engine faltering.

4.5 TAKE-OFF PROCEDURE

4.5.1 BEFORE TAKE-OFF

Before you line up at the runway for takeoff:

- Check oil pressure and oil temperature.
- Check the magnetos at 1800 RPM. Allowed drop is 175 RPM (max. difference 50 RPM).
- Check Alternator Output.
- Move also the propeller control through its complete range to check operation and return to full "HIGH RPM" position.
  Turn boost pump "ON" (check indicator movement on the fuel pressure gauge).

NOTE

The RPM Gauge is electronically operated. To check the magnetos the RPM source switch must be set to the same magneto as the ignition switch. Otherwise the gauge will show zero.

4.5.2 TAKE-OFF

Set throttle smoothly to max and let the airspeed go up to 70-75 KIAS (130-139 km/h). A light pressure on the stick lifts the tail to horizontal position. Rotate the aircraft at 75 KIAS (139 km/h). On reaching climb speed of 100 KIAS (185 km/h) reduce the RPM to 2500 and proceed climbing.

4.6 CLIMB

The maximum continuous RPM is restricted to 2500.

Turn boost pump "OFF".

4.7 CRUISE

1. Altitude	- As selected
2. Throttle / RPM	- Adjust for cruising speed
3. Mixture	- Adjust for cruising speed
4. Trim	- As required
5. Fuel	- Check periodically

4.8 LANDING PROCEDURES

4.8.1 DESCENT

1. Throttle	- Reduce
2. Mixture	- "FULL RICH"
3. Trim	- Adjust
4. Fuel selector valve*	- "ACRO & CENTER" tank

NOTE

Although safe operation did not require the use of the tanks in a specific sequence, it is recommended to set fuel selector to "ACRO & CENTER" tank position!

4.8.2 APPROACH

1. Boost pump	- ON
2. Mixture	- Set to "RICH"
3. Airspeed	- Reduce to approach speed
4. Propeller	- Set to low pitch ("HIGH RPM")

NOTE

It is recommended to reduce the RPM during approach and landing in order to avoid unnecessary noise.
In case of "Go Around", RPM control must be set to max. RPM before applying power.

4.8.3 BEFORE LANDING

1. Landing approach	- Proceed
2. Airspeed on final	- Maintain 75 KIAS (139 km/h)
3. Elevator trim	- Adjust

NOTE

Stall speed will be

MTOW = 700 kg :	53 KIAS (98 km/h)
MTOW = 800 kg :	58 KIAS (107 km/h)
MTOW = 840 kg :	59 KIAS (109 km/h)

4.8.4 NORMAL LANDING

1. Landing	- Perform as practicable with respect to surface and weather condition
2. Touchdown	- 3 point landing

NOTE

The rudder is effective down to 30 KIAS (56 km/h) airspeed

3. Throttle	- CLOSE / IDLE
4. Braking	- Minimum required

4.9 GO-AROUND

Decide early in the approach if it is necessary to go around and then start go-around before too low altitude and airspeed are reached.

Proceed as follows:

1. RPM control	- "HIGH RPM" / Full forward
2. Throttle	- "OPEN" / TAKEOFF power
3. Airspeed	- Minimum 90 KIAS (167 km/h) Rotate to go-around attitude

4.10 SHUTDOWN

1. Boost pump	- "OFF"
2. Engine	- Run for 1 min. at 1000 RPM
3. Dead cut check	- Perform
4. Avionic master switch	- "OFF" (if installed)
5. Mixture	- "IDLE CUT OFF"
6. Ignition switch	- "OFF"
7. Master switch	- "OFF"

4.11 LEAVING THE AIRCRAFT

1. Canopy	- Close and lock
2. Aircraft	- Secure
3. Pitot cover	- Attach
4. Log book	- Comply

4.12 ACROBATIC MANEUVERS

4.12.1 GENERAL

NOTE

Prior to executing these maneuvers tighten harnesses and check all loose items are stowed. Start the maneuvers at safe altitude and max continuous power setting if not otherwise noted.

For maneuver limits refer to Section 2 LIMITATIONS.

After termination of acrobatic maneuvers the artificial horizon (if installed) must be reset if possible.

At high negative g-loads and zero g-periods it is normal that oil pressure and RPM indication might drop down momentarily returning to normal status at positive g-loads.

WARNING

The high permissible load factors of the airplane may exceed the individual physiological limits of pilot or passenger. This fact must be considered when pulling or pushing high g's.

4.12.2 MANEUVERS

CAUTION

Particular caution must be exercised when performing maneuvers at speeds above V_A [154 KIAS (285 km/h)]. Large or abrupt control inputs above this speed may impose unacceptably high loads which exceed the structural capability of the aircraft.

Acrobatics is traditionally understood as maneuvers like loop, humpty bump, hammerhead turn, aileron roll etc..

This manual does not undertake to teach acrobatics, however, it is meant to demonstrate the plane's capabilities.

For this reason maneuvers are divided into segments. The segments are described. Limitations are pointed out.

- Segment horizontal line:
  A horizontal line may be flown with any speed between V_S and V_{NE}.

- Segment "line 45° climbing":
  The plane will follow the line at max. power. The speed will not decrease below 80 KIAS (148 km/h).

- Segment "line 90° up":
  Any entry speed may be used. Out of a horizontal pull-up at 200 KIAS (370 km/h) the vertical penetration will be 2.500 ft. The speed will gradually decrease to 0.

NOTE

In extremely long "line 90° up" an RPM decay may occur. This is related to a loss of oil pressure. Positive g´s should be pulled immediately in order to protect the engine. Oil pressure will return immediately.

- Segment "line 45° diving":
  Throttle must be reduced in order to avoid exceeding V_{NE}.

- Segment "line 90° diving":
  Throttle must be reduced to idle in order to avoid exceeding V_{NE}.

Above segments may be filled up with aileron rolls on snap rolls. Watch V_A = 154 KIAS (285 km/h) for aileron rolls with max. deflection.
Snap rolls should not be performed at speeds above 138 KIAS (256 km/h).

- Segment "1/4 loop, climbing":
  The minimum recommended speed is 100 KIAS (185 km/h). If the maneuver is to be followed by a vertical line, a higher entry speed is required depending on the expected length of the line. A complete loop can be performed at speeds above 100 KIAS (185 km/h).

NOTE

Since the maximum horizontal speed is 185 KIAS (343 km/h), higher speeds should be avoided in acrobatics since an unnecessary loss of altitude would occur.

- Torque maneuvers:
  All maneuvers with high angular velocity associated with high propeller RPM must be considered dangerous for the engine crankshaft.

Although wooden composite propeller blades are used, the gyroscopic forces at the prop flange are extremely high.

CAUTION

If performing a gyroscopic maneuver such as flat spin, power on, or knife edge spin, reduce RPM to 2400 in order to minimize the gyroscopic forces.

4.12.3 SPIN

To enter a spin proceed as follows:

- Reduce speed, power idle
- When the plane stalls:
- Kick rudder to desired spin direction
- Hold ailerons neutral
- Stick back (positive spinning), Stick forward (negative spinning)

The plane will immediately enter a stable spin.

- Ailerons against spin direction will make the spin flatter.
- Ailerons into spin direction will lead to a spiral dive.

Above apply for positive and negative spinning.

To stop the spin:

- Apply opposite rudder
- Make sure, power idle
- Hold ailerons neutral
- Stick to neutral position

The plane will recover within 1/2 turn.
Recovery can still be improved by feeding in in-spin ailerons.

NOTE

If ever disorientation should occur during spins (normal or inverted) one method always works to stop the spin:

- Power idle
- Kick rudder to the heavier side
  (this will always be against spin direction)
- Take hands off the stick

The spin will end after 1/2 turn. The plane will be in a steep dive in a side-slip.
Recovery to normal flight can be performed easily.

NOTE

After six turns of spinning the altitude loss including recovery is 2800 ft.

SECTION 5

PERFORMANCE

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SECTION 5

PERFORMANCE

5.1 GENERAL

Performance data charts on the following pages are presented to facilitate the planning of flights in detail and with reasonable accuracy under various conditions. The data in the charts have been computed from actual flight tests with the aircraft and engine in good condition and using average piloting techniques.

It should be noted that the performance information presented in the range and endurance charts allow for 45 minutes reserve fuel at specified speeds. Some indeterminate variables such as engine and propeller, air turbulence and others may account for variations as high as 10% or more in range and endurance. Therefore, it is important to utilize all available information to estimate the fuel required for the particular flight.

5.1.1 PERFORMANCE CHARTS

Performance data are presented in tabular or graphical form to illustrate the effect of different variables. Sufficiently detailed information are provided in the tables so that conservative values can be selected and used to determine the particular performance figure with reasonable accuracy.

All speeds in this chapter are Indicated Air Speeds (IAS). The performance figures below are given under following conditions.

1. Maximum allowed weight 840 kg (1852 lbs) except otherwise stated
2. Takeoff and landing on concrete surface.
3. No wind.
4. Standard atmospheric condition.

5.1.2 DEFINITIONS OF TERMS

For definition of terms, abbreviations and symbols refer to section 1, General.

5.1.3 SAMPLE PROBLEM

CONDITIONS

Airfield Pressure Altitude: 1500 ft (457 m), Outside air temperature on ground: 20°C
Cruising flight in FL 80 [8000 ft (2438 m)], Outside air temperature in FL 80: 10°C

TAKEOFF

The takeoff distance with T/O weight of 840kg (1852 lbs) is shown by Fig. 5.5

Result: Ground Roll: 257 m
Total Distance to clear a 50 ft (15 m) obstacle: 407 m

RATE OF CLIMB

The rate of climb in FL 80 (2438 m) with an aircraft weight of 750 kg is shown by Fig. 5.6

Result: Rate of climb: 950 ft/min

TIME, FUEL AND DISTANCE TO CLIMB

The values from airfield level 1500 ft (457 m) to FL 80 (2438 m) with an aircraft weight of 800 kg is shown by Fig. 5.7:

Result: Time: (8.7 - 1.9) min = 6.8 min
Distance: (13 - 2.8) NM = 10.2 NM
Fuel consumption: (8.3 - 1.9) liters = 6.4 liters

CRUISING SPEED

The cruising speed in FL 80 (2438 m) with an aircraft weight of 840 kg and power setting of 65% is shown by Fig. 5.9.

Result: Cruising speed : 148 KTAS (274 km/h)

Cruise Altitude and Power Setting should be determined for most economical fuel consumption and several other considerations.

FUEL CONSUMPTION

The fuel consumption in FL 80 (2438 m) with an aircraft weight of 840 kg and power setting of 65% is shown by Tab. 5.12.

Result: Fuel consumption: 34 liters per hour

RANGE

The range in FL 80 (2438 m) with an aircraft weight of 840 kg (1852 lbs) and power setting of 65%, is shown by Tab. 5.12, including fuel for warm up and takeoff from SL, max continuous Power climb to cruising altitude, and a reserve of 19 liters (5 US Gal.) for 45 minutes with 45% Power. 5 liters (1.3 US Gal.) unusable fuel is taken into account.

Total fuel:	122 L	(32 US Gal.)
Warm up & T/O:	-4 L	(-1.05 US Gal.)
Reserve:	-19 L	(-5 US Gal.)
Unusable fuel:	-5 L	(-1.3 US Gal.)
Usable fuel:	94 L	(24 US Gal.)

Result: Range: 386 NM (714 km)

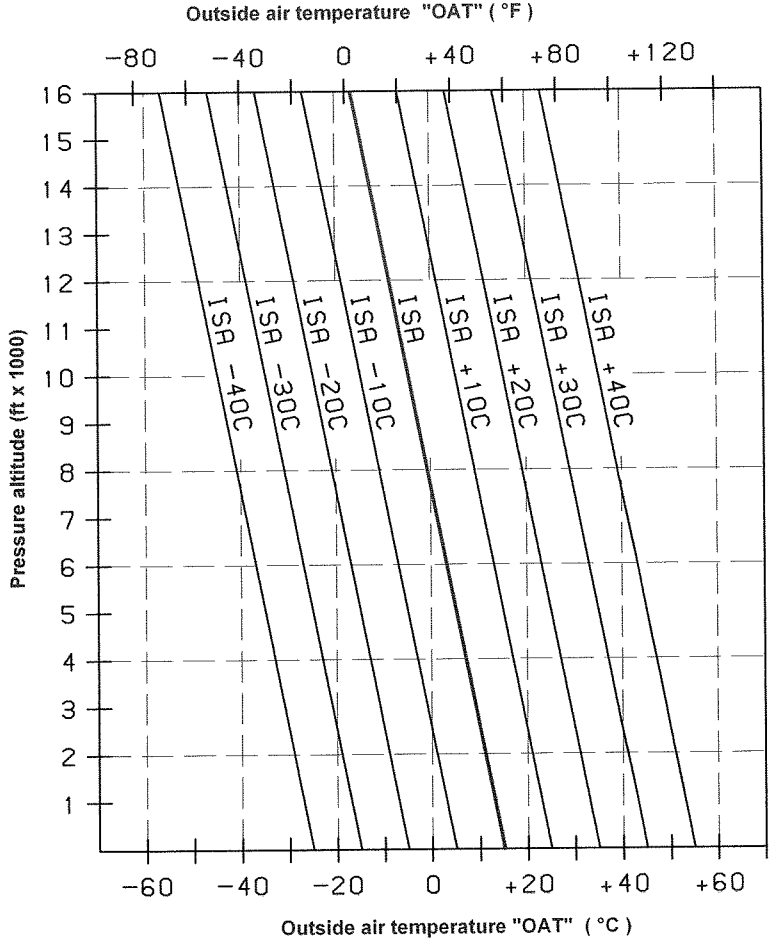
Endurance

The endurance in FL 80 (2438 m) is shown by Tab. 5.10.

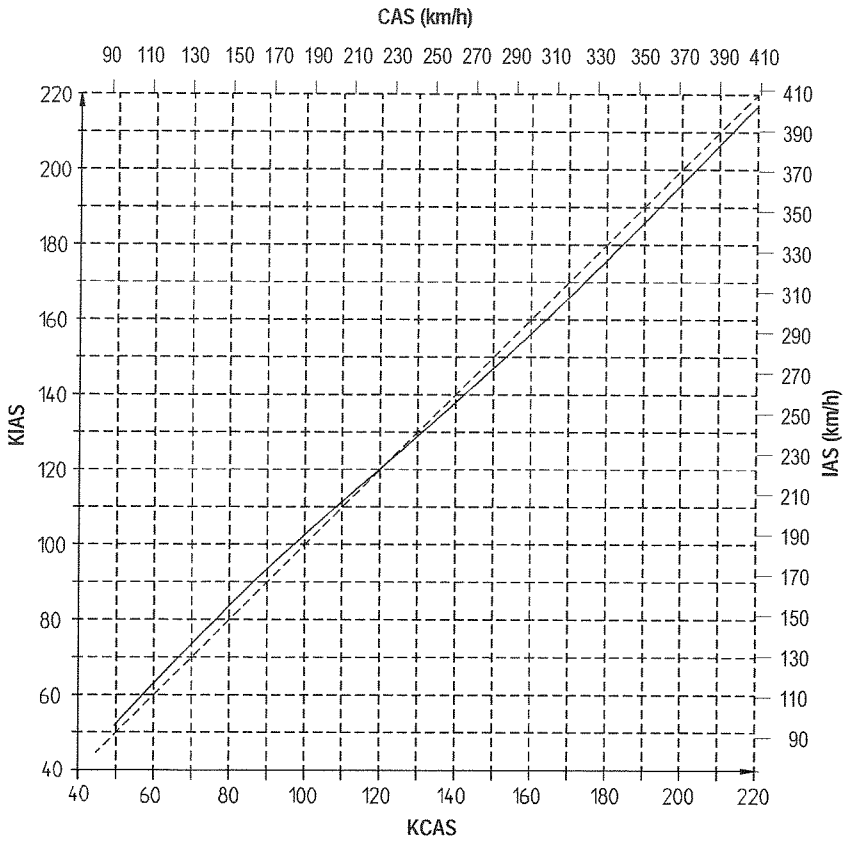
Result: Endurance: 2.66 hours

5.2 ISA CONVERSION

ISA Conversion of pressure altitude and outside air temperature



5.3 AIRSPEED CALIBRATION



NOTE

Indicated airspeed assumes zero instrument error

5.4 STALL SPEED

Condition:
Power Idle
Forward C.G. position

WEIGHT	CATEGORY	ANGLE OF BANK							
		0°		30°		45°		60°	
		KIAS (km/h)	KCAS (km/h)	**KIAS** (km/h)	KCAS (km/h)	**KIAS** (km/h)	KCAS (km/h)	**KIAS** (km/h)	KCAS (km/h)
840 kg (1852 lbs)	**NORMAL**	59 (109)	56 (104)	64 (119)	60 (111)	72 (133)	67 (124)	82 (152)	79 (146)
800 kg (1763 lbs)	**ACRO** (2 seat)	58 (107)	55 (102)	63 (117)	59 (109)	70 (130)	65 (120)	81 (150)	78 (144)
700 kg (1543 lbs)	**ACRO** (1 seat)	53 (98)	51 (94)	58 (107)	55 (102)	65 (120)	61 (113)	75 (139)	72 (133)

Max. altitude loss during stall recovery is approximately 100 ft (30 m).

5.5 TAKEOFF PERFORMANCE

Power: 2700 RPM and full throttle before brake release
Landing runway: paved level dry runway

For every 5 kts (9 km/h) headwind, the takeoff (T/O) distance can be decreased by 5%. For every 2 kts (4 km/h) tailwind [up to 10 kts (19 km/h)], the (T/O) distance is increased by 10%. On a solid, dry and plain Grass Runway, the T/O distance is increased by 15%.

Weight: 700 kg (1543 lbs) Liftoff speed (T/O): 70 KIAS (130 km/h)
Takeoff distance in meter Obstacle clearance speed over 15m (50ft) : 75 KIAS (139 km/h)

	-20°C OAT		-10°C		0°C		10°C		20°C		30°C		40°C	
PA	T/O	50 ft	T/O	50 ft	T/O	50 ft	T/O	50 ft	T/O	50 ft	T/O	50 ft	T/O	50 ft
Sea level	116	183	129	205	144	228	160	253	177	280	195	309	214	340
2000	135	213	151	239	168	266	186	295	206	327	227	360	250	396
4000	157	249	176	279	196	311	218	345	241	382	266	422	293	464
6000	184	292	206	327	230	364	255	405	283	448	312	495	344	545
8000	216	343	242	384	270	429	301	476	333	528	368	583	405	642

Takeoff distance in ft

	-20°C OAT		-10°C		0°C		10°C		20°C		30°C		40°C	
PA	T/O	50 ft	T/O	50 ft	T/O	50 ft	T/O	50 ft	T/O	50 ft	T/O	50 ft	T/O	50 ft
Sea level	381	600	423	673	472	748	525	830	581	919	640	1014	702	1115
2000	443	699	495	784	551	873	610	968	676	1073	745	1181	820	1299
4000	515	817	577	915	643	1020	715	1132	791	1253	873	1385	961	1522
6000	604	958	676	1073	755	1194	837	1329	928	1470	1024	1624	1129	1788
8000	709	1125	794	1260	886	1407	988	1562	1093	1732	1207	1913	1329	2106

Weight: 840 kg (1852 lbs) Liftoff speed (T/O): 73 KIAS (135 km/h)
Takeoff distance in meter Obstacle clearance speed over 15m (50ft) : 78 KIAS (144 km/h)

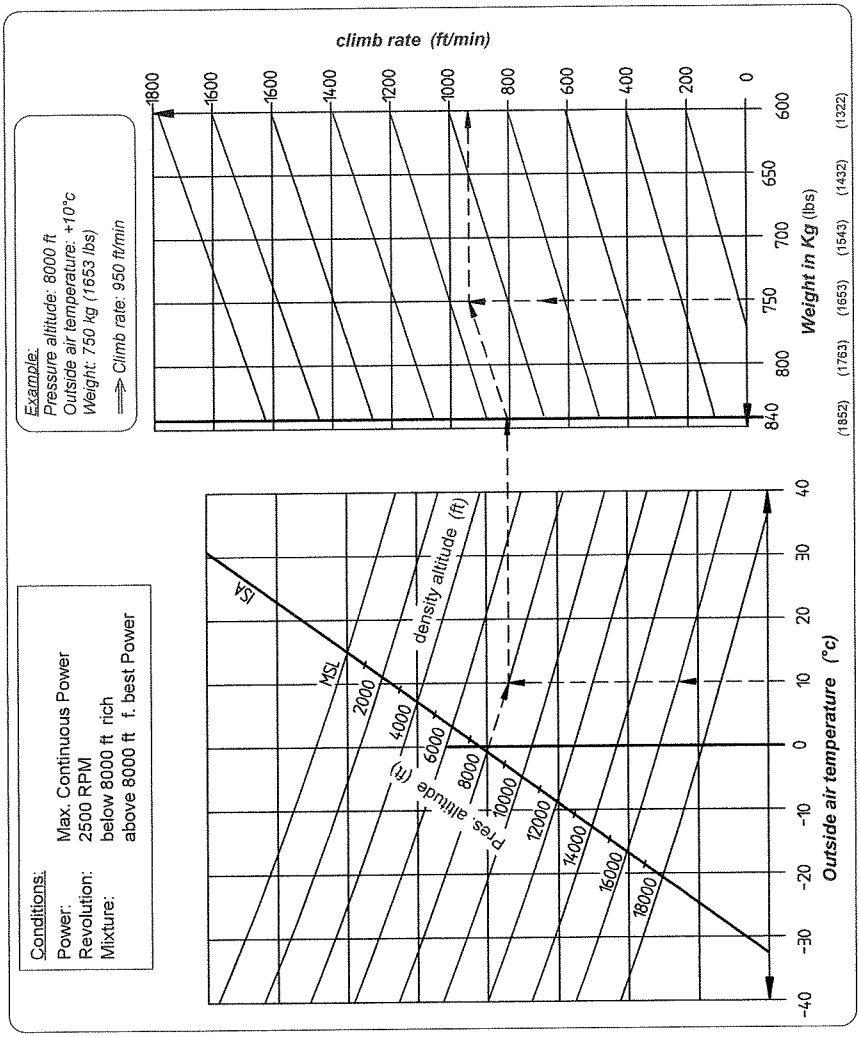
	-20°C OAT		-10°C		0°C		10°C		20°C		30°C		40°C	
PA	T/O	50 ft	T/O	50 ft	T/O	50 ft	T/O	50 ft	T/O	50 ft	T/O	50 ft	T/O	50 ft
Sea level	149	236	167	264	186	295	206	327	228	362	252	399	277	438
2000	174	275	194	308	217	343	240	381	266	422	293	465	323	512
4000	203	321	227	360	253	401	281	446	311	493	343	544	378	599
6000	238	377	238	422	297	470	330	523	365	579	403	639	444	703
8000	279	442	313	496	349	553	388	615	430	681	475	752	523	828

Takeoff distance in ft

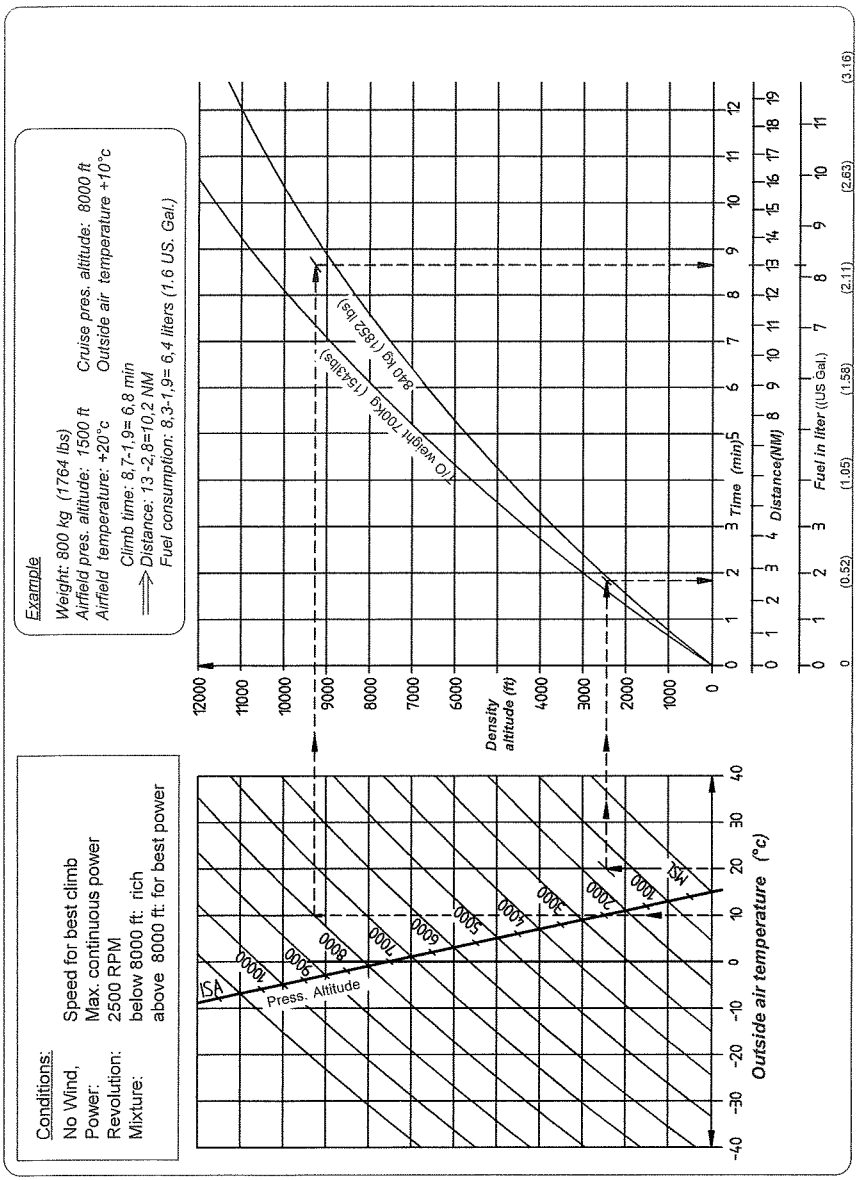
	-20°C OAT		-10°C		0°C		10°C		20°C		30°C		40°C	
PA	roll	50 ft	roll	50 ft	roll	50 ft	roll	50 ft	roll	50 ft	roll	50 ft	roll	50 ft
Sea level	489	774	548	866	610	968	676	1073	748	1188	827	1309	909	1437
2000	571	902	636	1010	712	1125	787	1250	873	1385	961	1526	1060	1680
4000	666	1053	745	1181	830	1316	922	1463	1020	1617	1125	1785	1240	1965
6000	781	1237	781	1385	974	1542	1083	1716	1198	1900	1322	2096	1457	2306
8000	915	1450	1027	1627	1145	1814	1273	2018	1411	2234	1558	2467	1716	2717

5.6 RATE OF CLIMB PERFORMANCE

Weight	Airspeed during climb KIAS (km/h)		
	sea level	5000 ft	12000 ft
840 Kg (1852 lbs)	86 (159)	84 (156)	82 (152)
700 Kg (1543 lbs)	84 (156)	82 (152)	80 (148)

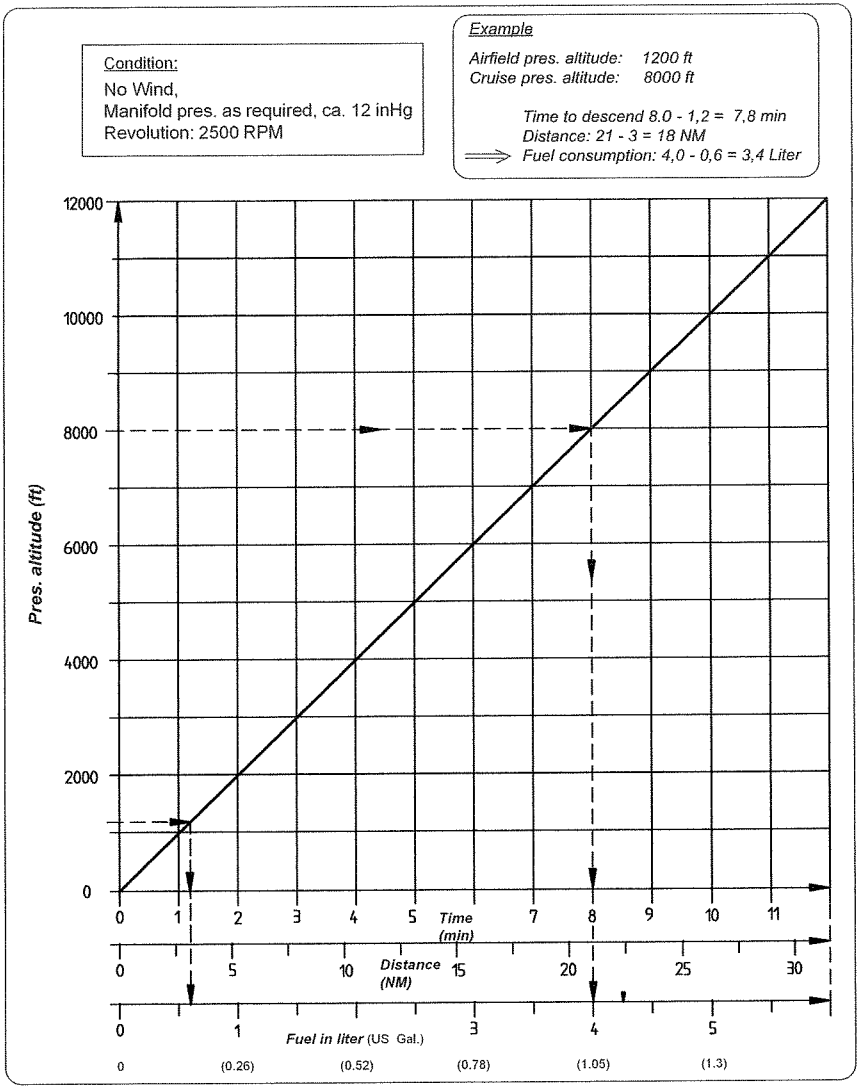


5.7 TIME, FUEL AND DISTANCE TO CLIMB

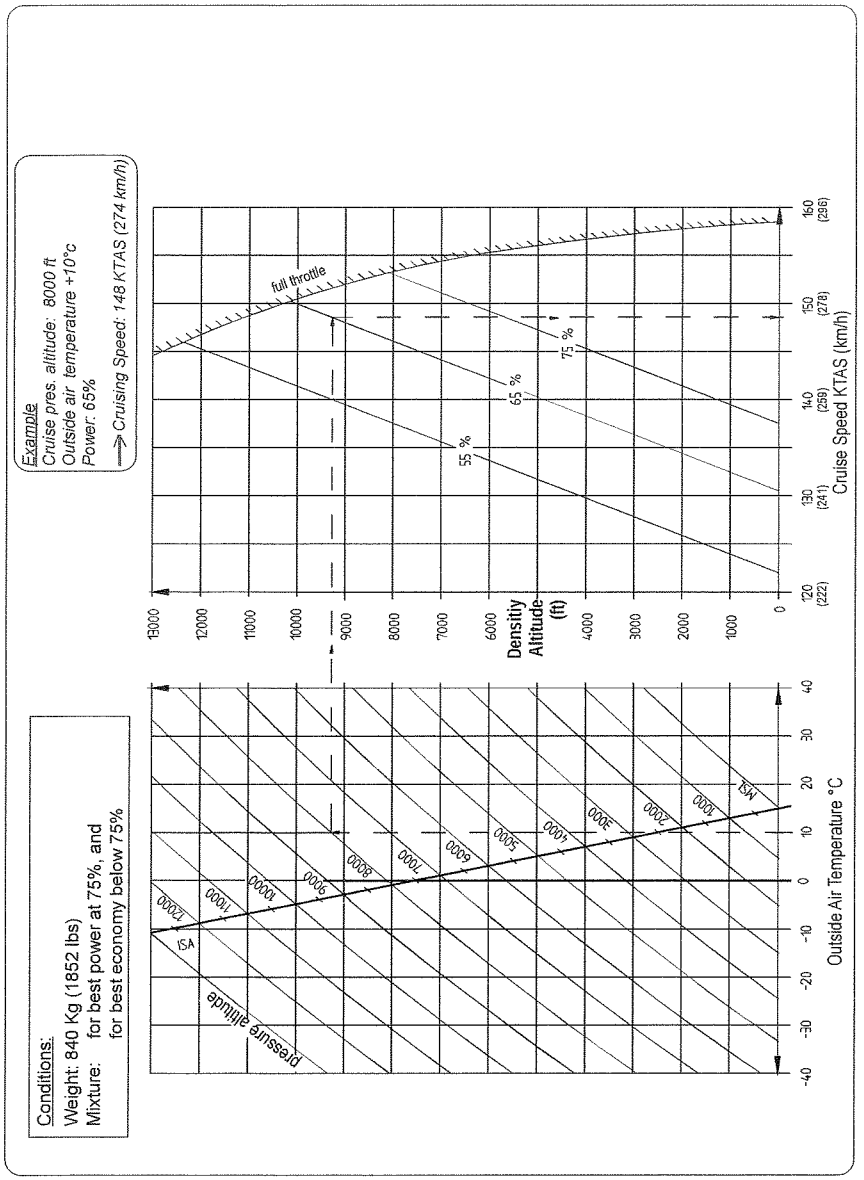


5.8 TIME, FUEL AND DISTANCE TO DESCEND

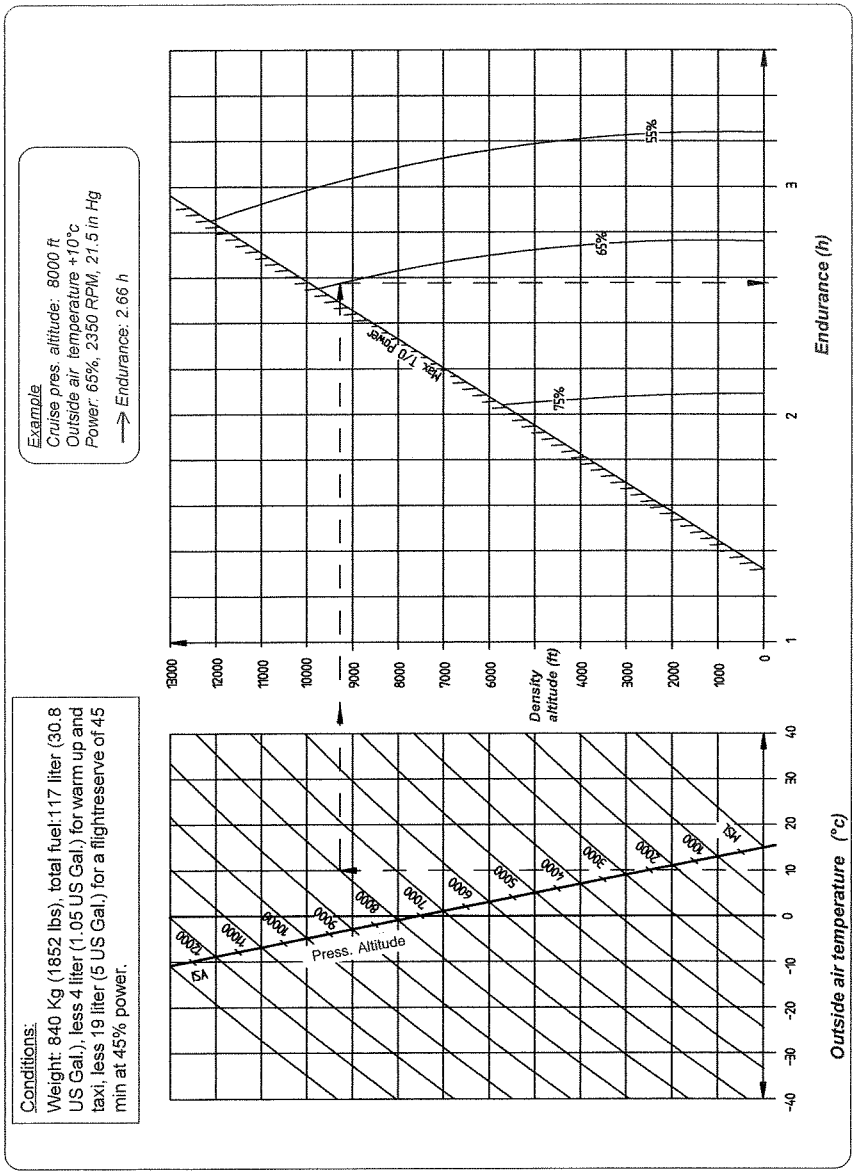
During the descent adjust the engine power for V = 150 KIAS and a rate of descent of 1000ft/min. Furthermore the engine temperature has to be kept in the green range. The diagramm below may be used for all A/C masses and temperatures.



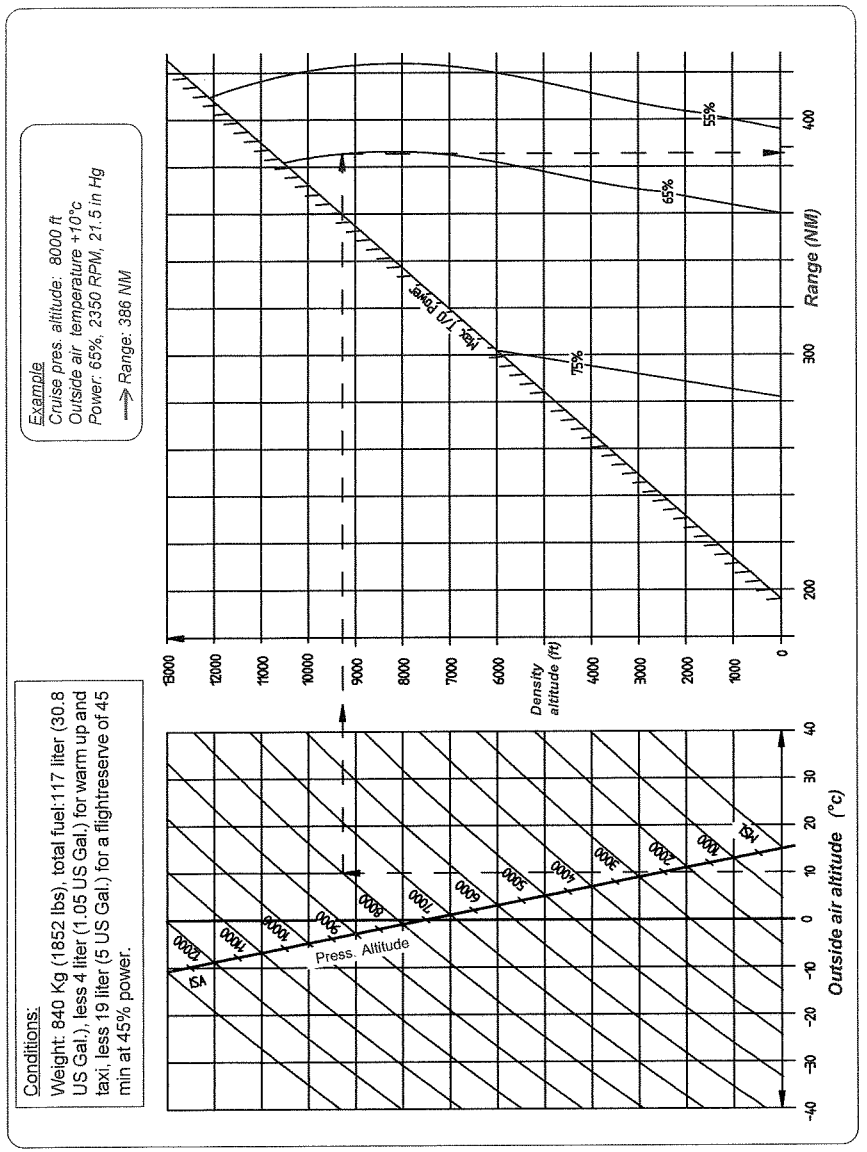
5.9 CRUISING SPEED



5.10 ENDURANCE



5.11 RANGE


Conditions:
Weight: 840 Kg (1852 lbs), total fuel:117 liter (30.8 US Gal.), less 4 liter (1.05 US Gal.) for warm up and taxi, less 19 liter (5 US Gal.) for a flightreserve of 45 min at 45% power.
Example
Cruise pres. altitude: 8000 ft
Outside air temperature +10°c
Power: 65%, 2350 RPM, 21.5 in Hg
Range: 386 NM
Press. Altitude
Outside air altitude (°c)
Density altitude (ft)
Range (NM)
MSL
MSL V/D Points
75%
65%
55%

5.12 CRUISE PERFORMANCE

Range and Endurance values for a T/O weight of 840 kg (1852 lbs) including fuel for warm-up and takeoff from SL, max. cont. Power climb to cruising altitude, and a reserve of 19 liters (5 US Gal.) for 45 minutes with 45% power. 5 liters (1.3 US Gal.) unusable fuel is taken into account. (At ISA - Conditions.)

PA	Eng.	Manif. Press.	Power Setting		Fuel Consumption		TAS		IAS		Endur. ①	Range ①	Mixture ②
[ft]	[RPM]	[IN HG]	[%]	[Hp]	[l/h]	(gal/h)	[Kts]	(km/h)	[Kts]	(km/h)	[h]	[NM]	Best ...
2000	2400	25,0	75	150	46	(12,2)	141	(261)	135	(250)	2,05	288	Power
	2300	23,5	65	130	34	(9,0)	134	(248)	128	(237)	2.75	368	Economy
	2000	23,5	55	110	29	(7,7)	126	(233)	122	(226)	3.21	405	Economy
4000	2400	24,5	75	150	46	(12,2)	145	(269)	135	(250)	2,05	294	Power
	2200	23,8	65	130	34	(9,0)	138	(256)	128	(237)	2.73	374	Economy
	2000	23,1	55	110	29	(7,7)	130	(241)	123	(228)	3.18	412	Economy
6000	2500	23,1	75	150	46	(12,2)	150	(278)	135	(250)	2.05	303	Economy
	2200	23,0	65	130	34	(9,0)	142	(263)	128	(237)	2,71	382	Economy
	2000	22,8	55	110	29	(7,7)	134	(248)	123	(228)	3.15	421	Economy
8000	2350	21,5	65	130	34	(9,0)	146	(270)	128	(237)	2.70	389	Economy
	2050	21,5	55	110	29	(7,7)	138	(256)	122	(226)	3,12	428	Economy
10000	2500	19,9	65	130	35	(9,2)	150	(278)	128	(237)	2.60	380	Economy
	2200	20,0	55	110	30	(7,9)	142	(263)	121	(224)	2.98	416	Economy
12000	2300	18,3	55	110	31	(8,2)	146	(270)	122	(226)	2.85	408	Economy

NOTE

① For temperatures above/ below Standard (ISA), increase/decrease Range 1,7% and Endurance 1,1% for each 10°C above/below Standard Day Temperature for particular altitude.

② **Leaning <u>with</u> exhaust gas temperature (EGT) gage**

For the adjustment **"Best Power"**, first lean the mixture to achieve the top exhaust temperature (peak EGT) and then enrich again until the exhaust temperature is 100°F lower than peak EGT.

For the adjustment **"Best Economy"**, simply lean the mixture to achieve the top exhaust temperature (peak EGT).

Leaning <u>without</u> exhaust gas temperature (EGT) gage and flowmeter

Slowly move mixture control from "Full rich" position towards lean position. Continue leaning until slight loss of power is noted (Loss of power may or may not be accompanied by rough engine run). Then enrich until engine runs smoothly and power is regained.

CAUTION

Always return the mixture to full rich before increasing power settings.

5.13 LANDING DISTANCE

Power: Idle and propeller of full fine
Landing runway: Paved level dry runway
For every 5 kts (9 km/h) headwind, the landing distance (L/D) can be decreased by 10%.
For every 2 kts (4 km/h) Tailwind [up to 10 kts (19 km/h)], the landing distance (L/D) is increased by 10%. On a solid, dry and plain Grass Runway, the landing distance (L/D) is increased by 15%.

Weight: 700 kg (1543 lbs) Landing speed at 15m (50ft) : 70 KIAS (130 km/h)
Landing distance in meter Side slip from 15m (50 ft) down to ca.5 m (15ft)

	-20°C OAT		-10°C		0°C		10°C		20°C		30°C		40°C	
PA	L/D	50 ft	L/D	50 ft	L/D	50 ft	L/D	50 ft	L/D	50 ft	L/D	50 ft	L/D	50 ft
Sea level	150	341	164	371	178	403	192	436	208	471	223	507	240	544
2000	166	377	181	411	197	446	213	483	230	521	247	561	266	603
4000	185	418	201	456	218	495	236	535	255	578	274	622	295	668
6000	205	465	223	506	242	549	262	595	283	642	305	691	327	742
8000	228	517	248	563	269	611	292	661	315	714	339	768	364	825

Landing distance in ft

	-20°C OAT		-10°C		0°C		10°C		20°C		30°C		40°C	
PA	L/D	50 ft	L/D	50 ft	L/D	50 ft	L/D	50 ft	L/D	50 ft	L/D	50 ft	L/D	50 ft
Sea level	492	1119	538	1217	584	1322	630	1430	682	1545	732	1663	787	1785
2000	545	1237	594	1348	646	1463	699	1585	755	1709	810	1841	873	1978
4000	607	1371	659	1496	715	1624	774	1755	837	1896	899	2041	968	2192
6000	673	1526	732	1660	794	1801	860	1952	928	2106	1001	2267	1073	2434
8000	748	1696	814	1847	883	2005	958	2169	1033	2343	1112	2520	1194	2707

Weight: 840 kg (1852 lbs) Landing speed at 15m (50ft) : 73 KIAS (135 km/h)
Landing distance in meter Side slip from 15m (50 ft) down to ca.5 m (15ft)

	-20°C OAT		-10°C		0°C		10°C		20°C		30°C		40°C	
PA	L/D	50 ft	L/D	50 ft	L/D	50 ft	L/D	50 ft	L/D	50 ft	L/D	50 ft	L/D	50 ft
Sea level	194	440	211	479	229	520	248	563	268	608	288	654	310	703
2000	215	487	234	531	254	576	275	623	297	673	319	724	343	778
4000	238	540	259	588	282	639	305	691	329	746	354	803	380	863
6000	265	600	288	653	313	709	338	767	365	828	393	892	422	958
8000	294	667	320	727	348	789	376	854	406	921	437	992	470	1065

Landing distance in ft

	-20°C OAT		-10°C		0°C		10°C		20°C		30°C		40°C	
PA	L/D	50 ft	L/D	50 ft	L/D	50 ft	L/D	50 ft	L/D	50 ft	L/D	50 ft	L/D	50 ft
Sea level	636	1444	692	1572	751	1706	814	1847	879	1995	945	2146	1017	2306
2000	705	1598	768	1742	833	1890	902	2044	974	2208	1047	2375	1125	2552
4000	781	1772	850	1929	925	2096	1001	2267	1079	2448	1161	2635	1247	2831
6000	869	1969	945	2142	1027	2326	1109	2516	1198	2717	1289	2927	1385	3143
8000	965	2188	1050	2385	1142	2589	1234	2802	1332	3022	1434	3255	1542	3494

SECTION 6

WEIGHT AND BALANCE AND EQUIPMENT LIST

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6.1 GENERAL

This section describes the procedure for establishing the basic weight and moment of the aircraft. Sample forms are provided for reference. Procedures for calculating the weight and movement for various operations are also provided. A comprehensive list of all equipment available for this aircraft is included. It is the responsibility of the pilot to ensure that the aircraft is loaded properly.

6.2 AIRCRAFT WEIGHING PROCEDURE

The aircraft weight is determined by weighing all three wheel loads simultaneously by three scales with the aircraft levelled.
(Upper fuselage reference line horizontal)

Datum line for weight arms x is the fire wall.

X1 = distance: fire wall - main wheel

X2 = distance: fire wall - tail wheel

XN = distance: fire wall - item N

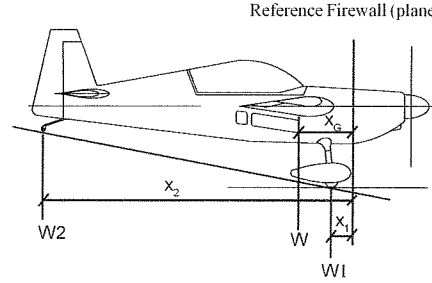
XG = distance: fire wall - Center of Gravity

W1 = Sum of weights indicated by the two scales below the main wheels

W2 = Weight indicated by the scale below the tail wheel

W = Total weight = W1 + W2

$$XG = \frac{(W1 \times X1) + (W2 \times X2)}{W} = \text{C.G. position}$$



$$W = W1 + W2, \quad XG = \frac{(W1 \times X1) + (W2 \times X2)}{W}$$

If a new weight is added to the known old weight and C.G. position the resulting new weight and C.G. can be obtained by a simple calculation:

Situation before adding item:

Wo, Xo = Airplane weight, C.G. position
Wn, Xn = Weight, distance from fire wall of item to add

New Weight of airplane and new C.G.:

$$W = Wo + Wn$$
$$XG = \frac{Wo \times Xo + Wn \times Xn}{W} : \text{C.G. position}$$

6.2.1 OWNERS WEIGHT AND BALANCE RECORD

Enter below all weight change data from aircraft log book.

EXTRA 200	SERIAL NUMBER: 008			REGISTRATION: N149C		
Date	Description of modification	Weight change Added (+), Removed (-)			Running empty weight	
		Wt./kg [lbs]	Arm/cm [inch]	Moment/kg*cm [lbs*inch]	Wt./kg [lbs]	Moment/kg*cm [lbs*inch]
1.17.1997	Empty weight as delivered	___	___	_____	554 kg	38260 kg*cm
02.15.1997	See Next Page.				1234.2 [lb]	34557.6 [lbs*in]
04.06.1998	See Next Page.				1236.1 [lb]	34610.8 [lbs*in]
08.06.1999	See Next Page.				1253.6 [lb]	34734.3 [lbs*in.
08.25.2008	See Next Page				1255 lb.	36411 [lb*in]

WEIGHING REPORT EA 300/200

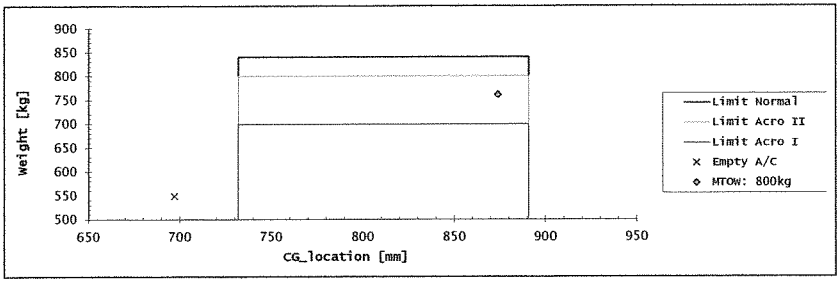
Extra Aircraft Product Support - USA -
Southeast Aero Services Inc. Saint Augustine, FL.

Weighing EA-300/200, SN08, dated 25.08.2008
Equipmentlist dated 25.08.2008 , Wing Tank Standard

	Arm	Mass	Moment
	mm	kg	kgmm
Main wheel load (LH)	400	265.0	106000
Main wheel load (RH)	400	254.0	101600
Tail gear load	5000	35.0	175000
Fuel, center	350	0.0	0
Fuel Wing, Standard 62 Kg	501	0.0	0
	501	0.0	0
Engine oil, 7 Qts		incl.	
Empty Weight	691	554.0	382600

Example: 2 pilots, cat. acrobatic

MTOW: 800kg	CG	Mass	Moment
Empty AC	691	554.0	382600
Fuel center tank	350	22.0	7700
Fuel Wing, standard	501	0.0	0
Fuel Wing, long range	501	0.0	0
PIC+parachute	1850	95.0	175750
PAX+parachute	1043	95.0	99085
In Flight Condition	868	766.0	665135.0



Aircraft inspector:	Date:	Stamp:	Sign:
Claude Bonner	25.8.2008	3040442IA	

Southeast Aero Services Inc.

St. Augustine Airport 385 Hawkeye View Lane.
St. Augustine Fl. 32095

Aircraft Make: Extra Model: EA-300/200 S/nr.: 008

08.25.2008 N149C

Smoke system removed.

Aircraft Weighed With Fuel Tanks ____Full X Empty

	WEIGHT		ARM		
Tail Scale Reading	92.0	X	196.8	=	18105.6
L. Main Scale Reading	592.0	X	15.7	=	9318.1
R. Main Scale Reading	571.0	X	15.7	=	8987.5
Totals	1255.0				36411.2

36411.2 (Total Moment) divided by 1255.0 (Total Empty Weight) equals 29.0 (New C.G.)

The New Empty Weight Is 1255.0 Pounds.

The New Center Of Gravity is 29.0 Inches Aft Of Datum.

The New Useful Load Is 597.0 Lbs at 1852.0 Lbs. Gross Weight. Normal Category
288.0 Lbs at 1543.0 Lbs. Gross Weight. Acrobatic Category 1
508.0 Lbs at 1763.0 Lbs. Gross Weight. Acrobatic Category 2

See Aircraft Flight Manual For Center Of Gravity Limits

Claude A. Bonner
AP3040442IA

Southeast Aero Services Inc.
4900 U.S. Hwy 1 North
St. Augustine, Fl. 32095

Extra Aircraft Product Support U.S.A
All Models

Date:08/06/1999

Supplemental Weight And Balance Data
And Supplemental Equiptmer' List
Computed

Make: Extra Flugzeubau

Model: Ea-300/200 S/N:0 ,8 n149c

Owner: Johnson Aviation Ltd.

Address: 201 N Dup nt h vy.

City:New Castle,

State: De. Zip:19720-6400

Item	Weight	Arm	Moment
Aircraft-Standard	123[illegible]	28.0	34618.8
Installed:			
1) Smoke Generating Pump, Solenoid Valve, Electrical Wiring And System Plumbing.	5.5	-3.0	-16.5
2) Smoke Oil Tank With Accessories	8.0	12.0	96.0
3) Smoke Refill Pump	4.0	9.0	36.0

		ACRO 1	ACRO 2
Total Empty Weight	: 1253.6		
Total Moment	: 34734.3		
Empty Weight C/G	: 27.7		
Gross Weight	: 1858.0	1549.0	1770.0
Useful Load	: 604.4	295.4	516.4

Signature:
Claude A. Bonner
A&P 433257924IA

SOUTHEAST AERO SERVICES INC.
4900 U.S. HWY 1 NORTH
ST. AUGUSTINE, FL. 32095

EXTRA AIRCRAFT PRODUCT SUPPORT U.S.A
ALL MODELS

DATE:04/06/98

SUPPLEMENTAL WEIGHT AND BALANCE DATA
AND SUPPLEMENTAL EQUIPTMENT LIST
Computed

MAKE: EXTRA FLUGZEUBAU MODEL: EA-300/200 S/N:008 N149C

OWNER: AERO SPORT INC. ADDRESS: 4900 US 1 NORTH

CITY:ST AUGUSTINE STATE: FL ZIP:32095

ITEM	WEIGHT	ARM	MOMENT
AIRCRAFT-METRIC	559.7	70.1	39243.6
AIRCRAFT-STANDARD	1234.2	28.0	34584.7
Removed:			
1) Factory Cabin Heat 2" firewall valve	-0.3	-2.0	0.6
Installed:			
1) Factory Cabin Heat 3" firewall valve	0.4	-2.5	-1.0
2) Factory Cabin Heat 3" regulator valve	0.5	2.0	1.0
3) Factory Cabin Heat 3" cowling duct	0.2	-28.5	-8.6
4) Factory Cabin Heat 3" Regulator Valve Control	1.0	42.0	42.0

		ACRO 1	ACRO 2
TOTAL EMPTY WEIGHT :	1236.1	1549.0	1770.0
TOTAL MOMENT :	34618.8		
EMPTY WEIGHT C/G :	28.0		
GROSS WEIGHT :	1858.0		
USEFUL LOAD :	621.9	312.9	533.9

SIGNATURE: _____
CLAUDE A. BONNER
A&P 433257924IA

Superseded

SOUTHEAST AERO SERVICES INC.
4900 U.S. HWY 1 NORTH
ST. AUGUSTINE, FL. 32095

EXTRA AIRCRAFT PRODUCT SUPPORT U.S.A
ALL MODELS

DATE:02/15/97

SUPPLEMENTAL WEIGHT AND BALANCE DATA
AND SUPPLEMENTAL EQUIPTMENT LIST

MAKE: EXTRA FLUGZEUBAU

MODEL: EA-300/200 S/N:08 N149C

OWNER:AERO SPORT INC.

ADDRESS: 4900 US 1 NORTH

CITY:ST AUGUSTINE

STATE: FL ZIP:32095

ITEM	WEIGHT	ARM	MOMENT
AIRCRAFT-METRIC	554.0	69.1	38,260.0
AIRCRAFT-STANDARD	1,221.6	27.6	33,691.4
INSTALLED:			
1) ACK TECH E-01 ELT	4.0	115.0	460.0
2) TERRA TRT-250D TRANSPONDER	1.5	54.0	81.0
3) TPX ENCODER AT-3000	0.5	34.0	17.0
4) TERRA TPX ANTENNA	0.3	35.0	10.5
5) GARMIN GNC-250 GPS/COMM	2.5	54.0	135.0
6) NAT INTERCOM SYSTEM A48, or	0.8	50.0	40.0
7) GARMIN GPS ANT	0.2	134.0	26.8
8) 2 1/4" FRONT ACC LERO METER	0.8	22.0	18.2
9) 3 1/8" REAR ACCELERO METER	1.20	54.00	64.8
10) SHADIN FUEL TRANSDUCER SYSTEM	0.8	50.0	40.0

		ACRO 1	ACRO 2
TOTAL EMPTY WEIGHT	: 1,234.2	1,549.0	1,770.0
TOTAL MOMENT	: 34,584.7		
EMPTY WEIGHT C/G	: 28.0		
GROSS WEIGHT	: 1,858.0		
USEFUL LOAD	: 623.8	314.8	535.8

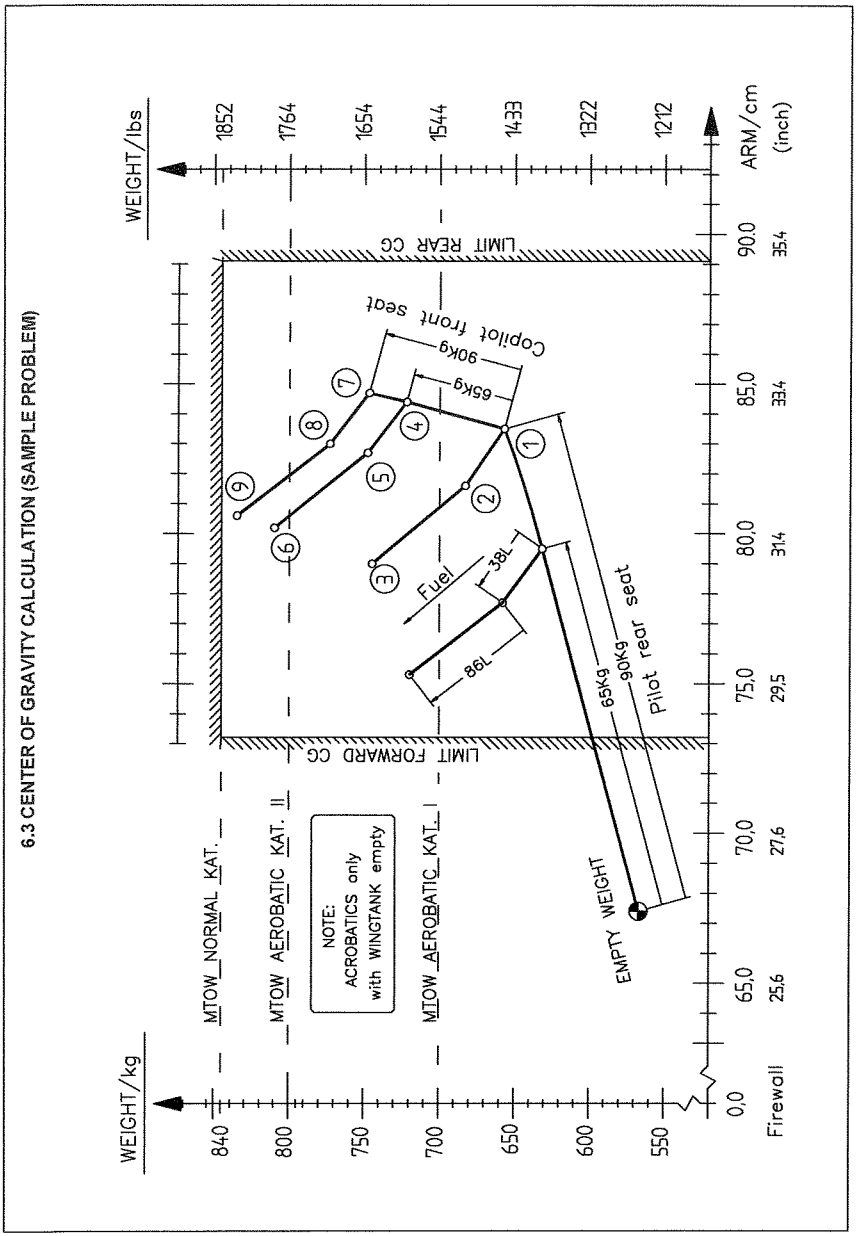
SIGNATURE:_____

CLAUDE A. BONNERA&P 433257924IA

6.3 CENTER OF GRAVITY CALCULATION (SAMPLE PROBLEM)

Position	PILOT Rear Seat		ACRO-TANK Fuel 36 Liter (9.5 US GAL)		COPILOT Front Seat		WING-TANK Fuel 86 Liter (22.6 US GAL)	
	(kg)	(lbs)	(kg)	(lbs)	(kg)	(lbs)	(kg)	(lbs)
(1)	90	198.5	-	-	-	-	-	-
(2)	90	198.5	25.9	57.1	-	-	-	-
(3)	90	198.5	25.9	57.1	-	-	62	136.7
(4)	90	198.5	-	-	65	143.3	-	-
(5)	90	198.5	25.9	57.1	65	143.3	-	-
(6)	90	198.5	25.9	57.1	65	143.3	62	136.7
(7)	90	198.5	-	-	90	198.5	-	-
(8)	90	198.5	25.9	57.1	90	198.5	-	-
(9)	90	198.5	25.9	57.1	90	198.5	62	136.7

6.3 CENTER OF GRAVITY CALCULATION (SAMPLE PROBLEM)



6.3.1 SAMPLE

Takeoff Condition:		
Pilot On Rear Seat	90.0 kg	(198.5 lbs)
Copilot On Front Seat	65.0 kg	(143.3 lbs)
"Acro & Center" Fuel	25.9 kg	(57.1 lbs)
86 l Fuel In Wing Tanks	62.0 kg	(136.7 lbs)
Aircraft Empty Weight	567.0 kg	(1250 lbs)
	========	==========
	809.9 kg	(1785.6 lbs)

To find C.G., follow line "Pilot Rear Seat" from Empty Weight to "90 kg" [198.5 lbs] (Point "1"). Continue on line "Copilot Front Seat" to 65 kg (Point "4"). Now follow line "Fuel" via Point "5" to Point "6":

FIND:	Weight	~ 810 kg	(1785.7 lbs)
	C.G.	~ 80.2 cm	(31.5 inch)

6.3.2 WEIGHT AND BALANCE RECORD SHEET

	WEIGHT (Kg)	ARM (cm)	MOMENT (Kg cm)
EMPTY WEIGHT:			
PILOT:			
COPILOT:			
"ACRO & CENTER" - FUEL:			
"WING TANK" - FUEL:			
	Σ W =		**Σ (W x X) =**

$$XG = \frac{\Sigma (W \times X)}{\Sigma W} = ___ cm$$

6.4 LOADING WEIGHTS AND MOMENTS

OCCUPANTS : max. 2

WEIGHT Pilot + Parachute		PILOT REAR SEAT Arm = 185cm *(73 inch)*		COPILOT FRONT SEAT Arm = 93.7 cm *(37 inch)*	
KG	*LBS*	KG x CM	*(IN x LBS)* MOMENT	KG x CM	*(IN x LBS)*
65	143	12012	*(10404)*	6091	*(5275)*
70	154	12936	*(11204)*	6559	*(5681)*
75	165	13860	*(12005)*	7028	*(6087)*
80	176	14784	*(12805)*	7496	*(6493)*
85	187	15708	*(13605)*	7965	*(6898)*
90	198	16632	*(14406)*	8433	*(7304)*

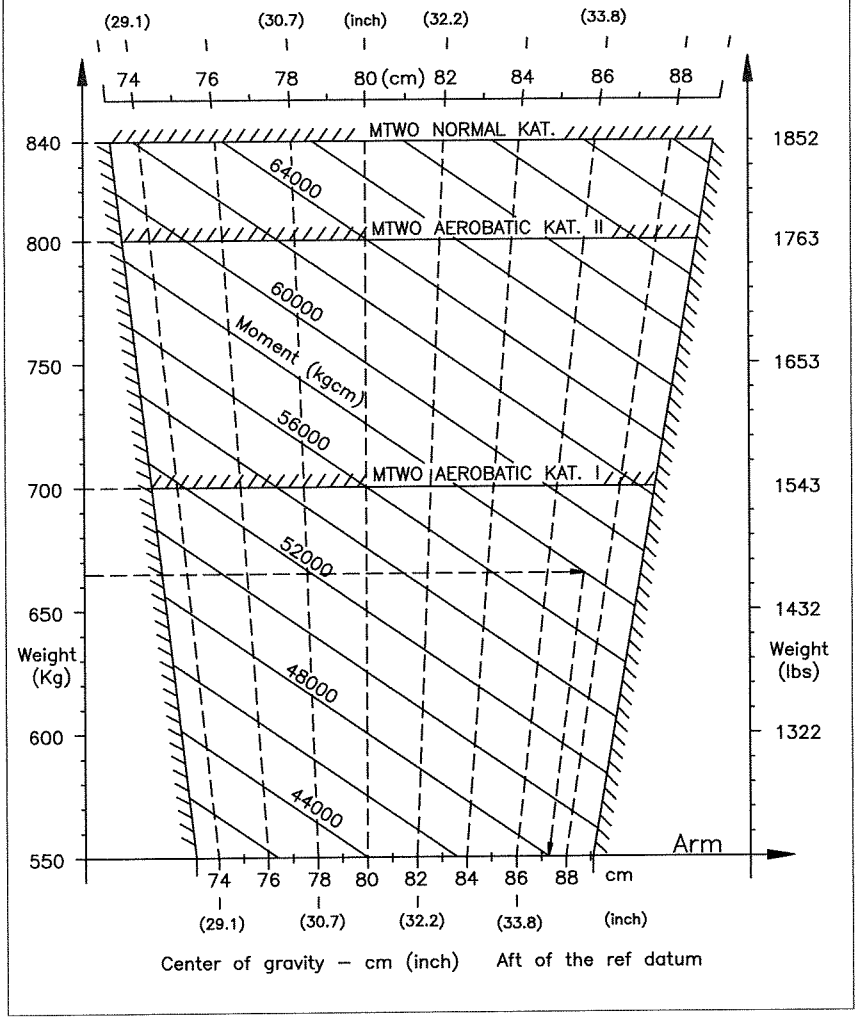
Fuel in the System: Max. 122 LITERS (32.1 US GAL.)

Fuel					
"ACRO & CENTER" TANK			WINGTANK		
LITER *(US GAL)*	KG *(LBS)*	KG x CM *(LBS x IN)*	LITER *(US GAL)*	KG *(LBS)*	KG x CM *(IN x LBS)*
5 (1.3)	3.6 (8)	126 (109)	10 (2.6)	7.2 (16)	361 (312)
10 (2.6)	7.2 (16)	252 (218)	20 (5.3)	14.4 (32)	721 (625)
15 (4.0)	10.8 (24)	378 (327)	30 (7.9)	21.6 (48)	1082 (937)
20 (5.3)	14.4 (32)	504 (437)	40 (10.6)	28.8 (63)	1443 (1250)
25 (6.6)	18.0 (40)	630 (546)	50 (13.2)	36.0 (79)	1804 (1562)
30 (7.9)	21.6 (48)	756 (655)	60 (15.8)	43.2 (95)	2164 (1875)
36 (9.5)	25.9 (58)	932 (799)	70 (18.5)	50.4 (111)	2525 (2187)
			80 (21.1)	57.6 (127)	2886 (2499)
			86 (22.7)	61.9 (136)	3102 (2687)

6.5 WEIGHTS AND MOMENTS LIMITS

Example:

At 655 kg and 58000 kgcm the C.G. location is 87.0 cm aft of ref. datum



6.6 EQUIPMENT LIST EXTRA-200 S/N: 008

QTY	ITEM	MANUFACT	PART NUMBER	WEIGHT (KG)	ARM (m)	Mark if installed	Required (R) Optional (O) Alternate (A)
1 1 1	Engine Magneto, LH Magneto, RH	Lycoming Slick Slick	AEIO-360-A1E 4372 4370	128.40 2.30 2.00	-0.51 -0.22 -0.22	✓ ✓ ✓	R R R
4	Shock Mounts	Lord	J-7402-37	1.70	-0.29		R
4	Shock Mounts	Barry Controls	94150-02	1.70	-0.29	✓	A
1	Exhaust System 4 in 1	Gomolzig	EA200-606000	7.35	-0.29	✓	R
1	Exhaust System 4 in 1 with Silencer	Gomolzig	EA200-606000/1	7.90	-0.27		O
1	Starter	Lycoming	MZ 4221	8.15	-0.67		R
1	Starter, lightweight	SKY-TEC (Lycoming)	149-12 LS (31A22 104)	3.65	-0.75	✓	A
1	Fuel Injector	Bendix	RSA-5 AD 1	3.46	-0.57	✓	R
1	Elec. Fuel Pump	Weldon Tool	8120-M	1.10	-0.04	✓	R
1	Elec. Fuel Pump	Weldon Tool	B8120-M	1.10	-0.04		A
1	Fuel Selector	Allen	6 S 122	0.19	0.26	✓	R
1	Oil Cooler	Stewart Warner	8406 R	1.40	-0.77	✓	R
1 1	Propeller Spinner	MT-Propeller Mühlbauer	MTV-12-B-C / C183 - 17e P-440	22.90	-1.02	✓	R R
1	Governor	Woodward	D210982	1.10	-0.79	✓	R
1	Governor	MT-Propeller	P-880-3	1.10	-0.79		A
1	Set Fuel, Oil & Sense Hoses in Eng. Comp.	Aeroquip	Type 303 with Firesleeve AE 102	6.15	-0.21	✓	R
1	Set Fuel Hoses in Cabin Comp.	Aeroquip	Type 303	1.38	0.21	✓	R
1	Sense Hoses (Cabin) (Oil, Fuel & MA Press)	Knapp/Hoerbiger	HS3AM and H3MM	0.30	0.77		R
1	RPM Vernier Control	ACS Products Co.	A-750-30-1200	0.71	0.55	✓	R
1	RPM Vernier Control	ACS Products Co.	A-750-30-960	0.60	0.49		A
1	Mixture Vernier Control	ACS Products Co.	A-750-20-1080	0.65	0.68	✓	R
1	Mixture Vernier Control	ACS Products Co.	A-750-20-960	0.60	0.60		A
1	Throttle Control	Teleflex Marine	F303 02750	0.50	0.60	✓	R
1	Altimeter, front	United Instr.	UI5934PD-3 A.134	0.60	0.52	✓	O
1	Altimeter, front (metric)	Winter	4110	0.33	0.55		O
1	Altimeter, rear	United Instr.	UI5934PD-3 A.134	0.60	1.50	✓	R
1	Altimeter, rear (metric)	Winter	4110	0.33	1.50		A
1	Altitude Blind Encoder	ACK	A-30	0.15	1.26	✓	O
1	Airspeed Indicator, front	Winter	6533 422	0.21	0.52	✓	O
1	Airspeed Indicator, front	United Instr.	UI8030 B.871	0.22	0.52		O
1	Airsp. Ind, front (metric)	Winter	6531 422	0.21	0.52		O
1	Airspeed Indicator, rear	Winter	6533 422	0.21	1.50	✓	R
1	Airspeed Indicator, rear	United Instr.	UI8030 B.871	0.22	1.50		A
1	Airsp. Ind., rear (metric)	Winter	6531 422	0.21	1.50		A

QTY	ITEM	MANUFACT	PART NUMBER	WEIGHT (KG)	ARM (m)	Mark if installed	Required(R) Optional(O) Alternate (A)
1	RPM Indicator	VDO	333 230 115 002 or 333 035 001G	0.31	1.49		R
1	RPM Indicator digital	Horizon	P100-230-464-00	0.68	1.49	✓	A
1	Magnetic Compass	Airpath	C 2300	0.25	1.50	✓	R
1	Accelerometer	Kollsman or Pioneer or Bendix	AN 5745	0.37	1.50	✓	O
1	Accelerometer	EXTRA	DSA 12	0.37	1.49		O
1	Accelerometer	EZE Technologies	DA-55	0.16	1.49		O
1	Turn & Bank Indicator	United Instr.	9501-2 / TSO C3b	0.55	1.50		O
1	Turn & Bank Indicator	Castleberry	C101 / TSO C101T	0.55	1.50		
1	VHF RADIO	Becker	AR 3201	0.90	1.46		R
1	VHF RADIO	Garmin	GNC 250	0.90	1.46	✓	A
1	GPS / NAV / COM	Garmin	GNS 430	2.95	1.40		O
1	Transponder	Terra	TRT-250D	0.89	1.40	✓	O
1	Transponder	Garmin	GTX 327	0.95	1.40		O
1	Transponder	Filser	TRT-600 (LAST)	0.70	1.45		O
1	Transponder	Filser	TRT-800	0.70	1.45		O
1	Transponder	Garmin	GTX 330	1.50	1.40		O
1	Transponder	Bendix/King	KT 73	1.64	1.40		O
1	Transponder	Becker	BXP6401-1	0.80	1.45		O
1	Transponder	Becker	BXP6401-2	0.80	1.45		O
1	Transponder	Becker	ATC-2000	1.20	1.40		O
1	Transponder	Becker	ATC-4401	0.73	1.45		O
1	Transponder Antenna	Comant Industries	CI 105	0.11	0.08	✓	O
1	Transponder Antenna	Bendix/King	KA 60	0.11	0.08	✓	O
1	COM Antenna	Moba	210FA	0.10	4.02		R
1	COM Antenna	EXTRA	83205A	0.10	4.02	✓	A
1	COM Antenna	Pointer	P1 3001-10	0.05	4.02		A
1	ELT	ACK Technologies	E-01	1.00	2.60	✓	O
1	Oil-Press. Indicator	UMA	3-311-32 or 10-12400	0.05	1.52		R
1	Oil Press./Oil Temp Ind.	Amitek or Christen	61943	0.51	1.52	✓	A
1	Oil Temp. Indicator	Westach	2A9-2	0.05	1.52		R
1	Oil Temp Sensor	Westtach	W399-S9	0.08	-0.20		R
1	Oil Press./ Oil Temp. Ind. (3 1/8")	Westach	3DA3-3MM or 3DA3-3KV	0.14	1.50		A
1	Oil Temp. Sender	Westach	W399-S9	0.08	-0.20		A
1	Oil Press Sensor	Westach	387-100MM or 387-100KV or 387-11MM	0.12	0.04		A
		or Mediamate	9308704	0.12	0.04		A
1	Oil Press / Oil Temp Ind. (2 1/4")	Westach	2DA3-3MM or 2DA3-3KV	0.09	1.50		A
1	Oil Temp. Sender	Westach	W399-S9	0.08	-0.20		A
1	Oil Press Sensor	Westach	387-100MM or 387-100KV or 387-11MM	0.12	0.04		A
		or Mediamate	9308704				
1	Fire Extinguisher	H3R	12533-003	.51	1.52		A

QTY	ITEM	MANUFACT	PART NUMBER	WEIGHT (KG)	ARM (m)	Mark if installed	Required(R) Optional(O) Alternate (A)
1	EGT/CHT Indicator	Westach	2 DA 1	0.07	1.52	✓	O
1	EGT Probe	Westach	712-2 DWK	0.06	-0.46	✓	O
1	CHT Probe	Westach	712-7 DK	0.05	-0.50	✓	O
1	Manifold press. Indicator	UMA	7-100-20	0.13	1.52		R
1	Fuel Pressure Ind.	UMA	3-313-10	0.05	1.52		R
1	Man Press / Fuel Press	United Instr.	UI6331 H.52	0.54	1.50	✓	A
1	Man Press / Fuel Flow	United Instr.	UI6331 H.212	0.54	1.50		A
1	Fuel Cont. Probe Wing (Wing Tank)	VDO	226 801 015 001 G	0.12	0.51	✓	R
1	Fuel Cont. Ind. Wing (Wing Tank)	VDO	301 271 036 001 K or 301 030 001 G	0.08	1.50	✓	R
1	Fuel Cont. Probe Acro & Center Tank	VDO	224 082 006 097 R	0.20	0.36	✓	R
1	Fuel Cont. Ind. Acro & Center Tank	VDO	301 272 052 001 K or 301 030 002 G	0.14	1.50	✓	R
1	Ammeter	VDO	190 004 039 002 or 190 037 002 G	0.08	1.50	✓	R
1	Volt/Ammeter	Electronics Intern.	VA-1A	0.26	1.51		A
1	Volt/Ammeter	Electronics Intern.	VA-1A-50	0.22	1.50		A
1	Shunt	Electronics Intern.	S-50	0.09	1.40		A
1	Alternator 60Amp	Electrosystems	ALX 8421 LS	5.90	-0.69	✓	R
1	Voltage Regulator	Lamar	B-00371-25	0.15	0.02	✓	R
1	Low Vol. Monitor	Lamar	B-00378-4	0.03	0.02		R
1	Alternator 65 Amps	Bosch	0 120 489 935	4.60	-0.72		A
1	Alternator 55 Amps	Bosch	0 120 489 917	4.20	-0.72		A
1	Alternator 55 Amps	Bosch	0 120 489 469	4.20	-0.72		A
1	Battery	Sonnenschein	DRYFIT A212/24G	8.60	-0.11		R
1	Battery	Aero Batteries	SBS J30X	9.70	-0.11		A
1	Battery	Exide/Sonnenschein	Dryfit A512/25G5	9.65	-0.11		A
1	Battery	B&C	BC100-MJ	11.29	-0.11	✓	A
1	Batt. Charger Plug	EXTRA	146 19 20	0.02	0.26	✓	O
1	External Power Recept.	Diverse	AN 2552-3A	1.46	1.77	✓	O
1	Ext. Power Solenoid	Switches Kidde	22735	0.40	-0.03	✓	O
1	Ext. Power Solenoid (cont. operation)	White-Rodgers	70-111-225-5	0.40	-0.03		O
1	Main Bus Solenoid	White-Rodgers	70-111-226-5	0.40	-0.03	✓	R
1	Starter Solenoid	Switches Kidde	22735	0.40	-0.03	✓	R
1	Ignition switch	TCM	10-357200-1	0.15	1.50	✓	R
1	Main Bus Fuseholder	MTA	PN 03.00360	0.03	-0.01		O
1	Main Bus Strip Fuse (40 Amps)	MTA	PN 02.00300		-0.01		O
1	PCB Auto Fuse	EXTRA	83290.1	0.01	-0.03		O
1	Low Voltage Light	OAK	MS 25041-2	0.02	1.53		R
1	Switches	Cutler Hammer	div.	0.04	1.52	✓	R
3	Circuit Breakers	Potter&Brumfield	div.	0.14	1.51		R
5	Switch/Circuit Breakers	Potter&Brumfield	div.	0.25	1.51	✓	R

QTY	ITEM	MANUFACT	PART NUMBER	WEIGHT (KG)	ARM (m)	Mark if installed	Required(R) Optional(O) Alternate (A)
1	Strobe Light (single)	Whelen	A 470-W	0.13	0.85		R
1	Strobe Power Supply	Whelen	A490 T	0.27	0.85		R
2	Strobe/Nav LTS (RH/LH)	Whelen	A 600-PG/PR-14	0.23	0.80	✓	A
2	Strobe Power Supply	Whelen	A 490 T	0.54	0.80	✓	A
1	Stall Sensor	EXTRA	73106.1	0.07	0.42	✓	R
1	Stall Warn Horn	EMAG	EM-S 110P	0.13	1.37	✓	R
2	Main Wheel Tires	Diverse	5.00-5 / 6PR	3.90	0.35	✓	R
2	Main Wheel Rim	Cleveland	40 - 151	4.00	0.35	✓	R
2	Brake Assy	Cleveland	30 - 164	1.40	0.35	✓	R
2	Brake Cylinder front	Cleveland	10 - 20 E	0.55	0.15		R
2	Brake Cylinder rear	Cleveland	10 - 20 E	0.55	0.95		R
2	Brake Cylinder front	Matco	MC4E	0.55	0.15	✓	A
2	Brake Cylinders rear	Matco	MC4E	0.55	0.95	✓	A
1	Tail Wheel 5"	EXTRA	steerable	5.50	4.70		A
1	Tail Wheel 6" Assy (Soft)	Special Products Aviation Inc.		5.90	4.70	✓	A
1	Brake Fluid Reservoir	ACS	A-315	0.20	-0.04	✓	R
1	Safety Belt Assy. Front	Hooker	1 013 030	2.90	1.10		R
1	Safety Belt Assy. Front	Hooker	1 011 230 With Ratchet	3.30	1.80	✓	A
1	Safety Belt Assy. Rear	Hooker	1 011 230 With Ratchet	3.30	1.80	✓	R
1	Standard Canopy	EXTRA	27301.000-LV	11.00	1.45	✓	R
1	Single Seat Canopy	EXTRA	87410.000-LV	9.20	1.45		O
1	Long Range Tank	EXTRA	87901	1.80	0.62		O
1	Leather Interior	EXTRA	Assy	1.50	1.55	✓	O
1	Aresticard Holder	EXTRA	Assy	0.09	1.51	✓	O
1	Airtow Hook	EXTRA/TOST	83607A0	0.50	5.05		O
1/2	Sighting Device LH/RH	EXTRA	84801.10	each 0.44	1.53	✓	O
2	Wing Tie Down Rings	EXTRA	83801.2-01	0.04	1.18	✓	O
1	Rear Seat Pan Stand.	EXTRA	77201.4	1.40	1.94		R
1	Rear Seat Pan Adjust.	EXTRA	87403.1	1.40	1.94	✓	A
1	Horizontal Tail Standard	EXTRA	37201.1	20.00	4.24	✓	R
1	Horizontal Tail (CRP)	EXTRA	87932	19.00	4.25		A
1	Vertical Tail Standard	EXTRA	37101.1	9.80	4.54	✓	R
1	Vertical Tail (CRP)	EXTRA	87931	9.00	4.54		A
1	3" Cabin Heat System	EXTRA	86606.000	0.99	0.38	✓	O
1	Knapp Brake System	EXTRA	FF0001	0.60	0.77	✓	O
1	Shadin Fuel Flow Sys	Shadin	912021	0.36	1.27	✓	O
~~1~~	~~Smoke System~~	~~Field Approved~~		~~7.93~~	~~0.16~~	✓	~~O~~
1	Intercom	NAT Tech	AA83	0.36	1.27	✓	O

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SECTION 7

DESCRIPTION AND OPERATION
OF AIRCRAFT AND SYSTEMS

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SECTION 7

DESCRIPTION AND OPERATION OF AIRCRAFT AND SYSTEMS

7.1 THE AIRCRAFT

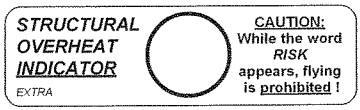
The aircraft EXTRA 200 is designed and developed by EXTRA Flugzeugproduktions- und Vertriebs- GmbH, Dinslaken 46569 Hünxe, Federal Republic of Germany, in accordance with the U.S. Federal Aviation Regulations, part 23, categories normal and acrobatic to fulfil the primary flight training, normal operation rules and acrobatic training up to the unlimited acrobatic level.

EXTRA 200 is a light weight, robust, single piston-engined, two-seat aircraft with a fuselage structure in tig-welded steel-tube construction.

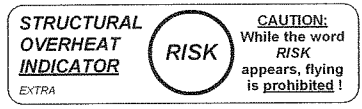
The landing gear, wing, and tail are made of epoxy, reinforced with glass- and carbonfiber. The items are qualified up to 72°C (161,6°F). Not to exceed this temperature limit an appropriate colour specification for composite structure is given by the manufacturer document EA-03205.19.

To check the temperature inside the cockpit (potential "green house" effect) a reversible temperature indicator (*STRUCTURAL OVERHEAT INDICATOR*) is applied on the upper side of the wing main spar in the carry-through section. After reaching the temperature limit of 72°C (161,6°F) the word "*RISK*" appears on the red spot of this structural overheat indicator immediately and flying is prohibited. When the structure cools down below this temperature limit the word "*RISK*" disappears and you may go on with the preflight checklist.

a) Below 72°C (161,6°F)



b) At 72°C (161,6°F) or above



The standard aircraft is designed to operate within a range of ambient air temperature from -20°C to +44°C (-4°F to 111°F) at sea level. It is possible to start the engine using the aircraft battery at -20°C (-4°F) without preheating. Below -10°C (+14°F) OAT a special oil breather line must be adapted (available as kit).

7.2 FUSELAGE

The fuselage structure consists of a steel tube construction integrating the wing and empennage connections as well as the seats. The fuselage except the rear lower part, is faired with an aramid/carbon laminate shell. Within the exhaust area stainless steel sheet metal is used. The upper fuselage body surface is one part from firewall to vertical stabilizer including the correlated canopy frame. Only the lower rear part of the fuselage is covered with Ceconite® 102.

The canopy frame itself is constructed by carbon laminate. The canopy is one part, opens to the right and is held in the open position by a belt. Emergency jettisoning is achieved by simply unlatching the canopy. For additional pilots protection a roll bar is installed behind the rear pilots seat.

7.3 WINGS

The wing is of CRP construction. The dual chamber main spar - fulfilling the requirement for fail safe design - consists of carbon roving caps combined with CRP webs. Core foam is a PVC foam (Divinycell HT 50). The wing shell is built by a Honeycomb sandwich with CRP Laminates. On the surface there is a protective layer of GRP. To prevent buckling of the shell plywood ribs are used. In the area of the wingtanks is a layer of CRP laminate with an incorporated aluminium thread bonded to the metal fuselage structure as means of lightning protection.
The connection to the fuselage is arranged by two bolts piercing through the spar parallel to the centerline of the fuselage and two brackets at the rear spars. Integral fuel cells are provided in the leading edge of the wing extending from the root ribs to half the span of each R/L and L/H wing. The ailerons are supported at three points in spherical bearings pressed into aluminium brackets. To reduce pilot's hand forces the hinge line of the ailerons is positioned 25% of the aileron depth at the root and 21,5% at the tip. Furthermore the ailerons are equipped with "spades" to decrease pilot forces. Ailerons are controlled via the center bracket. To prevent flutter the ailerons are weight balanced in the overhanging leading edge.

7.4 EMPENNAGE

The EXTRA 200 possesses a cruziform empennage with stabilizers and moveable control surfaces. The rudder is balanced aerodynamically at the tip. Spars consist of PVC foam cores, CRP caps and GRP laminates. The shell is built by honeycomb sandwich with GRP laminates. Buckling is prevented by plywood ribs.
Deviating from the other control surfaces the spar webs of the surfaces of the elevator is built by CRP. On the R/H elevator half a trim tab is fitted with two hinges. The control surfaces are mounted in spherical bearings (exception: Trim tab). To prevent flutter rudder and elevator are mass balanced. The balance weight for the rudder is installed in the rudder tip while the balance weight for the elevator is mounted on the elongated center bracket of the elevator extending into the fuselage.

7.5 FLIGHT CONTROL SYSTEM

7.5.1 PRIMARY CONTROL SYSTEM

The EXTRA 200 is standard equipped with full dual primary flight controls including conventional stick-type control columns and adjustable rudder pedals. The primary control surfaces are operated through a direct mechanical linkage.

7.5.2 LONGITUDINAL FLIGHT CONTROL SYSTEM

The two control columns are interconnected by a torque tube. The control movements are from there transferred to the elevator by a push rod.

7.5.3 LATERAL FLIGHT CONTROL SYSTEM

Push and pull rods are connected by sealed ball bearings from the torque tube to the ailerons.

The ailerons are statically as well as dynamically balanced. (Dynamically with spades).

The ailerons are supported by lubricated, sealed bearings.

7.5.4 DIRECTIONAL FLIGHT CONTROL SYSTEM

The dual rudder pedals with brake pedals are adjustable and operate the rudder through a cable system. Springs keep the cables under tension when they are not operated.

7.5.5 SECONDARY CONTROL

The elevator trim control is located on the right side in the rear cockpit.

The canopy lock is operated from the outside by a handle on left side of the canopy by reaching into the cockpit through the window. Inside a handle is located in both cockpits, used for locking as well as for normal operation and for emergency release.

The starter/magneto switch is located on the left side of the instrument panel in front of the rear seat.

7.6 INSTRUMENTATION

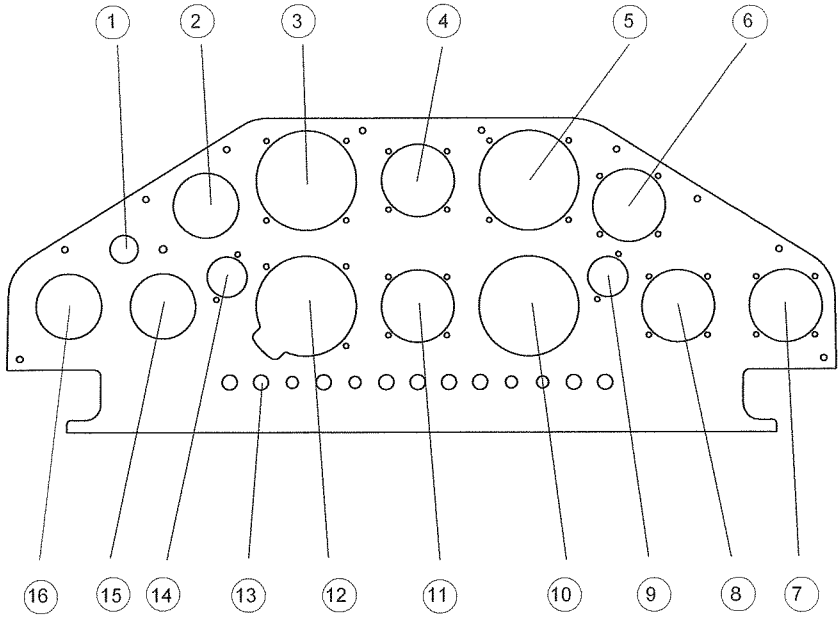
The EXTRA 200 is equipped with flight instruments in both cockpits.

Instruments and placards can be provided with markings in either metric or English units. The colour markings in instruments follow US-FAR, part 23 recommendation (see section 2).

7.6.1 INSTRUMENT PANEL (REAR COCKPIT)

For instrumet panel arrangement of the reat cockpit refer to Fig. 7.6.1, which includes standard and optional equipment marked as such.

Fig. 7.6.1:



Standard	Optional	Position	Item
x		1	Magneto Selector switch & starter
x		2	Amperemeter
x		3	Airspeed indicator
x		4	Magn. Direction indicator
	x	5	g-meter
	x	6	EGT / CHT
	x	7	COM
x		8	Oil temperature
x		9	Oil pressure
x		10	RPM indicator
x		11	Manifold pressure
x		12	Altimeter
x		13	Switches / Circuit breaker
x		14	Fuel pressure
x		15	Fuel Quantity "Acro & Center" Tank
x		16	Fuel Quantity "Wing Tank"
	x	17	Vertical speed indicator
	x	18	Turn and bank indicator
	x	19	Artificial horizon
	x	20	Intercom button
	x	21	Directional gyro
x		22	RPM control, Prop governor
x		23	Mixture control
x		24	Throttle lever
x		25	Stick
x		26	Radio button
x		27	Fuel shutoff valve
x		28	Trim lever and indicator
x		29	Boost pump

NOTE

This list may be modified by the minimum equipment requirements of individual certifying authorities!

7.6.2 INSTRUMENT PANEL (FRONT COCKPIT)

The front instrument panel respectively the front cockpit is equipped with the following positions.

24	Throttle
25	Stick
27	Fuel shutoff valve

The following equipment is optional:

3	Airspeed indicator
4	Magn. Direction indicator
5	g-meter
10	RPM indicator
12	Altimeter
17	Vertical speed indicator
18	Turn and bank indicator
19	Artificial horizon
20	Intercom button
21	Directional gyro
26	Radio button

7.7 LANDING GEAR

The landing gear is a composite construction with a multichamber fibreglass spring in a tail-wheel design.

The main wheels have a size of 5-5.50 and they are equipped with hydraulic disc brakes.

The tail wheel has a solid rubber tire with full-swivel capability.

7.8 SEATS, SEAT BELTS

The seats are ergonomically shaped composite designs. The back rest is adjustable on the ground with 2 quickpins in different positions and angle. The rear pedal-to-seat distance can be varied in different positions. In the front cockpit there is no possibility to adjust either the pedals nor the seat. The seat belt assembly consists of a left and a right shoulder strap, two left and two right lap belts and a negative-g-strap. All belts are adjustable. As each lap belt features a single point release, they are redundant for safety during aerobatics maneuvers. If one release is opened unintentionally, the second one guarantees full safety. For safe operation the releases are arranged in a way that one has to be closed to the right side, the other one to the left. During acrobatic maneuvers the seat belt system should be tightened firmly.

7.9 CANOPY

The canopy of front and rear cockpit is manufactured in one section. The canopy can be manually operated and opened by lifting to the right.
In the open position there is a rod mounted to the fuselage rear of the canopy which must be connected to the canopy preventing slam down unintentionally.
Interior canopy locking handles, located on the left side on the canopy of each cockpit must be pulled together to unlock the canopy from the inside.
To open the canopy from the outside there are no separate handles, this means it must be opened by reaching through the small window (bad weather window) and proceed as mentioned above (interior opening).

7.10 POWER PLANT

7.10.1 ENGINE

The power plant consists of one Textron-Lycoming six-cylinder, horizontally opposed, air-cooled, direct drive, fuel injection engine type with inverted oil system.
The rated maximum T/O Power is 200 HP at 2700 RPM.

Engine specification:

Textron-Lycoming AEIO-360-A1E

For the present TBO refer to latest issue of Textron - Lycoming SERVICE LETTER No. L 201.

The following accessories are included in the power plant installation:

Fuel Injector:	Bendix
Magnetos:	Slick
Alternator:	Electrosystems
Starter:	Lycoming
Fuel pump:	Gates Lear

Shielded ignition system
Propeller governor drive
Transistor voltage regulator
Overvoltage relay

The engine is operated with the following manual controls:

Throttle control, dual
RPM control
Fuel mixture control

The propeller governor monitors the RPM automatically and prevents overspeeding. In the event that oil pressure is lost the propeller is automatically adjusted to coarse pitch in order to avoid overspeeding.

The use of 100/100LL aviation grade fuel (AVGAS 100) is the minimum grade recommended by the manufacturer of the AEIO-360-A-1E engine.

For continuous operation 115/145 aviation fuel is the maximum grade.

7.10.2 OIL SYSTEM

The oil is cooled by an oilcooler mounted on the left hand side in the engine compartment. The oil level is determined by a dip-stick.

A thermostatic valve is fitted upstream of the oil cooler. This valve ensures a quick warm-up of the oil after engine start.

Oil capacity and grades:

Oil:

Max sump capacity		8 qts.
Min sump capacity	Acrobatic	6 qts.
	Normal	4 qts.

For temperatures and oil grades refer to Section 1.7.

7.10.3 ENGINE INSTALLATION

The engine is supported by four shock mounts (type LORD or BARRY CONTROLS), to the tig-welded steel tube engine mount which is attached to the fuselage with four bolts on the firewall axis.

The engine cowling is divided into two parts, a lower and an upper part both made of glass-fibre/carbonfibre reinforced epoxy. The parts are fixed by a number of screws and the upper cowling has a separate hatch for easy access to the oil dip-stick.

7.10.4 PROPELLER

The standard propeller is a 3-blade wood composite, constant speed propeller type MTV-12-B-C/C183-17e. The propeller has a diameter of 1,83 m.

7.10.5 THROTTLE

Dual control (cub-type) mounted on the left side in cockpit.

7.10.6 MIXTURE

Vernier-control located at left side of rear cockpit (red knob).

7.10.7 RPM-CONTROL

Vernier-control on the left side of the rear cockpit.
Preselection of RPM possible due to constant speed governor (blue knob).

7.10.8 FUEL SELECTOR VALVE

Dual control. A rotary fuel selector valve is mounted behind the firewall on the right side of the fuselage. A torque tube connects the valve to both cockpit handles. Pull and turn the handle 90° to open the valve to the Acro & Center tank. A further 90° turn switches to the Wing tank fuel supply.

Position down: "OFF"
Position left: "ACRO & CENTER"
Position up: "WING TANK"

7.10.9 EXHAUST SYSTEM (OPTIONAL)

The EXTRA 200 can be equipped with a complete 4 in 1 System including silencer.

7.11 FUEL SYSTEM

The fuel system consists of two separate, independent tanks:

- "Acro & Center tank" in the fuselage
- "Wing tank" in the wing

Wing tank:

The root section of each wing in front of the main spars forms an integral fuel tank providing two interconnected tanks with 86 liters (22.7 US Gal.) total capacity. Each side of the wing has a 2" diameter filler cap for gravity refuelling. Due to the interconnection the fuel level of the left and right integral tank will equalize during refuelling within reasonable time. For max. fuel capacity, the first filled side have to be filled once again! Furthermore the wing tank is connected directly to the engine by the fuel selector valve. Unusable fuel is approximately 1 liter (0.3 Us Gal.).

Acro & center tank:

The Acro tank of 9 liters (2,3 US Gal.) is mounted under the Center tank. This Center tank of 27 liters (7.1 US Gal.) is mounted in front of the main spar of the wing. The Acro tank is connected with the Center tank in a gravity feed system. The Center tank has a 2" diameter filler cap for gravity refuelling. Unusable fuel is approximately 4 liters (1.1 US Gal.)

Adequate venting is provided in each tank by a main ventilation-tube, ending outside the fuselage at the right side.

Fuel pump

In addition to the engine driven fuel pump an electrically driven auxiliary fuel pump (boost pump) with by-pass, having sufficient capacity to feed the engine at takeoff power, is fitted as a safety device against failure of the engine-driven pump. The boost pump switch is located on the instrument panel.

A fuel filter with drain is installed between the fuel selector valve and the boost pump. Separate drains are located at the lowest point of each tank system.

Normal float type transducers and electrically operated fuel indicators are used.

7.12 ELECTRICAL SYSTEM

The electrical system is supplied by a 12 V alternator with rectifier, transistor voltage regulator. The alternator is mounted on and driven by the engine.

The field current is controlled by the voltage regulator to nominal 13.8 V under all load conditions. The masterswitch is located on the rear instrument panel.

Circuit protection against overvoltage is provided by the voltage regulator.

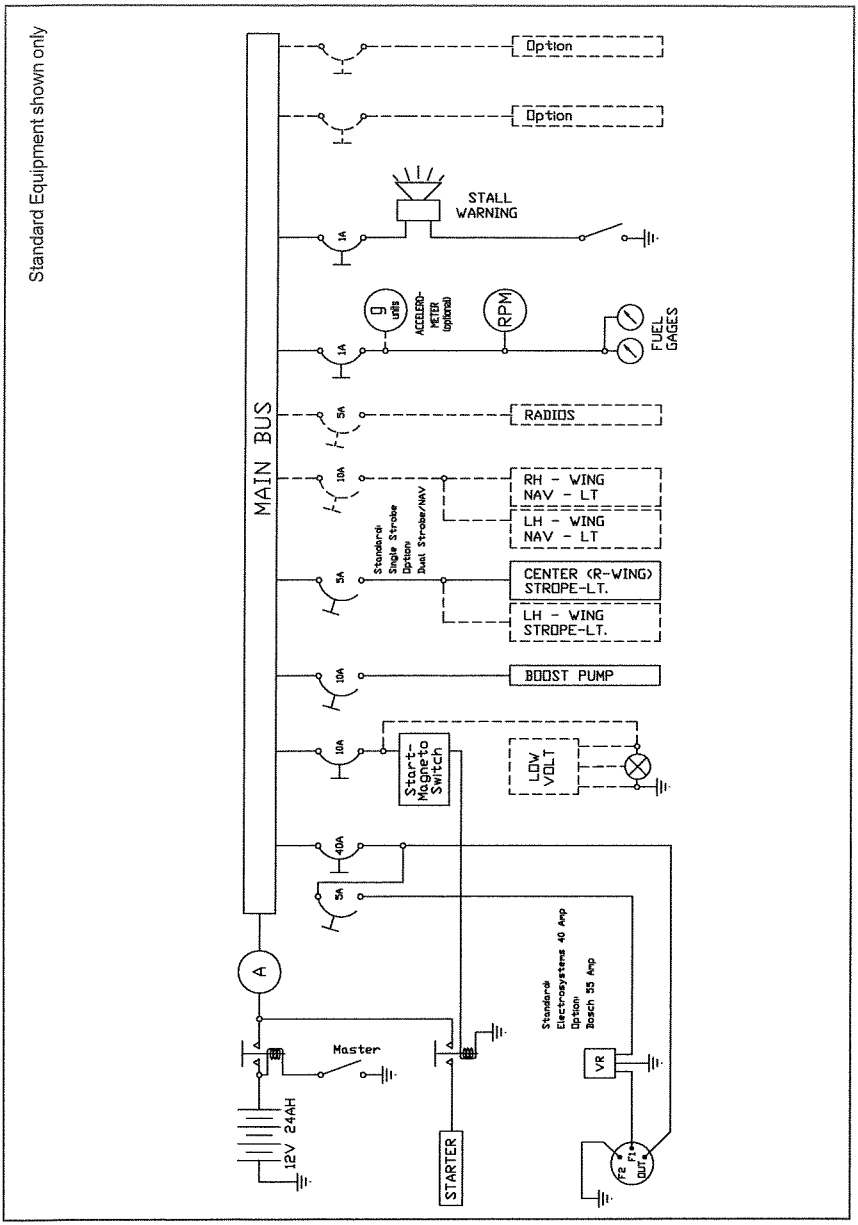
The maximum load taken from the alternator is 40 amp.

A 12 V leak proof battery is connected across the alternator output to stabilize the supply and to maintain all essential services in the event of an alternator failure and when the engine is not operating. The battery is mounted behind the firewall.

All electrical circuits are protected by circuit breakers located on the rear instrument panel and they are easily accessible to the pilot during flight.

The electrical system is adequately noise suppressed to ensure satisfactory operation of the radio equipment.

All wires, switches, circuit breakers etc. are manufactured to related aeronautical specifications.


Standard Equipment shown only
MAIN BUS
Option
Option
STALL WARNING
1A
g
ACCELERO-METER (optional)
RPM
FUEL GAGES
1A
5A
RADIOS
10A
RH - WING NAV - LT
LH - WING NAV - LT
Standard: Single Strobe
Option: Dual Strobe/NAV
5A
CENTER (R-WING) STROPE-LT.
LH - WING STROPE-LT.
10A
BOOST PUMP
10A
Start-Magneto Switch
LOW VOLT
40A
5A
A
Master
12V 24AH
STARTER
Standard: Electrosystems 40 Amp
Option: Bosch 35 Amp
VR
P2
P1
OUT

7.13 CABIN ENVIRONMENT CONTROL

A ventilation system in the canopy on the left side is provided for the supply of fresh air to the cabin. Left and right at the rear seat are eyeball-type adjustable vents.

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SECTION 8

HANDLING, SERVICING AND MAINTENANCE

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SECTION 8

HANDLING, SERVICING AND MAINTENANCE

8.1 INTRODUCTION

a) The airplane owner should establish contact with the dealer or certified service station for service and information.

b) All correspondence regarding the airplane must include its serial number which is stamped on a plate on the L/H rear part of the fuselage.

c) A service manual with revision service may be procured from the manufacturer.

8.2 AIRPLANE INSPECTION PERIODS

As required by national operating rules all airplanes must pass a complete annual inspection every twelve calendar months. In addition to the annual inspection airplanes must pass a complete inspection after every 100 flights hours with a minor check after 50 hours.

The Airworthiness Authority may require other inspections by the issuance of airworthiness directives applicable to the aircraft, engine, propeller and components. The owner is responsible for compliance with all applicable airworthiness directives and periodical inspections.

8.3 PILOT CONDUCTED PREVENTIVE MAINTENANCE

Pilots operating the airplane should refer to the regulations of the country of certification for information of preventive maintenance that may be performed by pilots. All other maintenance required on the airplane is to be accomplished by appropriately licensed personnel. Airplane dealer should be contacted for further information

Preventive maintenance should be accomplished with the appropriate service manual.

8.4 ALTERATIONS OR REPAIR

Alterations or repairs of the airplane must be accomplished by licensed personnel.

8.5 SERVICING

In addition to the airplane inspection periods (8.2) information for servicing the aircraft with proper oil and fuel is covered in Section 2 (Limitations) and Section 7 (Descriptions and Operation).

8.6 GROUND HANDLING

a) Due to its low weight and the free swivelling tail wheel two persons can easily move the airplane by hand.

b) To tie down the airplane M6 nut plates are provided in the wing tips where ring bolts can be screwed in. The tail wheel leg can be used as third point to tie down the airplane.
If the aircraft is parked in the open, it must be protected against the effects of weather, the degree of protection depending on severity of the weather conditions and the expected duration of the parking period. When the airplane is parked in good weather conditions for less than a half day park the aircraft headed into the wind and place wheel chocks at the main wheels.

c) To level the aircraft, the tail wheel is rested on a balance and jacked to a position that the fuselage reference line (upper fuselage stringer tube) is horizontal. There are two engine hoists provided on the top of the engine which can be used to lift the airplane with a crane. (Tail wheel resting on ground)

SECTION 9

SUPPLEMENTS
Doc-No. EA-07701.1

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9		Supplements	4 p.
9.2	☐	(Valid for SN 01 - 02 only)	10 p.
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902	☐	Reserved	0 p.
903	☐	Electronic Accelerometer	8 p.
904	☐	Emergency Locator Transmitter	10 p.
905	☑	External Power	6 p.
906	☑	Digital RPM Indicator	6 p.
907	☐	Long Range Wing Tank Capacity	6 p.
908	☐	Airtow Hook	8 p.
909	☐	Reserved	0 p.
910	☐	Reserved	0 p.
911	☐	Single Seat Canopy	4 p.
912	☐	FILSER TRT 600 Transponder	8 p.
913	☐	FILSER TRT 800 Transponder	8 p.
914	☐	GARMIN GTX 327 Transponder	6 p.
915	☐	GARMIN GTX 330 Transponder	8 p.
916	☐	BENDIX/KING KT 76A Transponder	6 p.
917	☐	BENDIX/KING KT 73 Transponder	8 p.
918	☐	BECKER ATC 2000 Transponder	6 p.
919	☐	BECKER ATC 4401 Transponder	8 p.
920	☐	BECKER BXP 6401 Transponder	10 p.

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9 SUPPLEMENTS

9.1 INTRODUCTION

Section 9 "Supplements" of the Pilot's Operating Handbook contains all information, necessary for a safe and efficient operation of the airplane when equipped with one or more of the various optional systems and equipment not provided with the standard airplane.

9.2 NOTES

The described systems and equipment are certified by the LBA for the *EXTRA 200.* Pages and contents of this section may not be exchanged and alterations of or additions to the approved contents may be made without the EXTRA Flugzeugproduktions- und Vertriebs-GmbH/LBA approval. The editor has the copyright of this Supplements and is responsible for edition of revisions. The log of effective pages is found under section 0.4 of this Pilot's Operating Handbook.

Each Supplement section (e.g. steerable tailwheel) covers only a single system, device, or piece of equipment and is a self-contained, miniature Pilot's Operating Handbook. The owner is responsible for incorporating prescribed amendments and should make notes about these on the records of amendments. It is responsibility of the pilot to be familiar with the contents of relevant supplements.

POH Supplements must be in the airplane for flight operations when the subject equipment is installed or special operations are to be performed.

The Table of Contents shows all EXTRA Supplements available for the EXTRA 200. A check mark in the *Section* column indicates that the corresponding supplement must be included in this POH.

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SECTION 901

STEERABLE TAIL WHEEL

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901.7 DESCRIPTION OF THE SYSTEM 901-3

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901 STEERABLE TAIL WHEEL

901.1 GENERAL

To improve taxi and handling quality, the EXTRA 200 can be equipped with an optional steerable tailwheel. The deflection angle of this tailwheel is arranged by the rudder control up to plus/minus 30°. Exceeding this deflection the tailwheel has a full-swivel capability by a release mechanism.

901.2 LIMITATIONS

The operation limitations are not affected due to the use of the steerable tailwheel.

901.3 EMERGENCY PROCEDURES

There is no change of basic emergency procedures with the installation of the steerable tailwheel.

901.4 NORMAL PROCEDURES

They are no changes for the described normal procedures after installation of the steerable tailwheel. In addition to the existing normal procedures the light precompression of connector springs and movement of the rudder have to be checked during the preflight check.

901.5 PERFORMANCE

Changes in flight performance due to installation of the steerable tailwheel are not noticeable. The given basic performance data under section 5 are still valid.

901.6 WEIGHT AND BALANCE

A change of the running empty weight and resulting C.G. position after installation of the steerable tailwheel is neglectable, because of minor differences in weight and C/G between standard and optional steerable tailwheel.

901.7 DESCRIPTION OF THE SYSTEM

The 5 inch tailwheel has a solid rubber tire and is rotatable by means of a wheelfork, which is connected to a bearing steelsleeve. This steelsleeve itself contains also the release mechanic, which gives the wheelfork a full-swivel capability exceeding plus/minus 30° deflection. The steelsleeve is glued into the glasfiberspring, which is bolted to the tail hardpoint of the aircraft. The steering of the tailwheel is accomplished by a direct mechanic link (rudder control cable) from the rudder pedals. The steering deflection of the tailwheel is controlled by the rudder movement and damped by anti shimmy connector springs.

901.8 HANDLING, SERVICING AND MAINTENANCE

During 50 hour inspection, the bearing steelsleeve has to be lubricated on the point of lubricating. Additionally all parts of the tailwheel have to be inspected visually for deformations, cracks and corrosion.

SECTION 905

EXTERNAL POWER

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905 External Power

905.1 GENERAL

The EXTRA 200 can be equipped with two versions of an optional external power receptacle system. The "normal" system (PN 93102.16-01) provides the capability to start the engine independent of the board battery and is limited to this use. The "continuous operation" system (PN 93102.16-02) further allows feeding the electrical system for longer periods.

905.2 LIMITATIONS

The operation limitations are not affected due to the installation of the external power receptacle system. For the location of the external power receptacle and protection of the electrical connection cable against overheating the following placard has to be attached on the rear instrument panel with an indicator arrow to the receptacle:

EXTERNAL POWER 12V
DO NOT CRANK FOR MORE THAN 10 SECONDS!
Allow 20 seconds to cool-down between attempts. Repeat up to 6 times.
Then let starter cool for 30 minutes.

905.3 EMERGENCY PROCEDURES

Not affected.

905.4 NORMAL PROCEDURES

The following starting procedures are recommended, however, the starting conditions may necessitate some variation from these procedures.

1. Perform Pre-flight inspection.
2. Set propeller governor control to "High RPM" position.
3. Open throttle approximately 1/4 travel.
4. **Master switch "OFF"**
5. Put the external power plug into the board receptacle
6. Turn boost pump "ON".
7. Move mixture control to "FULL RICH" until a slight but steady fuel flow is noted (approximately 3 to 5 seconds) and return mixture control to "IDLE CUT-OFF".

   Turn boost pump "OFF".

CAUTION

Pay attention to objects and persons in the propeller operating area!
Hold the canopy tight!

8. Apply the brakes.
9. Engage starter.
10. When engine fires release the ignition switch back to "BOTH".
11. Pull the external power plug from the board receptacle.
12. Move mixture control slowly and smoothly to "FULL RICH".

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905.8 HANDLING, SERVICING AND MAINTENANCE

Not affected.

13. Check the oil pressure gauge. If minimum oil pressure is not indicated within 30 seconds, shut off the engine and determine trouble.
14. **Master switch "ON".**

905.5 PERFORMANCE

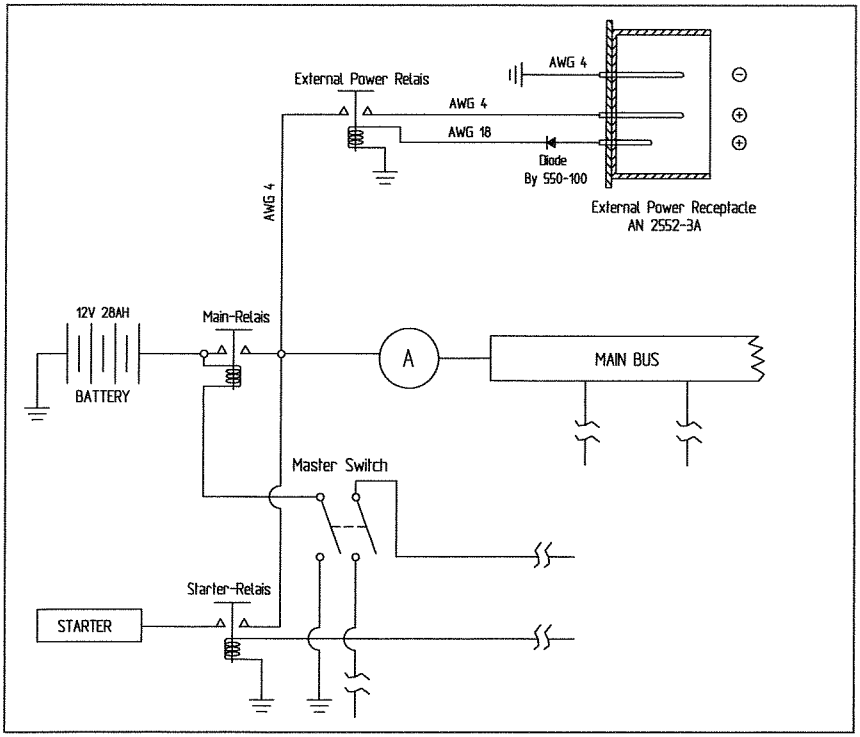
Not affected.

905.6 WEIGHT AND BALANCE

Not affected.

905.7 DESCRITPTION OF THE SYSTEM

The external power receptacle is attached left under the rear seat.
The main-relais is located at the left side of the firewall, above the starter-relais. For the avoidance of sparks, this relais does not switch before a safe contact from plug to receptacle is ensured. During the engine start, the master switch has to be switch in **"OFF"**-position for the disconnection of the battery from the aircraft electric circuit.



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SECTION 906

DIGITAL RPM INDICATOR

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906 DIGITAL RPM INDICATOR

906.1 GENERAL

The EXTRA 200 can be equipped with the optional "P-1000" Digital RPM indicator alternativ to the mechanical VDO - RPM indicator.

906.2 LIMITATIONS

The operation limitations are not affected due to the installation of the "P-1000" Digital RPM indicator. The face of the indicator is placarded with the unchanged Engine RPM operating range. Additional the operating RPM ranges are indicated on the large green, yellow, and red LEDs. This LEDs are located on the upper right corner of the indicator face.

Green	Yellow	Red
2500 / 700	2700 / 2500	3500 / 2700

906.3 EMERGENCY PROCEDURES

Not affected.

906.4 NORMAL PROCEDURES

The *Normal Procedures* have to be changed in Chapter "4.5 *Take-Off Procedure"* section "4.5.1.*Before take off"*. If the P-1000 RPM indicator is installed, the mag-drop test has to be carried out in the following manner:

Magneto check

Engine RPM: 1800 min^{-1}

Pay attation to the three small LEDs in the "Status" area on the upper left corner of the P-1000 face:

Ignition switch position: LEFT
Status area: Left red LED illuminates
Display: shows RPM drop

Ignition switch position: RIGHT
Status area: Right red LED illuminates
Display: shows RPM drop

Ignition switch position: BOTH
Status area: Right and left red LED off illuminate
The middle LED is not allowed to alert, otherwise the difference is more than permissible.

NOTE

During the short circuit (grounding) of a single magneto, the respective red LED has to be illuminated. The maximal allowed RPM drop at 1800 min^{-1} is 175 min^{-1}. The maximal difference between the magnetos must not exceed 50 RPM (identify with the illuminated yellow LED.)

906.5 PERFORMANCE

Not affected.

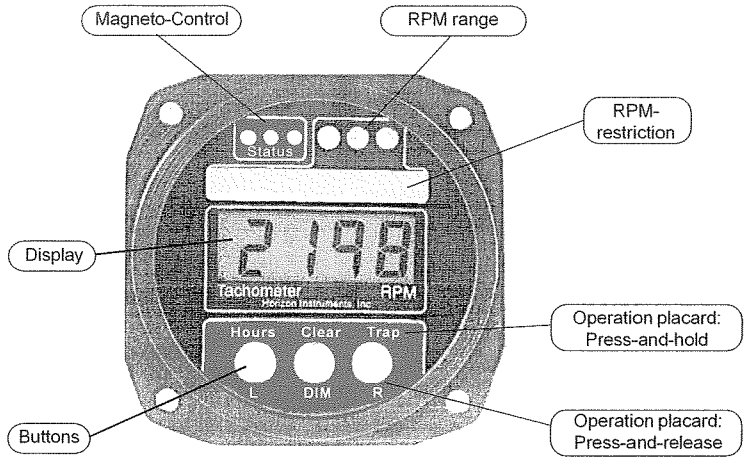
906.6 WEIGHT AND BALANCE

Not affected.

906.7 DESCRIPTION OF THE SYSTEM

The operation of the indicator is straight-forward. After power is supplied to the indicator, the engine is started, and the self tests are performed, the default display of the engine RPM appears on the display. The default display is insured via the use of internal timers that will restore the display to the current RPM even in the event that one of the panel buttons becomes stuck or defective.

Internally, two independent tachometers watch the pulses received from each magneto. Each tachometer is accurate to less than 1 RPM and can be individually enabled/disabled via buttons on the face of the indicator.



RPM RANGES

Engine operating ranges are indicated on the large green, yellow, and red LEDs.
This LEDs are located on the upper right corner of the indicator face.

MAGNETO-CHECK

Three small LED magneto system alert indicator lights are located within the "Status" area on the upper left corner of the indicator face.
The left and right red led alert indicator lights, when illuminated, indicate, because of loss of ignition signal to the tachometer, a possible malfunction of the respective left or right

magneto ignition system.
While performing a magneto check during engine run-up, the red alert indicator lights will illuminate, thus identifying the grounding of the respective right or left magneto systems.

Ignition switch	Tachometer Magneto	
	Right	Left
Both OFF	ON	ON
Left ON, Right OFF	OFF	ON
Right ON, Left OFF	ON	OFF
Both ON	OFF	OFF

Between the left and right magneto ignition system alert indicators is a yellow **RPM Synchronization indicator**. This small yellow indicator is illuminated when there is a difference of more than 50 RPM between the right and left tachometers.

This indicator also may flicker during extreme RPM excursions of the engine.

OPERATION BUTTONS

There are three panel buttons. Each button has two modes of operation.

PRESS-AND-HOLD operation mode

(press and hold for more than 2/3 of a second)

This operation mode is placarded above each button. (Hours, *Clear, Trap)*

Engine time *(Hours)*
The left button, upon depression, will cause the tachometer to display the non-fractional portion (0000.) of the current accumulated engine hours. When the button is released, the fractional part of the engine hours (.00) is displayed for a short period of time. The clock is started whenever the engine RPM exceeds 800 RPM and is recorded in real hours.

Clear (Clear)
The middle button clears the RPM trap. During depression of the switch, the RPM trap is zeroed. When the button is released, the trap will record the current engine RPM.

Engine RPM (Trap)
The right button will cause the tachometer to display the current contents of the RPM trap. This trap records the **highest engine RPM** achieved before the button was pressed.

PRESS-AND-RELEASE operation mode

(press and release in less than 2/3 of a second)

This operation mode is placarded below each button.*(L , DIM , R)*

Masks *(L , R)*
During normal operation, the tachometer presents the average of the left and right internal tachometers on the display. However, a mechanism **exists to mask** either tachometer from the display, leaving the remaining tachometer to determine magneto/ignition problems.

Quickly pressing and releasing the left button *(L)*, causes the tachometer to mask the left tachometer.

Quickly pressing and releasing the right button *(R)*, causes the tachometer to mask the right tachometer.

Dimmer *(DIM)*
Quickly pressing and releasing the middle button *(DIM)*, causes the tachometer to alternately dim or brighten the LED indicators (except the large red LED of the RPM Range).

906.8 HANDLING, SERVICING AND MAINTENANCE

Not affected.