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 TITLE: **AW139 – AMPI CHAPTER 4 ISSUE 9**
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N.A.		<i>L. Castaldi</i>	<i>13/2/18</i>	N.A.		N.A.	
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Systems	date	Diagnostics	date	Struct., FFC & Land. Syst	date	Software & CEH	date
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Rotors & Rotating Controls	date	Drive Systems	date	Flight	date	Lightning Protection (D.E.)	date
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N.A.		N.A.				<i>G. Bertolami</i>	<i>19/2/18</i>

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AWE009/02 Issue E

**LISTA DELLE REVISIONI - REVISION HISTORY**

REVISIONE ISSUE	DESCRIZIONE MODIFICHE CHANGE DESCRIPTION	DATA EMISSIONE ISSUE DATE
A N. PAG. 7 + 26 pages attached to Annex A	<i>First Issue</i>	13/02/2018

**DISTRIBUTION LIST**

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1 SCOPE

This report is issued to include the MAF (Manual Approval Form) information as referred in [1].

2 REFERENCES

- [1] C750-02-003.1 – DOA Handbook
- [2] AWEOP002 – Signature Procedure For Technical Documentation
- [3] 100-50-175 Compliance Verification Engineers (CVE) and Airworthiness Engineer (AE): Tasks Responsibilities, Operating Procedures

3 MANUAL APPROVAL FORM

The information pertinent to the MAF are reported in Annex A of this document. This report is issued in conformity with Ref. [2] with the clarification that both Area Approval and Authorization signatures have been undertaken by Authorization (CPE) approval.

The signatures for the technical content of the document AMPI CHAPTER 4 ISSUE 9 (reported in Annex A) for the responsibilities defined in [3] according to [1] are those reported on the first page of this document.

4 COMPLIANCE ASSESSMENT

The scope of this report is to provide compliance with the JAR 29 Amdt.3 requirement listed below:

JAR 29.1529 — Instructions for Continued Airworthiness".
Appendix A — Instruction for Continued Airworthiness
(Paragraph A29.1, A29.2 and A29.4



5 CERTIFICATION PROGRAM AND APPLICABILITY

This report is in accordance with the certification program as indicated in the following NDC:

- ✓ NDC 139G6220-008 - M/R DAMPER - IMPROVEMENT OF THE INSPECTION PROGRAM
- ✓ NDC 139G0255-035 - AMPI CHAPTER 4 - ISSUE 9

Note: the responsibility of the CVEs for the signature of this document is reported in the applicable CCLs.

6 CHANGE DESCRIPTION

The following change will be introduced in Chapter 4 issue 9:

Retirement lives

- ✓ The retirement life of the M/R damper Body End Assy P/N M006-01H046-041 every 2400 FH. In addition to this, a note will be added to specify that the M/R damper P/N 3G6220V001353 must be sent to the manufacturer when the M/R damper Body End Assy P/N M006-01H046-041 reaches 2400 FH.

Note that according to the results of the Compliance Statement 139G6220-008/02, a fatigue life of 39,000 FH has been demonstrated for the M/R damper P/N 3G6220V001353. LHD, on voluntary basis, decided to report in the AMPI Chapter 4 a retirement life of 2400 FH to align the Life of the damper with the Overhaul currently published in AMPI Chapter 5.

- ✓ The retirement life of the M/R blade P/N 3G6410A00131 will be extended from 16,800 FH or 50,200 landings to 20,000 FH or 50,200 Landings retaining the Rotor Starting or Stopping with wind speed above 33 kts - Life penalty factor;
- ✓ The retirement life of the M/R blade P/N 3G6410A00134 will be deleted;
- ✓ The retirement life of the TR Hub P/N 3G6420A00332 will be extended from 11,250 FH or 67,500 landings to 34,500 FH or 207,000 landings;
- ✓ The retirement life of the MGB Upper Fitting P/N 3G6330A00232 will be extended from 95,600 Landings to 133,240 Landings and the penalty factors for operation above 6400 Kg has been removed;
- ✓ Table 7 “*Operation above 6400 Kg (MTOW) – Life penalty factor*” will be modified to remove the RL024 “*MGB upper Fitting*” and to add RL021A (typo) “*MGB case, Top Assembly (with Circular Force Active Vibration Control System installed)*”.



Mandatory Inspections

- ✓ The Note 2 applicable to the inspections MI62-01 and MI64-01 will be changed to modify the wind speed value from 27 Kts to 33 Kts;
- ✓ The Note 7 will be changed to correct a typo. In particular the P/N 3G5306P22311 will be replaced by the P/N 3G5306P08513.



ANNEX A

TITLE

Manual Approval Form

Retirement lives

General

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References

Table 1 References

Data module/Technical publication	Title
No references	

Description

1 Introduction

The parts listed in the Table 2 - Retirement lives - List of affected parts must be mandatorily retired from service when the indicated retirement life is reached. This also applies to all those non-serialized standard parts which connect the identified assembly / component.

All retirement lives are expressed in Flying Hours (FH), unless otherwise specified. Flying hours are defined as those hours accumulated from take-off to landing.

The retirement lives of some parts are expressed in “landings” because their life is dependent upon the rotor start-stop cycles and the helicopter ground-air-ground cycles.

Note

If the number of landings applicable to the component is not known throughout the entire service life of the component, the number of landings shall be calculated multiplying the component flight hours by a factor of 6.

If not differently specified, the retirement lives are based on the following assumptions:

- 800 landings in 100 flying hours, including 400 rotor start-stop cycles.
- 20 percent of usage in CAT. A Training Operations, on yearly basis.
- For External Load Operations: 6 External Load Cycles and 3 Landings per Flying Hour, calculated on at least 100 FH basis
- For External Hoist Operations: 1.65 Lift per Flying Hour, calculated on at least 100 FH basis.

In the event that the actual usage exceeds these assumptions, the Operator shall contact the Manufacturer.

If parts with the same part number have been interchanged between different AW139 variants and different AW Helicopter models, the retirement life of the part must be restricted to the lowest value between those variants/models on which the part has been installed.

Note 1

For the following parts the Operator shall maintain the relevant log card and contact the Manufacturer prior to approaching the 40000 FH for confirmation of the applicability of the airworthiness instructions provided in this chapter:

- Fuselage Assembly, P/N 3G5300X00132
- Fuselage Structure Assembly, P/N 3P5300A00131
- Tail Assembly, P/N 3G5350A00135 plus retromod P/N 3G5309P02711
- Tail Assembly, P/N 3G5350A00136.

Note 2

For the following Tail Fin Assemblies the life limit is aligned with the life limit applicable to the corresponding Tail Assembly (refer to the related Log Card):

- Tail Fin Assembly, P/N 3G5351A00133 part of the Tail Assembly P/N 3G5350A00132
- Tail Fin Assembly, P/N 3G5351A00134 part of the Tail Assembly P/N 3G5350A00133 or 3G5350A00134
- Tail Fin Assembly, P/N 3G5351A00135 part of the Tail Assembly P/N 3G5350A00135
- Tail Fin Assembly, P/N 3G5351A00136 part of the Tail Assembly P/N 3G5350A00136.

2 Retirement lives - Affected parts

Refer to Table 2 - Retirement lives - List of affected parts.

3 Terms and definitions of Table 2

3.1 Reference (Ref)

This column gives a reference [RL (Retirement Lives) plus a progressive number] which identifies the listed components.

3.2 Part

This column gives the part description.

3.3 Part number

This column gives the part number.

3.4 Retirement life

This column gives the retirement life of the part. Unless specified differently, the retirement life is in flight hours.

Table 2 Retirement lives - List of affected parts

Ref	Part	Part number	Retirement life
RL001	Left Main Landing Gear Assembly (Liebherr landing gear installation)	3G3210V00133	21000 landings
	Left Main Landing Gear Assembly (Liebherr landing gear installation)	3G3210V00135	50000 landings
	Deleted		
	Right Main Landing Gear Assembly (Liebherr landing gear installation)	3G3210V00233	21000 landings
	Right Main Landing Gear Assembly (Liebherr landing gear installation)	3G3210V00235	50000 landings
	Deleted		
RL001A	Left Main Landing Gear Assembly (Liebherr landing gear installation)	3G3210V00831	25000 landings
RL001C	Left Main Landing Gear Assembly (Liebherr landing gear installation)	3G3210V00832	50000 landings
RL001B	Right Main Landing Gear Assembly (Liebherr landing gear installation)	3G3210V00931	25000 landings
RL001D	Right Main Landing Gear Assembly (Liebherr landing gear installation)	3G3210V00932	50000 landings
RL002	Nose Landing Gear Assembly (Liebherr landing gear installation)	3G3220V00133	50000 landings
	Deleted		
RL002A	Nose Landing Gear Assembly (Goodrich landing gear installation)	4G3220V00131	50000 landings
RL003	Main Retracting Actuator Assembly (Liebherr landing gear installation)	3G3230V00332	50000 landings
RL003A	Main Retracting Actuator Assembly (Goodrich landing gear installation)	4G3230V00233	50000 landings
RL004	Nose Retracting Actuator Assembly (Liebherr landing gear installation)	3G3230V00132	50000 landings

Table 2 Retirement lives - List of affected parts

Ref	Part	Part number	Retirement life
RL004A	Nose Retracting Actuator Assembly (Goodrich landing gear installation)	4G3230V00131	50000 landings
RL004B	Left MLG Trailing Arm Assembly (Goodrich landing gear installation)	4G3210V00331	50000 landings
RL004C	Right MLG Trailing Arm Assembly (Goodrich landing gear installation)	4G3210V00431	50000 landings
RL004D	MLG Shock Absorber Assembly (Goodrich landing gear installation)	4G3210V00531	50000 landings
RL004E	Forward Arm Assembly (Goodrich landing gear installation)	4F3210A00131	50000 landings
RL005	TGB Fitting	3G5351A01151	7200
RL006	Tail Assembly	3G5350A00132	19600 FH or 32000 landings whichever comes first (Para 5) (Note 8) (Note 12)
RL006A	Tail Assembly	3G5350A00136	See Note 1 Para 1
RL006B	Tail Assembly	3G5350A00135	32000 landings (Note 9)
RL007	Main Landing Gear LH Segment	3P5339A00332	80000 landings
	Main Landing Gear RH Segment	3P5339A00432	80000 landings
RL007A	Rear Fuselage Assembly	3P5340A00132	32000 landings (Note 10) (Note 11)
RL007B	Trailing Arm Pintle Pin (Goodrich landing gear installation)	3G5339A08351	37500 landings
RL008	Tailplane Attachment Fitting on Fin Upper	3G5351A01432	7600
	Tailplane Attachment Fitting on Fin Upper	3G5351A01532	7600
RL009	Tailplane Attachment Fitting on Fin Lower	3G5351A07531	10600
RL009A	Tailplane upper LH fitting	3G5309P03031	7600
RL009B	Tailplane upper LH fitting	3G5351A08931	7600
RL009C	Tailplane upper RH fitting	3G5309P03131	7600
RL009D	Tailplane upper RH fitting	3G5351A09031	7600
RL009E	Tailplane lower fitting	3G5351A09131	10600
RL010	Deleted		
RL010A	Rod End (Fin End)	3G5510A03931	4600 (Para 6)
RL010B	Rod Sleeve	3G5510A04152	10000

Applicable to: All

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Table 2 Retirement lives - List of affected parts

Ref	Part	Part number	Retirement life
RL010C	Rod End (Tailplane End)	3G5510A04052	10000 (Para 6)
RL010D	Tailplane Assembly	3G5510A00133	65000 (Para 6)
RL010E	Rod assembly (Tailplane)	3G5510A08931	4600 (Para 6)
RL011	Main Landing Gear Actuator Bolt on Sponson Bracket	3G3210L00151	160000 landings
RL012	Main Rotor Blade Assembly	3G6210A00131	20000 FH or 50200 landings whichever comes first (Table 4) (Note 4)
RL012A	Main Rotor Blade Assembly with Heater Mat	4G6210A00132	10000 FH or 50200 landings whichever comes first (Table 4) (Note 5)
RL012B	Deleted		
RL013	Main Rotor Hub Assembly	3G6220A00332	46000 (Para 5)
RL014	Deleted		
RL015	Main Rotor Rotating Swashplate	3G6230A00332	4800 (Para 4)
RL015A	Main Rotor Rotating Swashplate Assy reworked	3G6206P01331	4800 (Para 4)
RL015B	Main Rotor Rotating Swashplate	3G6230A00333	17300 (Para 4)
RL015C	Locking Ring Outer	3G6230A03153	7700
RL016	Main Rotor Scissors Attachment Flange Assembly	3G6220A00632	47450
RL016A	Main Rotor Scissors Attachment Flange Assembly	3G6220A00633	Unlimited
RL017	Main Rotor Damper Bolt	3G6220A01251	51700
RL018	Main Rotor Elastomeric Bearing	3G6220V00153	100000 landings
	Main Rotor Elastomeric Bearing	3G6220V00154	100000 landings
RL019	Upper Scissor Lever Assembly	3G6230A00832	14300
RL019A	Deleted		
RL019B	Lower Scissor Lever Assembly	3G6230A00931	20100
RL019C	Upper Scissor Lever Assembly	3G6230A00834	Unlimited
RL019D	Lower Scissor Lever Assembly	3G6230A00934	Unlimited
RL020	Deleted		
RL021	MGB Case, Top Assembly (without Circular Force Active Vibration Control System installed)	3G6320A01033	8800 FH or 23000 landings whichever comes first (Para 6) (Note 7)
	MGB Case, Top Assembly	3G6320A01035	120000 landings

Applicable to: All

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Table 2 Retirement lives - List of affected parts

Ref	Part	Part number	Retirement life
RL021A	MGB case, Top Assembly (with Circular Force Active Vibration Control System installed)	3G6320A01033	7600 FH or 20000 landings whichever comes first (Para 6) (Note 7)
RL021B	MGB case, Top Assembly	3G6320A16231	120000 landings
RL022	MGB Shaft, Main Rotor	3G6320A01852	15300 FH or 54000 landings whichever comes first (Table 3) and (Para 4) and (Para 5)
RL023	MGB Case, Main Assembly	3G6320A02734	5000 FH or 27000 landings whichever comes first (Para 5)
	MGB Case, Main Assembly	3G6320A02736	12500 FH or 61800 landings whichever comes first (Para 5)
RL023A	MGB Case, Main Assembly	3G6320A16331	12500 FH or 61800 landings whichever comes first (Para 5)
RL024	MGB Upper Fitting	3G6330A00232	133240 landings
RL025	MGB Antitorque Beam	3G6330A00532	20000 FH or 63000 landings whichever comes first (Para 5)
RL026	Deleted		
RL027	Deleted		
RL028	Deleted		
RL029	Deleted		
RL030	Deleted		
RL031	Deleted		
RL032	Deleted		
RL033	Tail Rotor Blade Assembly	3G6410A00131	10000
RL033C	Tail Rotor Blade Assembly	3G6410A00132	1200 FH or 3200 landings whichever comes first
RL033D	Tail Rotor Blade Assembly	3G6410A00133	40000 landings
RL033A	Tail Rotor Blade Assembly with Heater Mat	4G6410A00131	4033
RL033E	Tail Rotor Blade Assembly with Heater Mat	4G6410A00132	1200 FH or 3200 landings whichever comes first
RL033F	Tail Rotor Blade Assembly with Heater Mat	4G6410A00133	4033 FH or 40000 landings whichever comes first
RL033G	Tail Rotor Blade Damper bracket	3G6410A03631	27600
RL033B	Tail Rotor Blade Bolt	3G6420A00252	19100 (Note 2)

Applicable to: All

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Table 2 Retirement lives - List of affected parts

Ref	Part	Part number	Retirement life
RL034	Tail Rotor Hub Assembly	3G6420A00332	34500 FH or 207000 landings whichever comes first
RL035	Tail Rotor Elastomeric Spherical Bearing	3G6420V00153	5000
	Tail Rotor Elastomeric Spherical Bearing	3G6420V00154	5000
RL036	Tail Rotor Elastomeric Damper	3G6420V00455	20000
RL036A	Tail Rotor Elastomeric Damper	3G6420V00851	12600
RL037	Tail Rotor Shaft Drive Assembly, Number 1	3T6510A00141	100000 landings (Para 5)
	Tail Rotor Shaft Drive Assembly, Number 2	3T6510A00238	16000 FH or 44800 landings whichever comes first (Para 5) (Para 6)
	Tail Rotor Shaft Drive Assembly, Number 3	3T6510A00337	16000 FH or 44800 landings whichever comes first (Para 5) (Para 6)
RL038	Tail Rotor Drive Coupling	3T6510V00152	60000 landings (Para 5)
RL039	Deleted		
RL040	TGB Input Flange Coupling	3T6521A07353	100000 landings (Para 5)
RL041	IGB Output Flange Coupling	3T6521A07353	100000 landings (Para 5)
RL042	TGB Output Housing	3T6522A05338	48000 landings (Para 5)
RL042A	TGB Output Housing	3T6522A05341	100000 landings (Para 5)
RL043	TGB Centre Housing	3T6522A05144	10000 FH or 72200 landings whichever comes first
RL043A	TGB Centre Housing	3T6522A05147	23000 FH or 94200 landings whichever comes first (Para 5)
RL044	IGB Input Flange Coupling	3G6521A00231	100000 landings (Para 5)
	IGB Input Flange Coupling	3T6521A05657	100000 landings (Para 5)
RL045	IGB Housing assy centre	3T6521A05134	100000 landings (Para 5)
RL046	Deleted		

Table 2 Retirement lives - List of affected parts

Ref	Part	Part number	Retirement life
RL047	External (Rescue) Hoist Cable Assembly	BL-6260	1500 external hoist lifts or 4 years, whichever occurs first (see Para 4 for External Hoist Lift definitions)
	External (Rescue) Hoist Cable Assembly	BL-9149-8	1500 external hoist lifts or 4 years, whichever occurs first (see Para 4 for External Hoist Lift definitions)
	External (Rescue) Hoist Cable Assembly	42325-298	1500 external hoist lifts or 4 years, whichever occurs first (see Para 4 for External Hoist Lift definitions)
RL048	Antitorque Beam Bolts	3G6330V00451 3G6330L00351	56000
RL049	Deleted		
RL050	Deleted		
RL051	Deleted		
RL052	Tail Rotor Servoactuator	3G6730V00731	44000 (Note 6) (Para 5)
RL053	Main rotor damper body end assembly	M006-01H046-041	2400 (Note 13)

Notes

- 1 The specified limit is intended since first opening of the sealed bag.
- 2 Record the effective accumulated life of the Tail Rotor Blade Bolt P/N 3G6420A00252 on applicable Log Card at receiving date of Chapter 4 Issue 3. Otherwise, if the effective accumulated life is unknown record a forfeit accumulated life of 9000 FH. This is applicable to all bolts including those in stock.
- 3 Deleted.
- 4 If the number of landings is not known throughout the entire service life of the Main Rotor Blade Assembly P/N 3G6210A00131, the number of landings shall be calculated multiplying the blade flight hours by a factor of 4.
- 5 If the number of landings is not known throughout the entire service life of the Main Rotor Blade Assembly P/N 4G6210A00132, refer to the note reported at Para 1 of AMPI CHAPTER 04 AIRWORTHINESS LIMITATIONS - Retirement lives- General (39-A-04-10-00-00A-028E-P).
- 6 Record the effective accumulated life of the Tail Rotor Actuator P/N 3G6730V00731 on applicable Log Card at receiving date of Chapter 4 Issue 3 Change 5. Otherwise, if the effective accumulated life is unknown record a forfeit accumulated life of 11000 FH. This is applicable to all Tail Rotor Actuator including those in stock.
- 7 If the part is once installed on an AW139 helicopter equipped with the Circular Force Active Vibration Control System, the applicable life limit is restricted permanently to 7600 FH or 20000 landings.

Table 2 Retirement lives - List of affected parts

Ref	Part	Part number	Retirement life
8	When retromod P/N 3G5309P02711 and P/N 3G5309P01812 are applied to the Tail Assembly P/N 3G5350A00132, 3G5350A00133 or 3G5350A00134, the retirement life is extended to 19600 FH or 78600 landings, whichever comes first. The External Load Operation penalty factor is retained.		
9	When retromod 3G5309P02711 is applied to the Tail Assembly P/N 3G5350A00135, the life limit is not applicable but the Operator shall maintain the relevant log card and contact the Manufacturer prior to approaching the 40000 FH for confirmation of the applicability of the airworthiness instructions provided in this chapter.		
10	When retromod 3G5306P38811 is applied to Rear Fuselage 3P5340A00132 or 3P5340A00133, the life limit of 32000 landings is superseded. The Operator shall maintain the relevant log card and contact the Manufacturer prior to approaching the 40000 FH for confirmation of the applicability of the airworthiness instructions provided in this chapter.		
11	If Rear Fuselage 3P5340A00134 or 3G5340A11131 is installed, the life limit of 32000 landings is superseded. The Operator shall maintain the relevant log card and contact the Manufacturer prior to approaching the 40000 FH for confirmation of the applicability of the airworthiness instructions provided in this chapter.		
12	When retromod P/N 3G5309P01511 is applied before having accumulated 12500 FH or 50000 landings, whichever comes first, to the Tail Assembly P/N 3G5350A00132, 3G5350A00133 or 3G5350A00134, equipped with retromod P/N 3G5309P02711 and 3G5309P01812, the retirement life is extended to 33700 FH. The External Load Operation penalty factor is retained		
13	For main rotor damper body end assembly replacement, send P/N 3G6220V01353 to the Manufacturer.		

Table 3 CAT. A Training Operations - Life penalty factor

Ref	Part	Part number	Life penalty factor
RL012	Deleted		
RL012A	Deleted		
RL012B	Deleted		
RL022	MGB Shaft, Main Rotor	3G6320A01852	1.7
RL031	Deleted		

Note**CAT. A Training Operations**

- For the parts listed in Table 3 - CAT. A Training Operations - Life penalty factor a life penalty must be mandatorily applied whenever CAT. A Training Operations are performed.
- Flying hours (FH) spent in CAT. A Training must be multiplied by the specified factor. The penalty is applicable only to flying hours.
- FH spent in CAT. A Training Operations are defined as the total flying Hours (from take off to landing) when at least one time the training mode is activated.
- E.g.: MGB Shaft, Main Rotor: Flight time spent during CAT. A Training Operations = 1 FH. Total accumulated FH = 1 x 1.7 = 1.7 FH.

Table 4 Rotor Starting or Stopping with wind speed above 33 kts - Life penalty factor

Ref	Part	Part number	Life penalty factor 33 kts<wind speed<45 kts	Life penalty factor 45 kts<wind speed<60 kts
RL012	Main Rotor Blade Assembly	3G6210A00131	3 landings	14 landings
RL012A	Main Rotor Blade Assembly with Heater Mat	4G6210A00132	3 landings	14 landings
RL012B	Deleted			

Note**Rotor Starting or Stopping with wind speed above 33 kts**

- For the parts listed in Table 4 - Rotor Starting or Stopping with wind speed above 33 kts - Life penalty factor a life penalty must be mandatorily applied.
- Increase the landings by the specified value for each Rotor Start or Stop carried out with wind speed above 33 kts.
- E.g.: Main Rotor Blade Assembly part number 3G6210A00131:
Rotor Start with 33 kts<wind speed<45 kts. Flight Time = 1 FH.
Rotor Stop with 33 kts<wind speed<45 kts. Landings = 1
Total accumulated FH = 1 FH.
Total consumed landings = 3 + 1 + 3 = 7 landings.
- E.g.: Main Rotor Blade Assembly part number 3G6210A00134:
Rotor Start with 45 kts<wind speed<60 kts. Flight Time = 1 FH.
Rotor Stop with 45 kts<wind speed<60 kts. Landings = 1.
Total accumulated FH = 1 FH.
Total consumed landings = 14 + 1 + 14 = 29 landings.
- E.g.: Main Rotor Blade Assembly part number 4G6210A00132:
Rotor Start with 33 kts<wind speed<45 kts. Flight Time = 1 FH.
Rotor Stop with 45 kts<wind speed<60 kts. Landings = 2.
Total accumulated time FH = 1 FH.
Total consumed landings = 3 + 2 + 14 = 19 landings.

4 External Hoist Operation

For the parts listed in Table 5 - External Hoist Lift - Life penalty factor a life penalty must be mandatorily applied whenever an external hoist lift is performed.

The external hoist lift is defined as an unreeling and recovery of the cable with a load attached to the hook, independent of the length of the cable that is deployed/recovered. An unreeling/recovery of the cable with no load on the hook is not considered to be a lift. Any operation where a load is applied for half the operation (i.e. unreeling or recovery) must be considered as one lift.

Increase the flying hour by the specified value for each external hoist lift. The penalty is applicable only to flying hours.

Table 5 External Hoist Lift - Life penalty factor

Ref	Part	Part number	Life penalty factor
RL015	Main Rotor Rotating Swash-plate	3G6230A00332	0.5 (Note 1)
RL015A	Main Rotor Rotating Swash-plate Assy reworked	3G6206P01331	0.5 (Note 1)
RL015B	Main Rotor Rotating Swash-plate	3G6230A00333	0.5 FH
RL022	MGB Shaft, Main Rotor	3G6320A01852	Note 2

E.g.: **Main rotor rotating swashplate**: Flight time = 4 FH. Number of rescue hoist lifts = 3. Total accumulated FH = $4 + (0.5 \times 3) = 5.5$.

Notes

- For non zero-time components, whose accumulated life has been counted without this penalty factor, the total accumulated life must be updated as follows:

Updated total accumulated flight hours = present total accumulated flight hours + $0.5 \times$ accumulated external hoist lifts.
- The life penalty factor due to external hoist operations has been deleted. For non zero-time components, whose accumulated life has been counted with the previous penalty factor, the total accumulated life may be updated as follows:

Updated total accumulated life = present total accumulated life - $2.5 \times$ accumulated external hoist lifts.

Present total accumulated life = 25 FH.

Accumulated external hoist lifts = 5.

Updated total accumulated life = $25 - 2.5 \times 5 = 12.5$ FH.

5 External Load Operation

For the following parts in Table 6 - External Load Operation - Life penalty factor a life penalty must be mandatorily applied whenever an External Load Cycle is performed.

An External Load Cycle is every external load lift using the Cargo Hook.

Increase the Flying Hours and/or Landings by the specified value, as applicable, for each External Load Cycle.

Table 6 External Load Operation - Life penalty factor

Ref	Part	Part number	Life penalty factor
RL006	Tail Assembly	3G5350A00132	0.5 FH
RL013	Main Rotor Hub Assembly	3G6220A00332	1 FH
RL022	MGB Shaft, Main Rotor	3G6320A01852	2 Landings and 1 FH
RL023	MGB Case, Main assembly	3G6320A02734	2 Landings
	MGB Case, Main assembly	3G6320A02736	2 Landings
RL023A	MGB Case, Main Assembly	3G6320A16331	2 Landings
RL025	Antitorque Beam	3G6330A00532	2 Landings and 1 FH
RL037	TR Shaft Drive Assembly - 1	3T6510A00141	2 Landings
	TR Shaft Drive Assembly - 2	3T6510A00238	2 Landings
	TR Shaft Drive Assembly - 3	3T6510A00337	2 Landings
RL038	TR Drive Couplings (KAMATICS)	3T6510V00152	2 Landings
RL039	Deleted		
RL040	TGB Input Flange Coupling	3T6521A07353	2 Landings
RL041	IGB Output Flange Coupling	3T6521A07353	2 Landings
RL042	TGB Output Housing	3T6522A05338	2 Landings
RL042A	TGB Output Housing	3T6522A05341	2 Landings
RL043A	TGB Centre Housing	3T6522A05147	2 Landings
RL044	IGB Input Flange Coupling	3G6521A00231	2 Landings
	IGB Input Flange Coupling	3T6521A05657	2 Landings
RL045	IGB Housing assy centre	3T6521A05134	2 Landings
RL052	Tail Rotor Servoactuator	3G6730V00731	0.5 FH (Note 1)

E.g.: **For M/R Hub:** Flight time = 2 FH. Number of External Load Cycles = 4. Total accumulated FH = $2 + (1 \times 4) = 6$ FH.

E.g.: **For M/R Shaft:** Flight time = 5 FH. Number of Lifts = 3. Number of landings = 6. Total accumulated FH = $5 + (1 \times 3) = 8$ FH. Total accumulated landings = $6 + (2 \times 3) = 12$ Landings.

Note

- 1 The life penalty must be mandatorily applied to the AW139 Helicopters equipped with the Tail Rotor Elastomeric Damper P/N 3G6420V00851.

6 Operation above 6400 kg (MTOW)

For the following parts in Table 7 - Operation above 6400 kg (MTOW) - Life penalty factor a life penalty must be mandatorily applied to every flight (from take off to landing) in case the total weight exceeds 6400 kg at least one time.

In this case, Multiply the Flying Hours and/or Landings by the specified value, as applicable.

Table 7 Operation above 6400 kg (MTOW) - Life penalty factor

Ref	Part	Part number	Life penalty factor
RL010A	Rod End (Fin End)	3G5510A03931	4.5 FH
RL010C	Rod End (Tailplane End)	3G5510A04052	4 FH
RL010D	Tailplane Assembly	3G5510A00133	4 FH
RL010E	Rod assembly (Tailplane)	3G5510A08931	4.5 FH
RL012	Deleted		
RL012A	Deleted		
RL012B	Deleted		
RL014	Deleted		
RL021	MGB Case, Top Assembly (without Circular Force Active Vibration Control System installed) (Note 1)	3G6320A01033	1.6 FH
RL021A	MGB Case, Top Assembly (with Circular Force Active Vibration Control System installed) (Note 1)	3G6320A01033	1.6 FH
RL024	Deleted		
RL034	Deleted		
RL037	Tail Rotor Shaft Drive Assembly, Number 2	3T6510A00238	2 Landings
	Tail Rotor Shaft Drive Assembly, Number 3	3T6510A00337	2 Landings

E.g.: **For ROD END (Fin End)**: Total daily flight hours = 10 FH. Daily FH related to operations above 6400 Kg = 3 FH. Total Accumulated FH = $(10-3) + (4.5 \times 3) = 20.5$ FH.

Note

- 1 Life penalty factor is not applicable to P/N 3G6320A01035 and subsequents.

7 Guidelines

Daily recording of the following parameters is recommended, refer to Fig 1 :

- Flight Hours
- Landings (coincides with the number of flights)
- Flight Hours performed during Cat. A training as per Note (ref. Table 3 - CAT. A Training Operations - Life penalty factor)
- Rotor starts above 33 kts wind speed as per Note (ref. Table 4 - Rotor Starting or Stopping with wind speed above 33 kts - Life penalty factor)
- Rotor stops above 33 kts wind speed as per Note (ref Table 4 - Rotor Starting or Stopping with wind speed above 33 kts - Life penalty factor)
- External hoist lifts as per Para 4
- External load cycles as per Para 5
- Landings related to operations above 6400 kg as per Para 6
- Flight Hours related to operations above 6400 kg as per Para 6 .

Eg.: 1

- Total Daily Flight Hours = 7.5 FH
- Daily Landings (coincides with the number of flight) = 6
- Daily Flight Hours performed during Cat. A training as per Note = 0.5
- Daily external load cycles as per Para 5 = 4

For the main rotor shaft the following applies:

$$\text{Consumed Life FH} = (7.5 - 0.5) + \boxed{\text{Cat. A operations}} + \boxed{\text{Ext. Load operations (FH)}} = 11.85 \text{ FH}$$

$$(1.7 \times 0.5) + (1 \times 4)$$

$$\text{Consumed Landings} = 6 + \boxed{\text{Ext. Load operations (Landings)}} = 14 \text{ landings}$$

$$(2 \times 4)$$

Eg.: 2

- Total Daily Flight Hours = 7.5 FH
- Daily Landings (coincides with the number of flight) = 6
- Daily Flight Hours performed during Cat. A training as per Note = 0
- Daily rotor starts above 33 kts wind speed as per Note = 2
- Daily rotor stops above 33 kts wind speed as per Note = 1
- Daily external hoist lifts per Para 4 = 0
- Daily external load cycles as per Para 5 = 7
- Daily Landings related to operations above 6400 kg as per Para 6 = 3
- Daily Flight Hours related to operations above 6400 kg as per Para 6 = 2.6

For the antitorque beam the following applies:

$$\text{Consumed Life FH} = 7.5 + \boxed{\text{External Load Cycles (FH)}} = 14.5 \text{ FH}$$

$$(1 \times 7)$$

$$\text{Consumed Landings} = 6 + \boxed{\text{External operations (Landings)}} = 20 \text{ landings}$$

$$(7 \times 2)$$

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Fig 1 Guidelines

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Mandatory inspections

General

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References

Table 1 References

Data module/Technical publication	Title
39-A-25-91-01-00A-310A-K	Rescue hoist - Bumper assembly, cable cutter and cable - General visual inspection
39-A-25-91-01-01A-310A-K	Hook assembly (rescue hoist) - General visual inspection
39-A-25-94-01-00A-310A-K	Double rescue hoist - Bumper assembly, cable cutter and cable - General visual inspection
39-A-25-96-01-00A-310A-K	Single rescue hoist - Bumper assembly, cable cutter and cable - General visual inspection
39-A-53-10-00-00A-310A-A	Forward section - Sponsons - General visual inspection
39-A-53-10-00-00A-31AA-A	Forward section - Main gearbox rod attachment structure - Detailed inspection
39-A-53-10-00-00A-31AB-A	Forward section - Anti-torque beam attachment structure - Detailed inspection

Table 1 References (Continued)

Data module/Technical publication	Title
39-A-53-10-00-00A-31AE-A	Forward section - Engine attachment internal backup structure - Detailed inspection
39-A-53-10-00-00A-31AV-A	Forward section - Frame STA 5700 - Detailed inspection
39-A-53-10-00-00A-31AW-A	Forward section - Rescue hoist external backup structure - Detailed inspection
39-A-53-10-00-00A-31AX-A	Forward section - Double rescue hoist external backup structure - Detailed inspection
39-A-53-10-00-00A-31AY-A	Forward section - Single rescue hoist external backup structure - Detailed inspection
39-A-53-40-00-00A-31AA-A	Tail section - Fin rib attachment - Detailed inspection
39-A-53-40-00-00A-31AB-A	Tail section - Detailed inspection
39-A-53-40-00-00A-31AC-A	Tail section - Tail gearbox attachment structure - Detailed inspection
39-A-53-40-00-00A-31AG-A	Tail section - Tail/rear fuselage attachment fittings - Detailed inspection
39-A-53-40-00-00A-31AH-A	Tail section - Aft spar to skin joint - Detailed inspection
39-A-62-11-00-00A-310A-A	Main rotor blade installation - Blades - General visual inspection
39-A-62-11-01-00A-31AB-A	Main rotor blade - Detailed inspection
39-A-62-22-00-00A-31AC-A	Main rotor head - Elastomeric bearings - Detailed inspection
39-A-62-22-00-00A-31AF-A	Main rotor head - Main rotor hub - Detailed inspection
39-A-62-31-06-00A-31AA-A	Swashplate - Rotating swashplate and duplex bearing attaching parts - Detailed inspection
39-A-64-11-00-00A-310A-A	Tail rotor blade installation - Blades - General visual inspection
39-A-64-11-00-00A-31AB-A	Tail rotor blade installation - Elastomeric bearings - Detailed inspection
39-A-64-21-00-00A-31AD-A	Tail rotor head installation - Lag damper elastomer - Detailed inspection
39-A-75-51-00-00A-31AA-K	Engine inlet particle separator system - Number 1 and Number 2 main panels - Detailed inspection
39-B-53-10-00-00A-31AB-K	Forward section - Frame STA 5700 - Detailed inspection
39-B-53-10-00-00A-31BA-K	Forward section - Main landing gear backup structures - Special detailed inspection
39-B-62-11-01-00A-31AA-A	Main rotor blade - Detailed inspection

Table 1 References (Continued)

Data module/Technical publication	Title
39-C-64-21-00-00A-31AA-A	Tail rotor head installation - Elastomeric lag damper - Detailed inspection
39-C-64-21-00-00A-31AB-A	Tail rotor head installation - Elastomeric lag damper rod-end - Detailed inspection

Description

1 Introduction

The parts listed in the following schedule must be mandatorily inspected when the indicated interval is reached.

2 Affected parts

Refer to Table 2 - Mandatory inspections - List of affected parts.

3 Terms and definitions of Table 2

3.1 Reference (Ref)

This column gives the unique reference [MI (Mandatory Inspections) plus the system number followed by a progressive number] which identifies the part.

3.2 Part

This column gives the part.

3.3 Task

This column gives the description of the inspection to perform.

3.4 Interval

This column gives the inspection interval for the part. Unless otherwise specified, the inspection interval is in Flying Hours (FH). Flying hours are defined as those hours accumulated from take-off to landing.

The inspection intervals of some parts are expressed in "landings" because their inspection intervals are dependent upon the rotor start-stop cycles and the helicopter ground-air-ground cycles.

Inspection intervals are mandatory and cannot be increased. Inspection intervals can be anticipated provided that the next inspections intervals do not exceed the values mandated in this chapter.

Example:

- Task interval: 100 FH. If task is performed at 85 FH, next task must be performed within 100 FH from the previous compliance.

3.5 Reference (DMC)

This column shows the Data Module Code which provides the instructions to perform the inspection.

Table 2 Mandatory inspections - List of affected parts

Ref	Part	Task	Interval	Reference (DMC)
MI25-01	D-Lock External Hoist Hook	General visual inspection for damage	Daily	39-A-25-91-01-01A-310A-K
MI53-01	TGB Fitting (fasteners in skin and spar)	Detailed inspection for cracks (both internal and external sides)	300	39-A-53-40-00-00A-31AC-A
MI53-02	Main Landing Gear LH Panel	General visual inspection for damage	300	39-A-53-10-00-00A-310A-A
MI53-03	Main Landing Gear RH Panel	General visual inspection for damage	300	39-A-53-10-00-00A-310A-A
MI53-04	Tail Structure Assembly (fin rib attachment area)	Detailed inspection for cracks (both internal and external sides)	100	39-A-53-40-00-00A-31AA-A
MI53-05	Tail Structure Assembly (tail cone RH/LH panels upper and lower joint)	Detailed inspection for cracks (both internal and external sides)	300	39-A-53-40-00-00A-31AB-A
MI53-06	Tail Structure Assembly (tail cone joint STA 11020)	Detailed inspection for cracks (both internal and external sides)	300	39-A-53-40-00-00A-31AB-A
MI53-07	Fuselage Structure Assembly (forward MGB rod backup structure)	Detailed inspection for cracks	300 (Note 11)	39-A-53-10-00-00A-31AA-A
MI53-08	Fuselage Structure Assembly (aft MGB rod backup structure and fitting part number 3P5330A11251)	Detailed inspection for cracks	300 (Note 8) (Note 10)	39-A-53-10-00-00A-31AA-A
MI53-09	Fuselage Structure Assembly (antitorque beam backup structure)	Detailed inspection for cracks	1300	39-A-53-10-00-00A-31AB-A
MI53-10	Tail Structure Assembly (tail/rear fuselage attachment fittings)	Detailed inspection for cracks	200 (Note 13)	39-A-53-40-00-00A-31AG-A

Applicable to: All

39-A-04-20-00-00A-028E-P

Table 2 Mandatory inspections - List of affected parts

Ref	Part	Task	Interval	Reference (DMC)
MI53-11	Tail Structure Assembly (aft/front spar to skin joint)	Detailed inspection for cracks (both internal and external sides)	100	39-A-53-40-00-00A-31AH-A
MI53-12	Fuselage Structure Assembly (frame STA 5700, RH/LH side)	Detailed inspection for cracks	100 (Note 5) (Note 6) (Note 7)	39-A-53-10-00-00A-31AV-A
MI53-13	Internal Engine Attachment Backup Structure	Detailed inspection for cracks	1200	39-A-53-10-00-00A-31AE-A
MI53-14	Fuselage Structure Assembly (external hoist backup structure) (Note 3)	Detailed inspection for cracks	4000 External Hoist Lifts (see Para 4)	39-A-53-10-00-00A-31AW-A
MI53-14	Fuselage Structure Assembly (external hoist backup structure) (Note 4)	Detailed inspection for cracks	4000 External Hoist Lifts (see Para 4)	39-A-53-10-00-00A-31AX-A 39-A-53-10-00-00A-31AY-A
MI53-15	Main landing gear backup structure LH/ RH side (Goodrich landing gear installation)	Special detailed inspection for cracks (by using eddy current inspection method)	1250 Landings	39-B-53-10-00-00A-31BA-K
MI53-16	Fuselage Structure Assembly (frame STA 5700, LH/RH side, lower area) (Goodrich landing gear installation)	Detailed inspection for cracks (nut and washer removal required only in the lowest attaching position)	7000 Landings	39-B-53-10-00-00A-31AB-K
MI62-01	Main Rotor Blade Assembly	General visual inspection for damage	50 (Note 2)	39-A-62-11-00-00A-310A-A
MI62-02	Main Rotor Elastomeric Bearing (part number 3G6220V00153)	Detailed inspection for cracks in the elastomer	50	39-A-62-22-00-00A-31AC-A
MI62-03	Main Rotor Elastomeric Bearing (part number 3G6220V00154)	Detailed inspection for cracks in the elastomer	50	39-A-62-22-00-00A-31AC-A
MI62-04	Main Rotor Hub Assembly	Detailed inspection (without removal of blades)	2500	39-A-62-22-00-00A-31AF-A

Applicable to: All

39-A-04-20-00-00A-028E-P

Table 2 Mandatory inspections - List of affected parts

Ref	Part	Task	Interval	Reference (DMC)
MI62-05	Deleted			
MI62-06	Main Rotor Rotating Swashplate	Inspection for integrity and loss of torque of the bolts and ring retaining the duplex bearing	402 (Note 12)	39-A-62-31-06-00A-31AA-A
MI62-07	Main Rotor Blade Assembly with Heater Mat (part number 4G6210A00132)	Detailed inspection (hammer tapping) of the bonding area between tip cap and blade (from STA 6300 to STA 6400) for debondings and/or delaminations (blade removal not required)	100	39-B-62-11-01-00A-31AA-A
MI62-08	Main Rotor Blade Assembly (part number 3G6210A00131)	Detailed inspection (hammer tapping) of the bonding area between tip cap and blade (from STA 6300 to STA 6400) for debondings and/or delaminations (blade removal not required)	100 (Note 9)	39-A-62-11-01-00A-31AB-A
MI64-01	Tail Rotor Blade Assembly	General visual inspection for damage	50 (Note 2)	39-A-64-11-00-00A-310A-A
MI64-02	Tail Rotor Elastomeric Spherical Bearing (part number 3G6420V00153)	Detailed inspection for cracks in the elastomer	100	39-A-64-11-00-00A-31AB-A
MI64-03	Tail Rotor Elastomeric Spherical Bearing (part number 3G6420V00154)	Detailed inspection for cracks in the elastomer	25	39-A-64-11-00-00A-31AB-A
MI64-04	Tail Rotor Damper	Detailed inspection for cracks in the elastomer	100	39-A-64-21-00-00A-31AD-A
MI64-05	Tail Rotor Damper (part number 3G6420V00851)	Detailed inspection of rod end for cracks in the elastomer	200	39-C-64-21-00-00A-31AB-A
MI64-06	Tail Rotor Damper (part number 3G6420V00851)	Detailed inspection of body for cracks in the elastomer	300	39-C-64-21-00-00A-31AA-A

Applicable to: All

39-A-04-20-00-00A-028E-P

Table 2 Mandatory inspections - List of affected parts

Ref	Part	Task	Interval	Reference (DMC)
MI71-00	Engine Air Particle Separator Left and Right Main Panels (part numbers 3G7160V00551 and 3G7160V00651)	Detailed inspection for cracks in the panels	25	39-A-75-51-00-00A-31AA-K

Notes

- 1 Deleted.
- 2 Rotor Starting or Stopping with wind speed above 33 kts. For each Rotor Start or Stop performed with wind speed above 33 kts the additional inspections reported in Table 6 - Rotor Starting or Stopping with wind speed above 33 kts - Additional inspections must be mandatorily carried out.
- 3 Helicopters AW139 that have the Breeze rescue hoist system installed.
- 4 Helicopters AW139 that have the Goodrich rescue hoist system installed.
- 5 The 100 FH interval is extended to 1200 FH if the retro modification P/N 3G5306P08512 is applied to the frames P/N 3P5338A13352 (LH) and 3P5338A13452 (RH).
- 6 The inspection requirement is not applicable to helicopters that install the following frame P/N: 3P5338A13354 (LH) and 3P5338A13454 (RH).
- 7 The 100 FH interval is extended to 1200 FH if the retro modification P/N 3G5306P08513 is applied to the frames P/N 3P5338A13353 (LH) and 3P5338A13453 (RH).
- 8 When the 3750 FH are reached, the 300 FH interval is reduced to 150 FH if the retro modification P/N 3G5306P36711 or P/N 3G5306P39211 is not applied.
- 9 The inspection requirement is applicable to Main Rotor Blade Assembly part number 3G6210A00131 when the 10000 FH are reached.
- 10 The 300 FH interval is extended up to 1300 FH if the Fuselage Structure assembly P/N 3G5300X00135 or the Cruciform fitting retromodification P/N 3G5306P36711 are installed on the rotorcraft.
- 11 The 300 FH interval is extended up to 1300 FH for all helicopters except S/N from 31003 to 31011.
- 12 The inspection requirement is not applicable to helicopters that install the Main Rotor Rotating Swashplate P/N 3G6230A00333.
- 13 The 200 FH interval is extended to 8350 landings for the following configurations:
 - Tail assembly P/N 3G5350A00136
 - Tail assembly P/N 3G5350A00135 plus retromodification P/N 3G5309P02711 (Part I of Optional BT 139-419)
 - Rear Fuselage Assy P/N 3P5340A00134 or P/N 3G5340A11131
 - Rear Fuselage assembly P/N 3P5340A00132 or 3P5340A00133 plus retromodification P/N 3G5306P38811 (Optional BT 139-418)

Table 2 Mandatory inspections - List of affected parts

Ref	Part	Task	Interval	Reference (DMC)
	- Tail assembly P/N 3G5350A00134 or P/N 3G5350A00133 or P/N 3G5350A00132 plus the following retromodifications: 1 P/N 3G5309P02711 (Part I of Optional BT 139-419) 2 P/N 3G5309P01812 (Optional BT 139-200)			

Table 3 Mandatory inspections - Rescue hoist system (Breeze)

Ref	Part	Task	Interval	Reference (DMC)
MI25-02	Rescue hoist	GVI of entire usable length of the cable for damage and condition	Prior the first use of the day	39-A-25-91-01-00A-310A-K

Table 4 Mandatory inspections - Double rescue hoist system (Goodrich)

Ref	Part	Task	Interval	Reference (DMC)
MI25-03	Rescue hoist cable	Do a GVI of the maximum length of cable used during the day operation for damage and condition	After the last flight of the day or before the first flight of the day if the rescue hoist is used or it is envisaged to be used	39-A-25-94-01-00A-310A-K

Table 5 Mandatory inspections - Single rescue hoist system (Goodrich)

Ref	Part	Task	Interval	Reference (DMC)
MI25-04	Rescue hoist cable	Do a GVI of the maximum length of cable used during the day operation for damage and condition	After the last flight of the day or before the first flight of the day if the rescue hoist is used or it is envisaged to be used	39-A-25-96-01-00A-310A-K

Table 6 Rotor Starting or Stopping with wind speed above 33 kts - Additional inspections

Ref	Part	Task	Interval	Reference (DMC)
MI62-01	Main Rotor Blade Assembly	General visual inspection for damage	Within 15 FH from the event	39-A-62-11-00-00A-310A-A

Table 6 Rotor Starting or Stopping with wind speed above 33 kts - Additional inspections

Ref	Part	Task	Interval	Reference (DMC)
MI64-01	Tail Rotor Blade Assembly	General visual inspection for damage	Within 15 FH from the event	39-A-64-11-00-00A-310A-A

4 External hoist lift

The external hoist lift is defined as an unreeling and recovery of the cable with a load attached to the hook, independent of the length of the cable that is deployed/recovered.

An unreeling/recovery of the cable with no load on the hook is not considered to be a lift.

Any operation where a load is applied for half the operation (i.e. unreeling or recovery) must be considered as one lift.

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